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Experimental investigation of failure in thin rings subjected to uniform external pressure

Demyttenaere, Jules Henry

Massachusetts Institute of Technology

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EXPERIMENTAL INVESTIGATION OF
FAILURE IN THIN RINGS SUBJECTED
TO UNIFORM EXTERNAL PRESSURE

Jules Henry Demyttenaere
and
William Joseph Norris

EXPERIMENTAL INVESTIGATION OF FAILURE IN THIN
RINGS SUBJECTED TO UNIFORM EXTERNAL PRESSURE

by

JULES HENRY DEMYTENAERE, LTJG., U.S. NAVY
B.S., United States Naval Academy

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U.S. NAVY
Academy

1954

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Letter on front cover:

EXPERIMENTAL INVESTIGATION OF
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TECHNOLOGY

JULES HENRY DEMYTENAERE
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INVESTIGATION OF FAILURE IN THIN
WALLS UNDER UNIFORM EXTERNAL PRESSURE

DAVID E. AERE, LTJG., U.S. NAVY
Graduate Naval Academy
(1949)

and

WILLIAM JOSEPH NORRIS, LTJG., U.S. NAVY
B.S., United States Naval Academy
(1949)

SUBMITTED IN PARTIAL FULFILLMENT OF THE
REQUIREMENTS FOR THE DEGREE OF
NAVAL ENGINEER

at the

MASSACHUSETTS INSTITUTE OF TECHNOLOGY

May, 1954



EXPERIMENTAL INVESTIGATION OF FAILURE IN THIN
RINGS SUBJECTED TO UNIFORM EXTERNAL PRESSURE

by

JULES HENRY DEMYTENAERE, LTJG., U.S. NAVY
B.S., United States Naval Academy
(1949)

TITLE: EXPERIMENTAL INVESTIGATION OF FAILURE IN THIN RINGS SUBJECTED TO UNIFORM EXTERNAL PRESSURE

AUTHORS: J. H. Demyttenaere, LTJG., U.S.N., and
W. J. Norris, LTJG., U.S.N.

Submitted to the Department of Naval Architecture and Marine Engineering on May 24, 1954, in partial fulfillment of the requirements for the degree of Naval Engineer.

ABSTRACT

The object of this thesis was to investigate experimentally the strain distributions and ultimate failures in thin circular and non-circular aluminum rings subjected to uniform external pressure.

A test apparatus was designed to permit the application of hydraulic pressure to the outer circumference of a ring placed between two sheets of Plexiglas. The small clearance between the edges of the ring and the Plexiglas was sealed by a continuous rubber gasket of slightly greater width fitted around the outer circumference of the ring. Except for friction at the gasket-Plexiglas interface, the ring was free of restraint.

Nine rings of aluminum alloy, 61S-T6, were tested to collapse; the variable in this series was out-of-roundness. Correlation between measured and predicted strains was obtained in five of the six rings in which the assumptions of the prediction were valid. A somewhat arbitrary strain distribution was observed in the two most circular rings. The four rings of moderate out-of-roundness collapsed very near predictions based upon a criteria related to the maximum stress level in the outer fibers while the three rings of relatively large out-of-roundness failed at pressures somewhat higher than predicted. The two most circular rings collapsed near a computed pressure when the minimum rather than average section thickness was used in the calculation. Collapse pressures were predicted most accurately when the stress used in conjunction with the failure criteria was defined by the point on the stress-strain curve at which marked non-linearity occurred.

An analysis of the results indicates that strain distributions in an out-of-round ring loaded by uniform external pressure can be predicted with satisfactory

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accuracy provided the assumptions made in the prediction are satisfied; namely, that the thickness to diameter ratio be the order of 0.0285 and that the initial configuration assumed is actually predominate. A failure criteria based upon the stress level in the outer fibers will, when the assumptions above are fulfilled, predict collapse quite accurately in rings of out-of-roundness to thickness ratio between 0.10 and 0.30. At larger values of out-of-roundness the criteria is conservative but not over cautious.

It is recommended that the experimentation be extended to include a quantitative investigation of the effects of deviations from a basic two lobe out-of-roundness such as might be obtained in a practical situation. Such an investigation should be conducted using rings of constant thickness in order to avoid many of the unnecessary complications encountered in the analysis of data obtained in this thesis.

accuracy provided the measurements made in the radiation
 are satisfied; namely, that the thickness is constant, that
 of the order of 1.025 and that the initial concentration
 measured is actually proportional. A further difficulty arises
 when the stress level in the outer fibers will, when the
 assumptions above are utilized, require certain modifications
 accordingly to that of 0.01-0.02 to 0.03-0.04, and the ratio
 between 0.10 and 0.20. At larger values of 0.01-0.02
 the criteria is considered for not over 0.01.

It is recommended that the experiment be extended
 to include a quantitative investigation of the stress of
 deviation from a basic two load two-ounce test as
 might be required in a practical situation. It is in-
 vestigative studies be conducted using rings of constant
 thickness in order to avoid many of the unnecessary compli-
 cations encountered in the analysis of data obtained in
 this study.

Cambridge, Massachusetts
May 24, 1954

Professor Leicester F. Hamilton
Secretary of the Faculty
Massachusetts Institute of Technology
Cambridge, Massachusetts

Dear Sir:

In accordance with the requirements for the degree of Naval Engineer, we submit herewith a thesis entitled, "Experimental Investigation of Failure in Thin Rings Subjected to Uniform External Pressure."

ACKNOWLEDGMENTS

The authors wish to express their appreciation to those individuals at the Massachusetts Institute of Technology, Boston Naval Shipyard, and the David Taylor Model Basin, who so generously gave their time and advice so that the objectives of this thesis might be accomplished.

Specifically the authors wish to thank Dr. E. Wenk, Jr., of the David Taylor Model Basin, who originally posed many of the problems involved; Dr. W. W. Murray, of the Massachusetts Institute of Technology, and his assistant Mr. Peter Stein for their most helpful advice on a multitude of details..

The authors are particularly grateful to their Thesis Supervisor, J. H. Evans, Assistant Professor of Naval Architecture, Massachusetts Institute of Technology, for his encouragement and advice during the progress of the thesis.

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The author wishes to express their appreciation to those individuals at the Massachusetts Institute of Technology, Boston Naval Shipyard, and the David Taylor Model Basin, who so generously gave their time and advice on what the objectives of this thesis might be accomplished.

Specifically the author wishes to thank Dr. E. Vess, Jr., of the David Taylor Model Basin, who originally posed some of the problems involved; Mr. W. W. Curry, of the Massachusetts Institute of Technology, and his assistant Mr. Peter Stein for their most helpful advice on a multitude of details. The authors are particularly grateful to their thesis supervisor, J. N. Evans, Assistant Professor of Naval Architecture, Massachusetts Institute of Technology, for his encouragement and advice during the progress of the thesis.

SYMBOLS AND ABBREVIATIONS

The following symbols and abbreviations are used throughout this report:

b = Width of ring, in inches

D = Mean diameter measured to neutral axis, in inches

E = Modulus of elasticity - initial slope of stress-strain curve, psi.

ϵ = Compressive strain, inches/inch or micro-inches/inch.

h = Thickness of ring, in inches.

I = Moment of inertia of ring cross-section about neutral axis, in in.⁴

P = Pressure, psi.

$P_{crit.}$ = Collapse pressure of a circular ring, psi.

psi. = Pounds per square inch.

R = Mean radius to neutral axis of ring, in inches.

R_o = Mean radius to outer circumference of ring, in inches.

σ = Compressive stress, psi.

σ_y = 0.2% proof stress, psi.

θ = Angular coordinate to designate positions on ring, in degrees.

u_o = Two-lobe out-of-roundness, maximum deviation from a true circle of the same perimeter as the two-lobe configuration, in inches.

u_1 = Three-lobe out-of-roundness, inches

u_2 = Four-lobe out-of-roundness, inches

SYMBOLS AND ABBREVIATIONS

The following symbols and abbreviations are used

throughout this report:

δ = Width of ring, in inches

Δ = Mean diameter measured to neutral axis, in inches

α = Radius of elasticity - initial slope of stress-strain curve, psi.

β = Compressive stress, in pounds per square inch

γ = Thickness of ring, in inches

λ = Moment of inertia of ring cross-section about neutral axis, in in.⁴

μ = Poisson's ratio

ν = Collapsed pressure of a closed ring, psi.

ρ = Density per square inch

σ = Mean stress to neutral axis at ring, in inches

σ_0 = Mean stress to outer circumference of ring, in inches

σ_c = Compressive stress, psi.

σ_t = Tensile stress, psi.

θ = Angular coefficient to diameter position on ring, in degrees

ω = Two-lobe out-of-roundness, maximum deviation from a true circle of the same diameter at the two-lobe configuration, in inches

ω_1 = Three-lobe out-of-roundness, inches

ω_2 = Four-lobe out-of-roundness, inches

Experimental Investigation of Failure in Thin
Rings Subjected to Uniform External Pressure

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I.

INTRODUCTION

Objective

The objective of this thesis is to investigate experimentally the strain distributions and ultimate failures obtained in thin circular and non-circular aluminum rings subjected to uniform external pressure.

Background

The slender column in axial compression is generally cited as the classic example of a buckling failure. The behavior of such axially loaded columns was subjected to an early theoretical analysis, and at the same time numerous investigators supplied experimental data relevant to the problem. The experimentation served two purposes. First, the conclusions substantiated the basic concepts and accuracy of the theoretical analysis. Second, certain practical aspects of the problem were magnified. Thus, through an analysis of experimental data and an appreciation of the difficulty in obtaining a perfectly loaded column, Moncrieff (1) was able to propose a practical criteria for column strength which included the assumption of an inherent eccentricity.

The failure of thin rings subjected to uniform external pressure is analogous to the buckling of columns.

INTRODUCTION

Objective

The objective of this thesis is to investigate experimentally the strain distributions and ultimate failures obtained in thin aluminum and non-aluminum aluminum alloys subjected to uniform tensile stresses.

Background

The slender column is a structural member of primary interest in the design of a machine structure. The behavior of such a column under load is dependent on an early theoretical analysis, and at the same time numerous investigators have applied experimental data relevant to the problem. The experimental work has been divided into two main groups. First, the compressive strength of the column is determined and accuracy of the theoretical analysis. Second, certain practical aspects of the problem have been investigated. These have been an analysis of experimental data and an investigation of the difficulty in obtaining a perfectly loaded column. However, it is well to review a practical objective for columns which are loaded for compression of an inherent uncertainty.

The failure of thin columns subjected to uniform external pressure is analogous to the buckling of columns.

The buckling of such uniformly loaded rings has been treated theoretically by investigators as early as Levy (2) and as recent as Boresi (3). In addition a clear presentation of the problem is given by Timoshenko (4). Despite the extensive theoretical analysis, experimentation has been lacking. In a search of the literature, the authors found no instance of a ring being collapsed experimentally under a uniform external load.

The lack of experimental data relative to the ring problem may be partially explained by the following:

1. The use of the ring as a structural member is not so widespread as the column. Hence the need for usable data was not pressing.
2. Many of the conclusions formulated as a result of column analysis were applicable to the ring problem.
3. The buckling of a cylinder subjected to external pressure is closely related to the buckling of a ring. Obviously much of the data gleaned from failure of cylinders also applies to rings. The collapse of a cylinder may be effected simply and conveniently in the laboratory.
4. A free ring loaded with uniform external pressure offers a difficult problem in the design of a test apparatus.

The buckling of such radially loaded rings has been treated theoretically by investigators as early as Levy (2) and as recent as Bessel (3). In addition a clear presentation of the problem is given by Timoshenko (4). Despite the extensive theoretical analysis, experimental work has been lacking. In a section of the literature the authors found an instance of a ring being collapsed experimentally under a uniform external load.

The lack of experimental data relative to the ring problem may be partially explained by the following:

1. The use of the ring as a structural member is not as widespread as the column. Hence the need for reliable data was not pressing.
2. Many of the conditions formulated as a result of column analysis were applicable to the ring problem.
3. The buckling of a cylinder subjected to external pressure is closely related to the buckling of a ring. Obviously much of the data gathered from failure of cylinders also applies to rings. The collapse of a cylinder may be effected singly and conveniently in the laboratory.
4. A ring loaded with uniform external pressure offers a different problem in the design of a test apparatus.

Despite the apparent validity of the theoretical analysis of ring buckling and the availability of related experimental data, there is a need for experimentation devoted specifically and directly to the collapse of a thin ring. This need arises not merely from the satisfaction to be gained by proving the validity of the concepts or assumptions of the pertinent theory but from a desire to obtain a practical perspective. Thus, some peculiarity of the ring problem may yet be unrecognized; practical expedients such as the assumption of an inherent out-of-roundness may be necessary. Furthermore, there is need for a substantial and tested criteria of failure which may or may not be related to the stress level at the inner or outer surface of the ring. It is to be appreciated that in out-of-round rings, failure results when stable equilibrium is not possible between an internal and external bending moment and is not a direct result of some specified stress in the outer fibers. In addition, the ring provides a convenient means of studying and evaluating out-of-roundness not only in rings but also in cylinders. In a theoretical approach out-of-roundness may be represented very simply by a Fourier series; there is need for a simple, yet reliable, measure of out-of-roundness which has a practical value.

Problem

The authors have undertaken the task of obtaining and

...the apparent validity of the theoretical
analysis of ring working and the availability of related
experimental data, there is a need for experimental
devices specifically and directly to the solution of a
ring ring. This need arises not merely from the neces-
sity to be gained by proving the validity of the concepts
or assumptions of the present theory but from a desire to
obtain a practical perspective. Thus, some peculiarity of
the ring problem may yet be unaccounted; practical experience
such as the assumption of an inherent out-of-roundness may
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in rings but also in cylinders. In a practical approach
out-of-roundness may be represented very simply by a
fourier series; some is need for a simple, yet reliable,
means of out-of-roundness which has a practical value.

Problem

The authors have undertaken the task of obtaining and

evaluating experimental data of the type referred to above.

Implied in this assignment are the following specific problems:

1. The design and construction of a test apparatus which will permit the application of a uniform external load to an essentially free ring.
2. The instrumentation of a ring such that the strain distribution may be determined.
3. A comparison of measured strain distribution and failure loads with theoretical predictions.
4. A review of experimental results for the purpose of pointing out the practical aspects of the ring buckling problem.

overlapping experimental base of the new interest to move.
Applied in this experiment are the following specific

procedures:

1. The design and construction of a test apparatus which will permit the application of a uniform external load to an essentially free ring.
2. The determination of a ring such that the static distribution may be determined.
3. A comparison of measured static distribution and finite loads with theoretical prediction.
4. A review of experimental results for the purpose of defining the practical aspects of the ring loading problem.

II.

PROCEDURE

Outline

The procedure followed in this thesis is to be presented under the following subheadings:

- (a) Selection of Material for Construction of Rings
- (b) Manufacture of Rings
- (c) Ring Dimensions and Instrumentation
- (d) Design and Manufacture of Test Apparatus
- (e) Proof Test of Apparatus
- (f) Test of Rings
- (g) Test of Compression Specimens
- (h) Evaluation of Data
- (i) Correlation of Data

PROCEDURE

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- (g) Test of Compression Specimens
- (h) Evaluation of Data
- (i) Correlation of Data

Selection of Material for Construction of Rings

Rings were machined from a length of 9" aluminum alloy (61S-T6) tubing of 1/4" wall thickness. The actual outside diameter as determined by the authors was 9.016".

The decision to use aluminum alloy test rings was based upon the following considerations:

1. To properly correlate measured and predicted strain distributions it was necessary to select a material in which the stress-strain relationship was essentially linear over a considerable range of stress. Aluminum alloy 61S-T6 was suitable in this respect since the estimated 0.2% proof stress was 40,000 psi. (5).
2. The predicted failure pressures for rings of reasonable diameter and thickness were satisfactory. Reasonable dimensions were defined as any diameter and thickness of tubing or pipe which was available and which could be accommodated by the facilities in the testing laboratory. A relatively high failure pressure was desirable in that sufficient accuracy could be obtained without the use of an extremely sensitive pressure measuring device. Yet the maximum pressure should not be so high as to require an excessively complicated and expensive test apparatus.

Selection of Material for Construction of Rings

Rings were specified from a length of 2' minimum alloy (613-T6) of 1 1/4" wall thickness. The actual outside diameter as determined by the supplier was 9.000".

The decision to use aluminum alloy cast rings was based upon the following considerations:

1. To properly correlate measured and predicted strain distributions it was necessary to select a material in which the stress-strain relationship was essentially linear over a considerable range of stress. Aluminum alloy 613-T6 was available in this respect since the estimated 0.2% proof stress was 40,000 psi. (5).

2. The predicted failure pressures for rings of reasonable diameter and thickness were satisfactory. Resonance dimensions were defined as any diameter and thickness of tubing or pipe which was available and which could be accommodated by the facilities in the testing laboratory. A relatively high failure pressure was desirable in that sufficient accuracy could be obtained without the use of an extremely sensitive pressure measuring device. For the various pressures should not be so high as to require an excessively complicated and expensive test apparatus.

Manufacture of Rings

The aluminum alloy tubing as received was cut into sections of 2 1/2" to 3 1/2" lengths. Out-of-roundness was then intentionally introduced in the majority of these pieces by placing the individual sections in a loading machine and applying a load. When the desired permanent set had been obtained the load was removed. Since essentially circular rings were to be machined from the remainder of the sections no deformation was introduced. In this manner a range of out-of-roundness was obtained which varied from $u_0 = 0.007$ " in the tubing as received to $u_0 = 0.296$ " in the section with the greatest intentional deformation.

As a result of the permanent set imposed, there existed an unknown residual stress at points stressed above the elastic limit. To insure linearity of measured strains over a maximum range, the sections were subjected to a solution heat treatment followed by the precipitation heat treatment required to obtain alloy 61S-T6. Uniformity of properties was insured by heat treating, as a group, all sections plus the material from which compression specimens were to be manufactured.

Wooden plugs about 1 inch in thickness were tailored to fit into one end of each section in order to facilitate final machining and to avoid distortion caused by the clamping jaws of the lathe. Rings of specified width were

Manufacture of Rings

The aluminum alloy tubing as received was cut into sections of $3\frac{1}{2}$ " to $5\frac{1}{2}$ " lengths. Out-of-roundness was then intentionally introduced in the majority of these pieces by placing the individual sections in a loading machine and applying a load. When the desired permanent set had been obtained the load was removed. Since essentially circular rings were to be machined from the remainder of the sections no deformation was introduced. In this manner a range of out-of-roundness was obtained which varied from $u_o = 0.007$ " in the tubing as received to $u_o = 0.120$ " in the section with the greatest intentional deformation. As a result of the permanent set imposed, there existed an unknown residual stress at points stressed above the elastic limit. To insure linearity of measured strains over a certain range, the sections were subjected to a relatively heat treatment followed by the precipitation heat treatment required to obtain alloy 4142. Uniformity of properties was insured by heat treating, as a group, all sections plus the material from which compression specimens were to be manufactured. Wooden plugs about 1 mm in thickness were drilled to the ends and of even section in order to facilitate final machining and to avoid distortion caused by the clamping force of the lathe. Rings of specified sizes were

then cut from the sections; this width was determined by a direct measurement of the clearance between the surfaces of the annular test chamber in the completed test apparatus. The rings as cut were generally 0.444" to 0.446" in width, the maximum deviation for a single ring being 0.001". The thickness of the rings remained the same as received.

Ring Dimensions and Instrumentation

The outside diameter of each ring was found by fitting a wire 0.01" in diameter around the outer circumference. The wire was scribed and then measured between the scribe marks on a 36" steel rule. An outside radius was computed by dividing the measured circumference by 2π and correcting for the thickness of the measuring wire. All radii thus determined were averaged to give a mean outside radius (4.508") for use in the computations.

The measuring wire was scribed at intervals of 30 degrees of arc length and again fitted around each ring. The scribe marks were transferred to the aluminum alloy. For those rings in which deformation had been introduced care was taken to locate one of the scribe marks as close as possible to the point of maximum diameter. The twelve stations so determined were designated 0° through 330° in 30° increments, with 0° at a point of maximum diameter. The thickness and width at each station were measured with small micrometers

then cut from the sections; this wire was determined by a direct measurement of the distance between the surfaces of the smaller steel cylinder in the completed test specimen. The rings of the wire were generally 0.444" to 0.445" in width, the maximum variation for a single ring being 0.001".

The thickness of the rings remained the same as described.

Wire Identification and Location

The outside diameter of each ring was found by dividing a wire 0.01" in diameter around the outer circumference. The wire was divided and then measured between the surface of the wire on a 36" steel rule. An outside reading was completed by dividing the measured circumference by π and multiplying for the thickness of the measuring wire. The rings were determined with reference to the wire which was used (4.500") for the in the comparison.

The measuring wire was used in intervals of 10 degrees of arc length and again divided about each ring. The rings were then transferred to the aluminum alloy. For rings in which deformation had been introduced after the rings were one of the series were as close as possible to the point of maximum interest. The better position was determined with reference to the maximum diameter. The thickness and width of each section were measured with small microscopes.

while the outside diameter was measured at stations and half stations with large outside micrometers.

Baldwin SR-4, type A-7, strain gages were cemented to the inner surface of the ring at each station with the gage length along the circumference. Photograph No. 1 shows the strain gages in position on the ring after the leads had been soldered in place. The leads were passed through the bottom of the test apparatus and connected to the multiple selector switchbox which in turn was connected to the strain indicator. The dummy gage was attached to a scrap piece of aluminum alloy which was placed in the vicinity of the test apparatus. Leads from the dummy were connected to the switchbox.

Design and Manufacture of Test Apparatus

The design of the apparatus and its associated equipment can be understood from photographs Nos. 2, 3, 4 and Figures I and II. The bucket pump had been previously tested up to a pressure of 1600 psi and was considered to be of sufficient capacity to supply oil to the pressure chamber enclosing the ring.

The test apparatus was made by the Mechanical Engineering Department machine shop at the Massachusetts Institute of Technology to plans and specifications furnished by the authors. It is to be noted that provision was made for a

While the outside diameter was measured at stations and built

stations with large outside micrometers.

Boiler No. 4, type 4-7, strain gages were cemented to

the inner surface of the ring at each station with the same

method along the circumference. Photograph No. 1 shows the

strain gages in position on the ring after the leads had

been soldered in place. The leads were passed through the

bottom of the test apparatus and connected to the multiple

selector switchbox which in turn was connected to the strain

indicator. The dummy gages were attached to a scrap piece of

aluminum alloy which was placed in the vicinity of the test

apparatus. Leads from the dummy were connected to the

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ing Department machine shop at the Massachusetts Institute

of Technology to plans and specifications furnished by the

author. It is to be noted that provision was made for a

Photograph No. 1

RING WITH STRAIN GAGES AND LEADS



FROM MEMPHIS
APRIL 21 1968
MEMPHIS, TENN



Photograph No. 2

TEST APPARATUS



5. ON 04/20/01

TEST AREA



Photograph No. 3

TEST APPARATUS



PROLOGUE NO. 1
MUTATION TEST

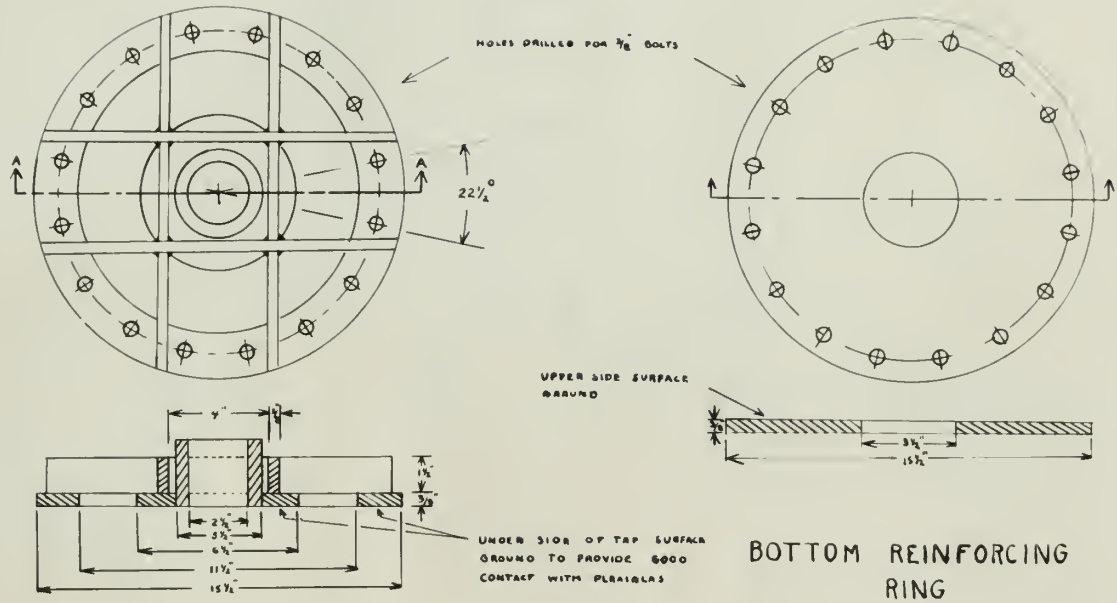
Photograph No. 4

TEST APPARATUS WITH CATHETOMETER MOUNTED

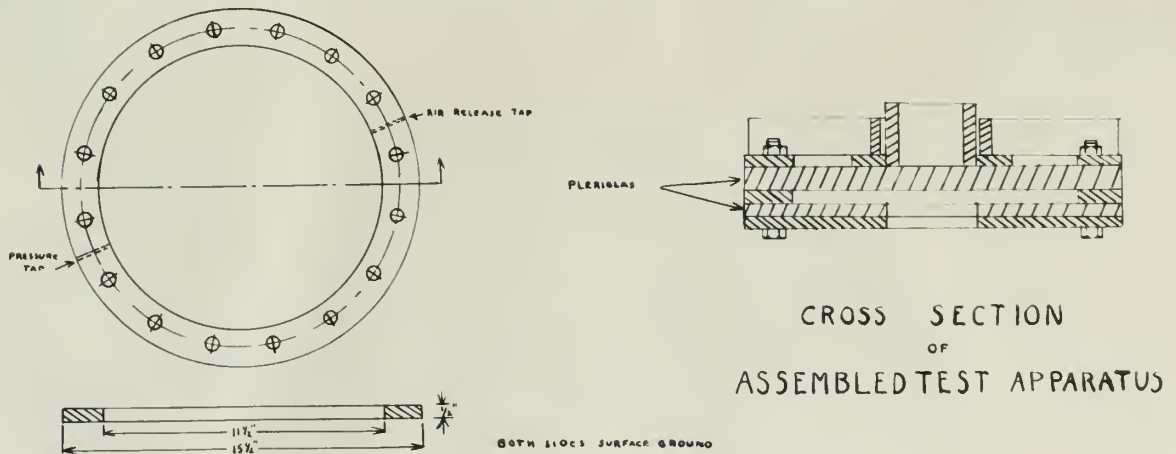


TEST SUBJECTS WITH CONTINUOUS RECORD
IN 1934-35

FIGURE I
TEST APPARATUS



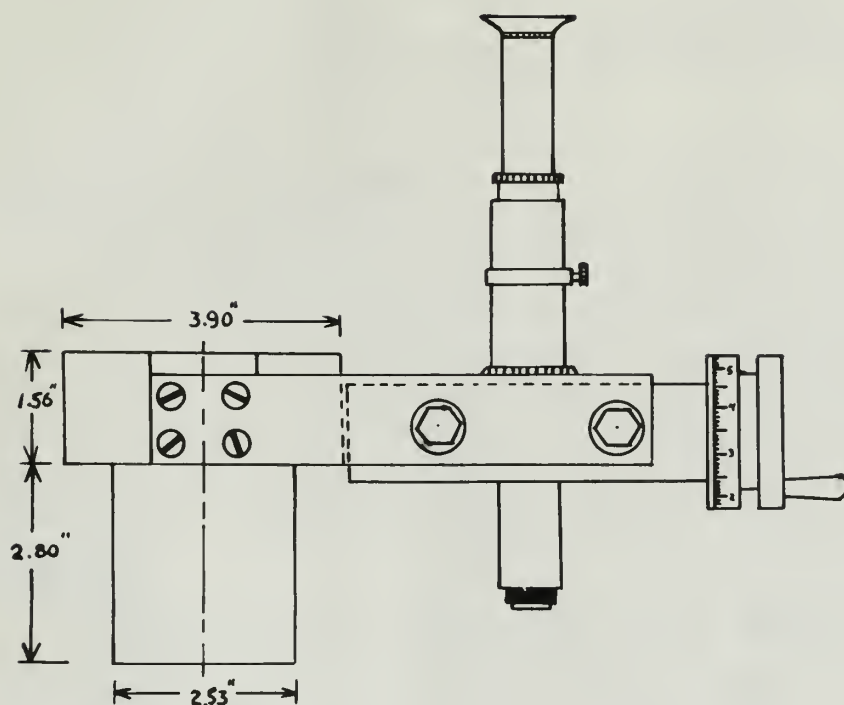
TOP ASSEMBLY



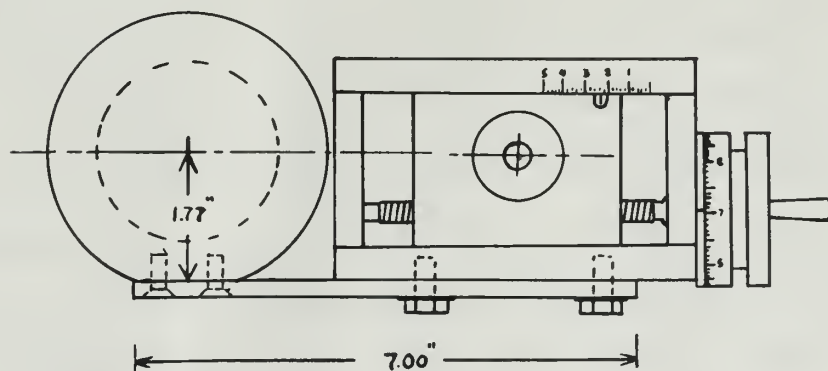
SPACER RING

4/13/54 YHD - 297

FIGURE II
CATHETOMETER MOUNTING



SIDE VIEW



TOP VIEW

7/24/54 YAD-Wgn

viewing chamber which would permit inspection of the rings during application of pressure. The pressure gage was furnished by the Department of Mechanical Engineering, MIT, and calibrated by the authors.

Proof Test of Apparatus

Upon completion of the test chamber by the machine shop, the complete test apparatus as indicated in Photographs Nos. 2, 3, and 4 was assembled and proof tested by the authors. Two preliminary test rings were machined from a section of the original aluminum tubing for use in this phase of the testing.

To maintain pressure in the test chamber behind the aluminum rings, a gasket of width slightly greater than $1/2$ " was placed around the outer circumference of the ring; several types of gasket material were tried. In each case the gasket material was cut to size, glued to the outer surface of the ring, and bound lightly by several strands of light string lying along the circumference. The ring was then placed in the test apparatus and the upper Plexiglas surface and top web assembly were bolted down. Oil was pumped into the test chamber with the bucket pump and the air vented through the air release tap. Pressure was built up behind the ring in the test chamber until, in most cases, the gasket material extruded between the Plexiglas surfaces

viewing chamber which would permit inspection of the ring during application of pressure. The pressure gauge was furnished by the Department of Mechanical Engineering, MIT, and calibrated by the authors.

Proof Test of Apparatus

Upon completion of the test chamber by the machine shop, the complete test apparatus as indicated in Photographs Nos. 2, 3, and 4 was assembled and proof tested by the authors. Two preliminary test rings were machined from a section of the original aluminum tubing for use in this phase of the testing.

To maintain pressure in the test chamber during the aluminum rings, a gasket of width slightly greater than $1\frac{1}{2}$ " was placed around the outer circumference of the ring; several types of gasket material were tried. In each case the gasket material was cut to size, fitted to the outer surface of the ring, and held tightly by several strands of light wire along the circumference. The ring was then placed in the test apparatus and the upper flange turned and top was assembly was bolted down. Oil was poured into the test chamber with the gasket pump and the air vented through the air release valve. Pressure was built up behind the ring in the test chamber until, in most cases, the gasket material stretched between the flanges without

and the edges of the ring. As a result of these trials, the best gasket material for sealing the ring edges appeared to be a rubber gasket cut from an ordinary truck tire inner tube approximately 9" in diameter. This type of gasket was used throughout the remainder of the proof tests and the final tests. Pressures up to 875 psi. were maintained in the test chamber using this gasket.

An Ames dial gage was mounted on the steel web assembly midway between the top inner and outer backing up rings to determine the maximum deflection of the Plexiglas surface as pressure was applied. A maximum deflection of 0.007" occurred at 875 psi. The deflection occurring at 500 psi. was of the order of 0.004".

It became apparent during these tests that some type of lubricant was necessary to eliminate or reduce to a minimum the friction between the rubber gasket and the Plexiglas surfaces. Several types of lubricant were tried and the best found to be black rubber-to-metal cement. Strains at a particular station were found to vary with the lubricant and gasket material when the same pressure was applied.

The cathetometer was mounted during the preliminary test phase and attempts were made to take deflection readings. Unfortunately the cathetometer proved impractical as a means

and the edges of the ring. As a result of these trials, the best tested material for sealing the ring was determined to be a rubber gasket one inch in diameter. This type of gasket was used throughout the remainder of the proof tests and the final tests. Pressures up to 875 psi. were maintained in the test chamber using this gasket.

An Ames dial gage was mounted on the steel wall assembly midway between the top inner and outer backing up rings to determine the maximum deflection of the Plexiglas surface as pressure was applied. A maximum deflection of 0.007" occurred at 875 psi. The deflection occurring at 500 psi. was of the order of 0.004".

It became apparent during these tests that some type of lubricant was necessary to eliminate or reduce to a minimum the friction between the rubber gasket and the Plexiglas surfaces. Several types of lubricant were tried and the best found to be a light rubber-to-metal cement. Trials of a particular reaction were found to vary with the lubricant and gasket material when the test pressure was applied.

The catheterometer was mounted during the preliminary test phase and attempts were made to take deflection readings. Unfortunately the catheterometer proved impractical as a means

of reading deflection for several reasons, viz.

1. The slight deflection of the upper Plexiglas surface distorted the ring as seen through the eye piece of the cathetometer.
2. The use of black rubber-to-metal cement as a lubricant obscured the aluminum ring in the test chamber when oil was introduced.

The proof testing period was of infinite value to the authors in planning and carrying out the final tests of the circular and non-circular rings.

Test of Rings

One result of the proof testing was the formulation of a standard test procedure. As a consequence, the remainder of the tests were made in a minimum amount of time and with little difficulty.

Nine rings were tested in accordance with the following steps:

1. Leads were soldered to strain gages, threaded through the bottom of the test chamber, and connected to terminals on the back of the multiple selector switchbox.
2. Black rubber-to-metal cement was applied to the rubber gasket and outer circumference of the ring. (Cement allowed to dry 3-5 minutes).
3. The rubber gasket was slipped around the outer

of testing facilities for several reasons, viz.

1. The slight deflection of the upper flexure surface illustrated the ring as seen between the eye piece of the microscope.

2. The use of blind rubber-to-metal contact as a lubricant obscured the aluminum ring in the test chamber when oil was introduced.

The proof testing period was of infinite value to the authors in planning and carrying out the final tests of the circular and non-circular rings.

Tests of Rings

The result of the proof testing was the formation of a standard test procedure. As a consequence, the remainder of the tests were made in a minimum amount of time and with little difficulty.

When rings were tested in accordance with the following steps:

1. Bands were selected to desired sizes, measured through the bottom of the test chamber, and connected to terminals on the back of the multi-selector switch.
2. Blind rubber-to-metal contact was applied to the rubber contact and outer circumferences of the ring. (Current allowed to dry 5-6 minutes).
3. The rubber contact was slipped around the outer

circumference of the ring and secured in place by four strands of light string. The gasket was adjusted so that the bevels extended an equal distance beyond the edges of the ring.

4. A thin film of #40 S.A.E. motor oil was applied to the inner Plexiglas surfaces.
5. The ring was centered in the test chamber.
6. The upper Plexiglas plate was slipped over the bolts and held down firmly by hand while the top web assembly was lowered into place.
7. All bolts were tightened.
8. Initial zero readings were set on all strain gages by adjusting switchbox set screws.
9. Air was vented through the air release tap while the test chamber was being filled with oil from the bucket pump.
10. Pressure in the chamber was increased in increments of 50 to 100 psi. depending upon the expected collapse pressure of the particular ring, 100 psi. increments being used for the more circular rings and 50 psi. increments for the more out-of-round rings. Strain gage readings were taken at each pressure increment.
11. Because the collapse of the more circular rings occurred rather quickly, the pressure was maintained

1. Adjustment of the ring and secured in place by two strands of light wire. The strands were adjusted so that the device extended an equal distance beyond the edge of the ring.
2. A thin film of 440 E.A.E. motor oil was applied to the inner flexible surfaces.
3. The ring was centered in the test chamber.
4. The upper flexible plate was slipped over the bolts and held down firmly by hand while the top and bottom assembly was lowered into place.
5. All bolts were tightened.
6. Initial zero readings were set on all strain gauges by adjusting switchbox set screws.
7. Air was vented through the air release tap while the test chamber was being filled with oil from the pressure pump.
8. Pressure in the chamber was increased in increments of 50 to 100 psi, depending upon the required collapse pressure of the test specimen, 100 psi. Instruments being used for the test specimen were 50 psi. Instruments for the core were 100-psi range. Strain gauge readings were taken at each pressure increment.
9. Between the collapse of the test specimen there occurred rather quickly, the pressure was maintained

momentarily at 5 psi. increments when the ring was near collapse. The collapse of a significantly out-of-round ring generally proceeded at a very slow rate. In this case the strain indicator was connected to read the maximum strain. Pressure was then increased in steps of 5 psi. or smaller, each pressure being maintained until the strain indication steadied.

12. When the strain indicator needle did not steady, failure occurred. Failures in the more circular rings were evidenced by an immediate drop in pressure and obvious deformations.

Collapse pressure was thus defined as that pressure at which static strains could not be maintained. Complete failure of all rings was characterized by large, visible deflections regardless of the rate at which collapse proceeded; once complete failure occurred the ring would not again sustain collapse pressure.

Test of Compression Specimens

The stress-strain curve for the material used in the fabrication of the rings for this investigation was determined by means of the "Single Thickness" compression test method. Three specimens of the aluminum alloy (61S-T6) were tested in the compression block as indicated in photographs Nos. 5 and 6. A complete description and

approximately 10 to 15 minutes from the time the
first explosion occurred. The collapse of a structure
out-of-town was generally preceded by a very
slow rise. In this case the first explosion was
followed by a very slow rise. The second explosion was
then followed by a rise of 2 in. or more, and
pressure being maintained until the second explosion
occurred.

12. When the whole structure began to rise,
failure occurred. Failure in the case of circular
trials were witnessed by an immediate drop in
pressure and obvious deformation.

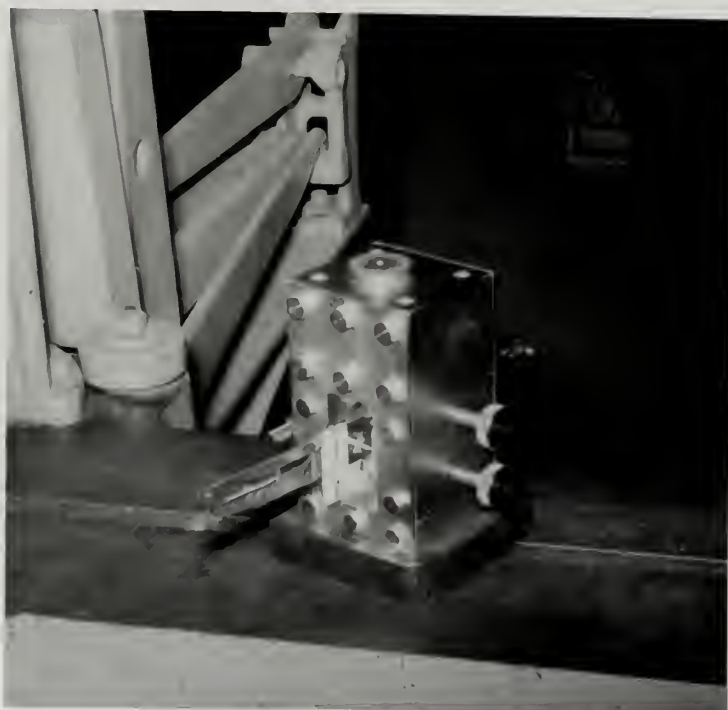
Collapses pressure was then defined as first pressure
at which static stress ratio was maintained. Complete
failure of all types was characterized by large, visible
deformation, regardless of the rate at which collapse proceeded;
once complete failure occurred the rise would not again
establish collapse pressure.

Test of Compression Specimens

The stress-strain curve for the material used in the
test is shown in the table for this investigation was made
known by means of the "Single Specimen" compression test
method. These specimens of the material also (101-71)
were tested in the compression mode as indicated in
photograph test 2 and 3. A complete description of

Photograph No. 5

HUGGENBERGER TENSOMETERS MOUNTED
ON SPECIMEN IN COMPRESSION BLOCK



Enclosure No. 1

RECEIVED BY THE
U.S. DEPARTMENT OF AGRICULTURE



Photograph No. 6

COMPRESSION BLOCK IN LOADING MACHINE



Photograph No. 9

CONFESSION BLOOD IN LAMBERT'S HANDS



bibliography for this method is given in reference (6). Compression tests in general are also discussed in reference (5).

Evaluation of Data

All data obtained during the testing period can be grouped as follows:

1. Dimensions
2. Pressure at which strains were recorded
3. Pressure at which ring collapsed
4. Strain readings
5. Compression Specimen test data.

With one exception all dimensions recorded were taken with micrometers and can be considered accurate to 0.0005". The circumference, as determined by the scribed wire, was read to the nearest 1/128" on the steel rule. However, the authors felt that the accuracy was of the order of 1/64", one scale division. The accuracy of the average outside radius then becomes 0.0025".

The pressure gage was calibrated by the authors prior to the tests using a dead-weight tester. Two calibration runs were made using increasing pressures and two with decreasing pressures. The maximum deviation from true was found to be 4 psi.; the largest difference between an up reading and a down reading for the same true pressure was also 4 psi. Used in conjunction with a calibration curve

graphology for this method is given in reference (5).

Compression tests in general are also discussed in

reference (5).

Evaluation of Data

All data obtained during the testing period can be

grouped as follows:

1. Dimensions
2. Pressure at which strains were measured
3. Pressure at which ring collapsed
4. Strain readings
5. Compression specimen test data.

With one exception all dimensions recorded were taken with calipers and can be considered accurate to 0.005". The circumference, as determined by the scriber wire, was read to the nearest 1/16" on the steel rule. However, the authors felt that the accuracy was of the order of 1/64", one scale division. The accuracy of the average diameter readings then becomes 0.005".

The pressure gauge was calibrated by the weight piston to the gauge using a dead-weight tester. Two determinations were made using increasing pressures and two with decreasing pressures. The maximum deviation from zero was found to be 4 psi; the largest difference between any two readings was a difference of 10 psi. The pressure gauge was also used in conjunction with a calibration curve

the gage was considered to be accurate to ± 2 psi. although the accuracy was considerably better over most of the pressure range. During the tests the indicated pressure could be held fairly constant. An accuracy of plus or minus one half a scale division ($\pm 2 \frac{1}{2}$ psi.) is probably conservative but will be assumed. The accuracy of the recorded pressures at which strain readings were taken then becomes $\pm 5 \frac{1}{2}$ psi.

With one additional consideration the above remarks also apply to the accuracy of the recorded collapse pressures. The collapse of rings with significant amounts of out-of-roundness occurred rather slowly. It would seem that the pressure could be inadvertently increased above the true collapse pressure, the only effect being an accelerated failure. However, in these instances the strain gage indicating the greatest strain was checked at small increments of pressure when the ring was near collapse. The pressure was not increased further until the strain gage reading steadied. In the authors' opinion the point of collapse was accurate to $- 5$ psi. for the very non-circular rings. On the otherhand, failure of the more circular rings occurred rather quickly. During the latter portion of a test run it was customary to pause momentarily at each 5 psi. increment of pressure. Again the authors estimate that the point of collapse was accurate to $- 5$ psi.

The work was continued to be carried on in the laboratory of the
the laboratory was considerably reduced over most of the
pressure range. During the tests the indicated pressure
could be held fairly constant. An accuracy of plus or
minus one half a scale division ($\pm 1/2$ gal.) is possibly
conservative but will be assumed. The accuracy of the
recorded pressure at which strain readings were taken
then becomes $\pm 1/2$ gal.

With one additional consideration the above remarks
also apply to the accuracy of the recorded collapse
pressure. The collapse of rings with significant amounts
of out-of-roundness occurred rather slowly. It would seem
that the pressure could be inadvertently increased above
the true collapse pressure, and only slowly bring it
back to the collapse pressure. However, in these tests the strain
rate indicating the pressure which was reached at small
increments of pressure when the ring was first collapsed.
The pressure was not increased further until the strain
rate became excessive. In the laboratory of the
of collapse was assumed to be $\pm 1/2$ gal. for the first two
collapse rings. On the other hand, failure of the rings
collapse rings occurred rather slowly. During the latter
portion of a test run it was necessary to increase the pressure
at about 1 gal. increment of pressure. Above the collapse
pressure that the point of collapse was assumed to be $\pm 1/2$ gal.

Considering both the accuracy of the pressure reading and the point of failure, the recorded collapse pressure is believed to be within $-10\frac{1}{2}$ psi. or $+5\frac{1}{2}$ psi. of true.

The accuracy of the strain gages used, as stipulated by the manufacturer, was $\pm 2\%$. The transverse sensitivity correction was insignificant, being of the order of -0.05% . An accuracy of one half the smallest scale division on the strain indicator, ± 5 micro-inches per inch, is generally assigned to the switchbox-indicator combination.

As a practical expedient plots of out-of-roundness which are presented in the RESULTS were derived in the following manner; the average diameter was subtracted from the measured diameter at stations and half stations; the difference was divided by two and plotted at diametrically opposite stations. It is to be appreciated that this method will define a symmetrical out-of-roundness curve but is truly representative of the initial configuration only when a symmetrical two lobe pattern predominates.

Correlation of Data

The experimental data is compared to theoretical predictions based upon the following formula:

$$\sigma = \frac{P R}{h} - \frac{6 P R u_0 \cos 2\theta}{h^2 (1 - P/P_{crit.})} \quad (1)$$

Equation (1) was developed for the case of circular tubes in reference (4) but applies equally well to rings; the

Considering both the accuracy of the pressure reading and the point of failure, the recorded collapse pressure is believed to be within $\pm 10-15\%$ of true. The accuracy of the strain gages used, as estimated by the manufacturer, was $\pm 2\%$. The transverse sensitivity correction was insignificant, being of the order of -0.05% . An accuracy of one half the smallest scale division on the strain indicator, ± 5 micro-inches per inch, is generally assigned to the micro-indicator combination. As a practical expedient plots of out-of-roundness which are presented in the RESULTS were derived in the following manner; the average diameter was subtracted from the measured diameter at stations and half stations; the difference was divided by two and plotted at diametrically opposite stations. It is to be appreciated that this method will define a symmetrical out-of-roundness curve but is only representative of the initial modification only when a symmetrical two lobe pattern predominates.

Correlation of Data

The experimental data is compared to theoretical predictions based upon the following formula:

$$(1) \quad \sigma = \frac{P}{A} - \frac{\delta P R}{A^2} \cos 2\theta$$

Equation (1) was developed for the case of internal pressure in a cylinder (4) but applies equally well to tension; the

assumptions made in the derivation may be found in the same reference. The major qualifications are that the ring be thin and that the initial configuration be given by $(R + u_0 \cos 2 \theta)$.

For purposes of computation the authors found it convenient to rearrange and modify equation (1)

$$\sigma = \frac{PR_0}{h} + \frac{6 P R u_0 h E \cos 2 \theta}{Eh^3 - 4PR^3} \quad (2)$$

$$\epsilon = \frac{PR_0}{Eh} + \frac{6 P R u_0 h \cos 2 \theta}{Eh^3 - 4PR^3} \quad (3)$$

Equation (2) was derived directly from equation (1) by the substitution of $3 Eh^3/12$ for $P_{crit.}$, R_0 for R , and rearranging. Equation (3) is obtained by dividing equation (2) by E . The use of the outside radius, R_0 , instead of the radius to the neutral axis, R , in the hoop stress term seemed well taken but changed the predicted strains an insignificant amount.

A common criteria for failure is yielding of the outer fibers due to the addition of hoop stress and the maximum bending stress. Then as a basis for comparison of experimental and theoretical collapse pressures in those cases where the out-of-roundness pattern is $u_0 \cos 2 \theta$, equation (2) can be reduced to

$$\sigma_{max.} = \frac{FR_0}{h} + \frac{6 PR u_0 h E}{Eh^3 - 4PR^3} \quad (4)$$

assumptions made in the derivation may be found in the same reference. The major difficulties are that the ring is thin and that the initial configuration be given by $(R + u_0 \cos \theta)$.

For purposes of comparison the results found in equation (1) are written as follows:

$$\begin{aligned} (1) \quad \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) + \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) &= \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) \\ (2) \quad \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) + \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) &= \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) \\ (3) \quad \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) + \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) &= \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) \end{aligned}$$

Equation (3) was derived directly from equation (1) by the substitution of $\frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right)$ for R , and rearranging. Equation (3) is obtained by dividing equation (2) by R . The use of the outside radius, R_0 , instead of the radius to the neutral axis, R , in the hoop stress term seemed well taken but changed the predicted stresses in a significant amount.

A common criterion for failure is failure of the outer fibers due to the addition of hoop stress and the bending stress. Then as a basis for comparison of experimental and theoretical collapse stresses in these cases where the only difference is the use of R or R_0 , equation

$$(4) \quad \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) + \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right) = \frac{1}{R} \frac{d}{d\theta} \left(\frac{R^2}{\sin \theta} \frac{d\theta}{d\theta} \right)$$

When $\sigma_{\max.}$ is assumed to be equal to σ_y , the pressure as determined by equation (4) is then the theoretical collapse pressure.

When D. is returned to the school for the purpose

of determining its position (4) it is then for educational

colleges transfer.

III

RESULTS

All results presented in this section were derived from data obtained by the authors.

Figure III is a plot of the data obtained during the compression tests of the aluminum alloy specimens and from which the value of Young's Modulus and the yield stress were determined.

Figure IV shows the results of the preliminary test which was made to determine the effect of various lubricants and gaskets upon strain. Lubricant #1 was black rubber-to-metal cement; Lubricant #2 was clear rubber cement. Gasket #1 had been used during the Proof Testing; Gasket #2 was of the same dimensions but had not been used previously.

Figures V, VIII, XI, XIV, and XVII show measured and theoretical circumferential strain distributions for one or more pressures. Theoretical curves are based upon equation (3).

Figures VI, IX, XII, XV, and XVIII are plots of out-of-roundness versus circumferential positions.

Figures VII, X, XIII, XVI, and XIX are plots of ring thickness versus circumferential position.

Figures XX-XXVI are a comparison of measured and predicted strains. The experimental points are an average of

Results

All results presented in this section were tabulated from data obtained by the authors.

Figure III is a plot of the data obtained during the recognition tests of the aluminum ship movement and from which the values of Young's modulus and the yield stress were determined.

Figure IV shows the results of the preliminary test which was made to determine the effect of various indentations and depths upon strain. Indentation #1 was blank rubber-to-metal contact; Indentation #2 was clean rubber contact. Indentation #3 was used during the first testing; Indentation #4 was of the same dimensions but had not been used previously.

Figures V, VII, IX, XIV, and XVII are arranged and interpreted chronologically as the difference between one or both pressures. Theoretical curves are based upon equation (2).

Figures VI, IX, XII, XV, and XVIII are plots of

of the difference between experimental positions.

Figures XI, X, XIII, XVI, and XIX are plots of the

theoretical curves corresponding to the

Figures XX-XXIV are a comparison of measured and theoretical values. The experimental values are shown in

strains read at the 0° and 180° positions, and 90° and 270° positions. The theoretical curves are based upon equation (3).

Figure XXVII shows experimental collapse points plotted on arguments of pressure and the ratio of out-of-roundness to average thickness.

Figure XXVIII~~X~~ is a comparison of experimental collapse pressures with several theoretical predictions. All theoretical curves are based upon equation (4). They differ in that Curve A is plotted for $\sigma_{\max} = \sigma_y = 35,000$ psi. and $h = h_{\text{ave}} = 0.250"$; Curve B is plotted for $\sigma_{\max} = 41,000$ psi. and $h = h_{\text{ave}} = 0.250"$; Curve C is plotted for $\sigma_{\max} = 41,000$ psi. and $h = h_{\text{min}} = 0.243"$.

average value of σ_y and σ_z positions, and σ_{yz} and σ_{zx} positions. The theoretical curves are based upon equation (3).

Figure XVII shows experimental collapse curves plotted on graphs of σ_y versus σ_z and the ratio of out-of-balance to average collapse.

Figure XVIII is a comparison of experimental collapse pressures with several theoretical predictions. All

theoretical curves are based upon equation (4). They differ

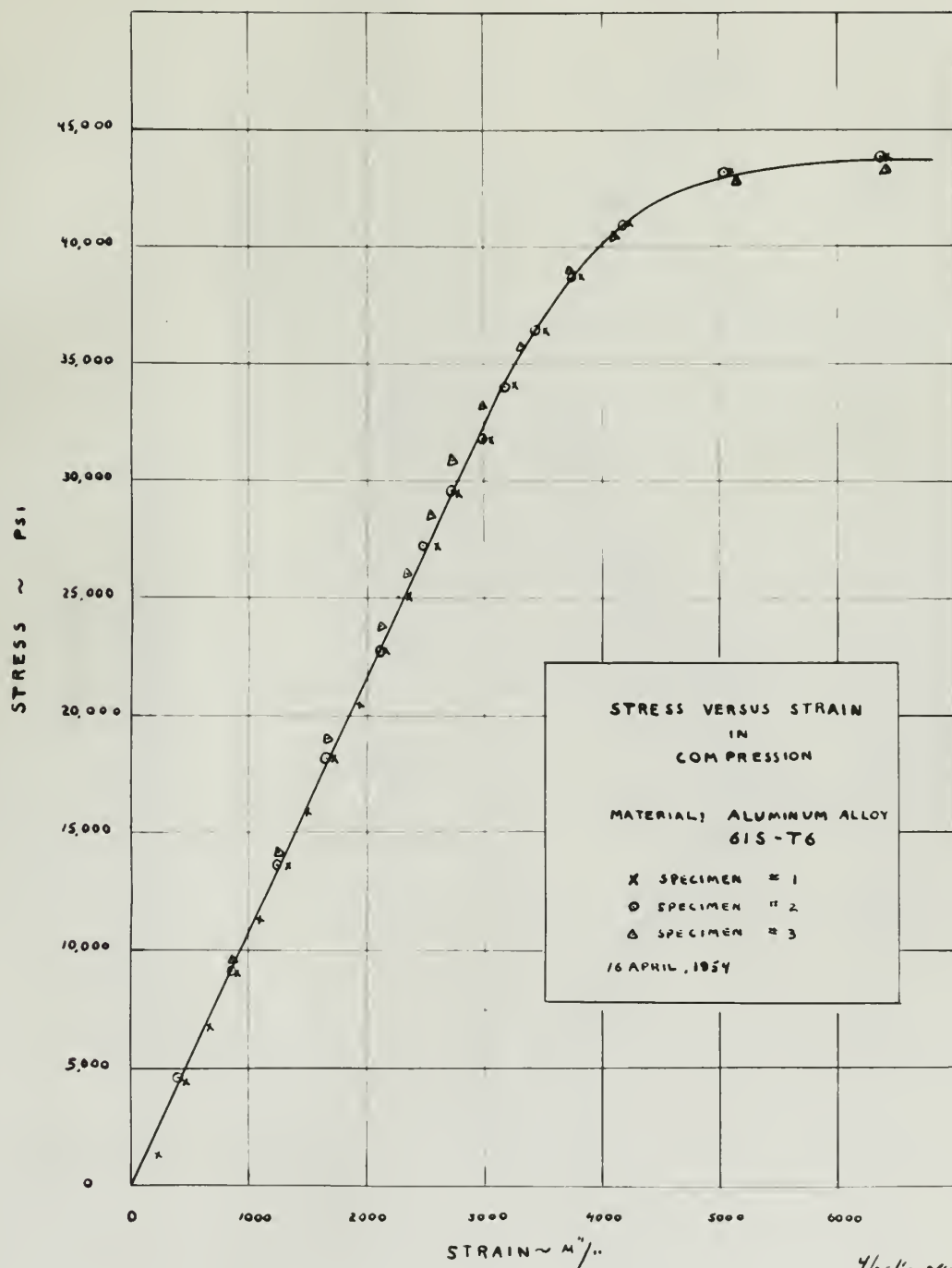
in that Curve A is plotted for $\sigma_y = 35,000$ psi.

and $\sigma_z = 0.350$; Curve B is plotted for

$\sigma_y = 41,000$ psi. and $\sigma_z = 0.350$; Curve C is

plotted for $\sigma_y = 41,000$ psi. and $\sigma_z = 0.250$.

FIGURE III
STRESS STRAIN CURVE



4/25/54 YHD -WJ 7

Figure IV

CIRCUMFERENTIAL STRAIN DISTRIBUTION FOR DIFFERENT LUBRICANTS AND GASKETS
RING NO.-PRELIMINARY TEST

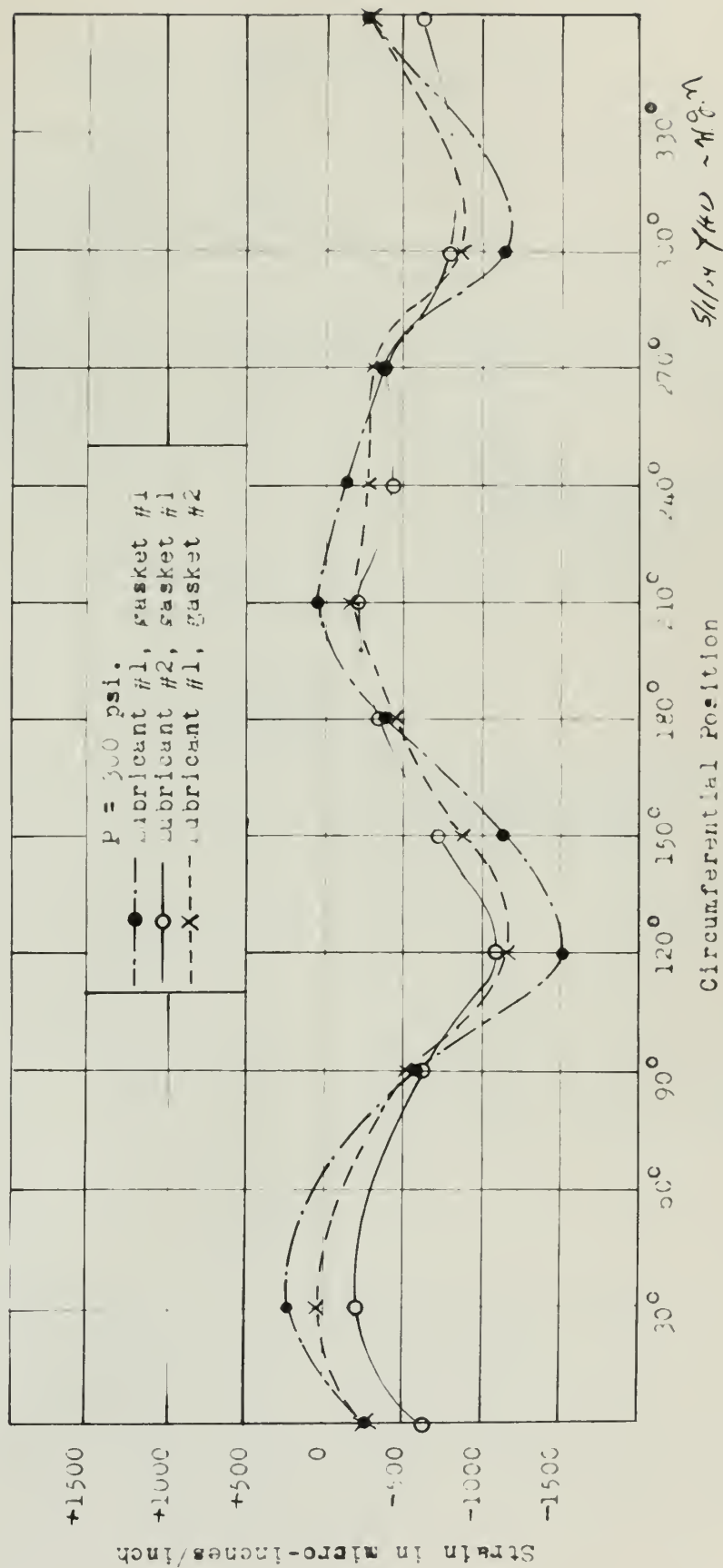


Figure V
CIRCUMFERENTIAL STRAIN DISTRIBUTION ON INNER SURFACE
RING NO. 2 $u_0 = 0.0118''$

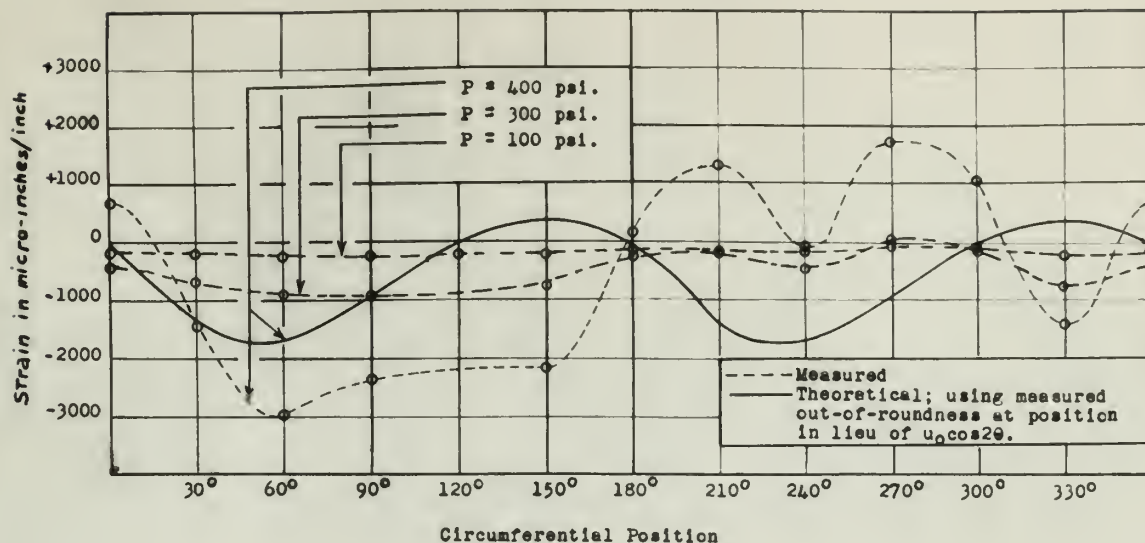


Figure VI
INITIAL CONFIGURATION
RING NO. 2

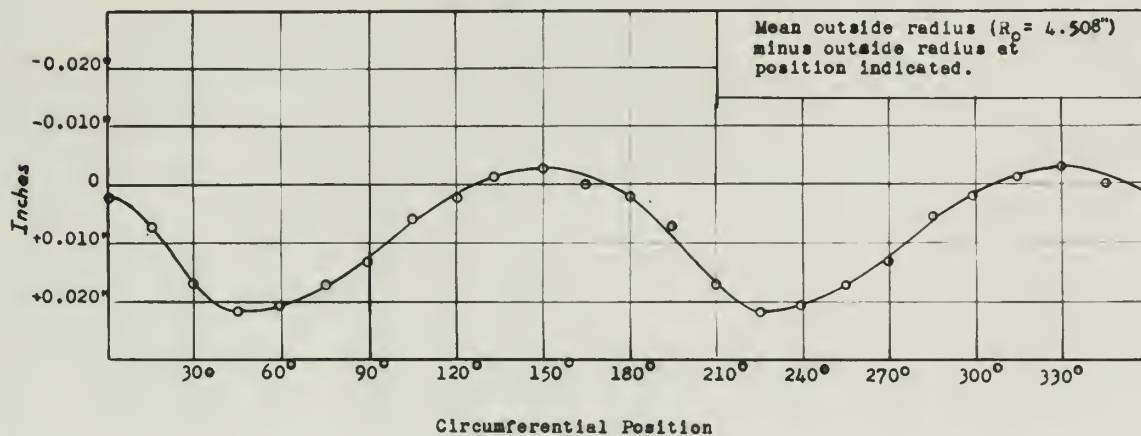


Figure VII
VARIATION IN THICKNESS
RING NO. 2

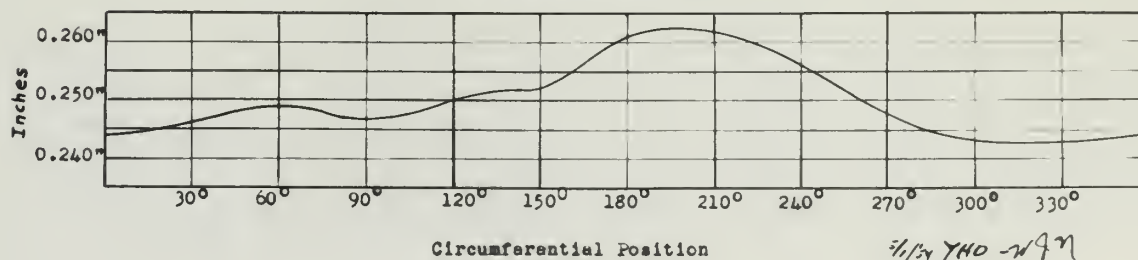


Figure VIII

CIRCUMFERENTIAL STRAIN DISTRIBUTION ON INNER SURFACE
RING NO. 3 $u_0 = 0.0268''$

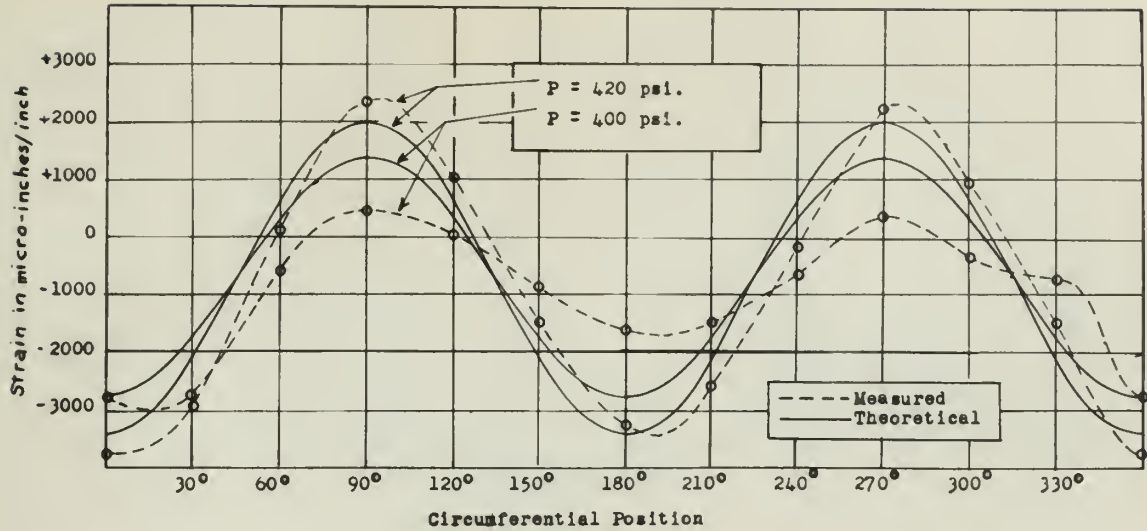


Figure IX

INITIAL CONFIGURATION
RING NO. 3

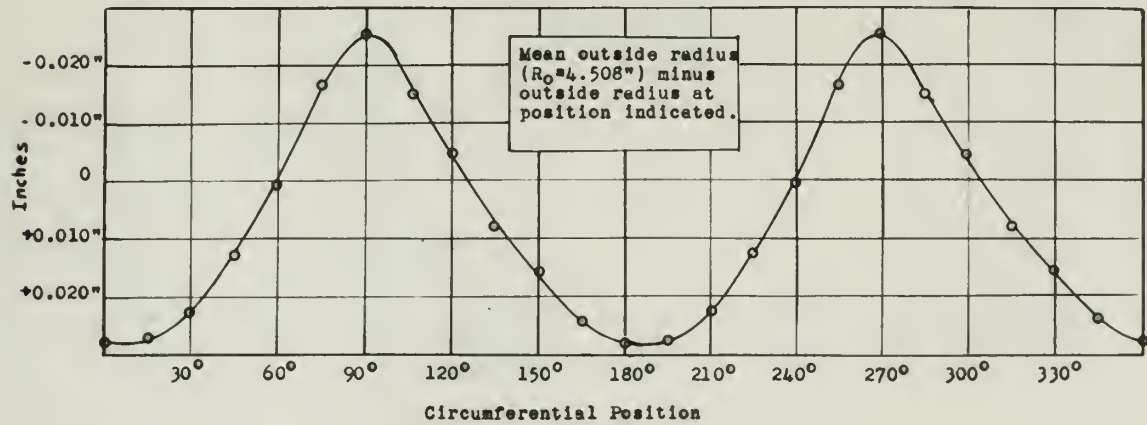


Figure X

VARIATION IN THICKNESS
RING NO. 3

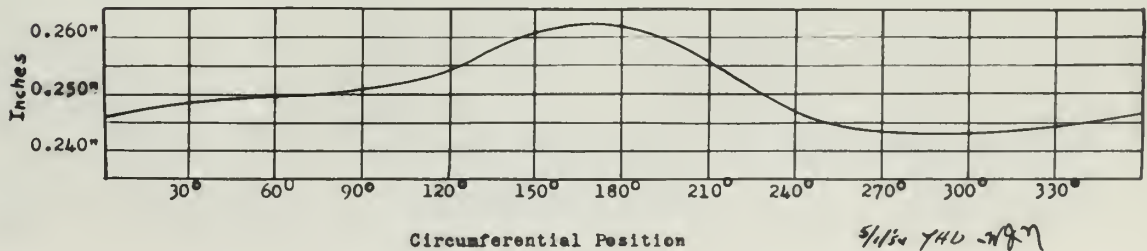


Figure **XI**

CIRCUMFERENTIAL STRAIN DISTRIBUTION ON INNER SURFACE
RING NO. 5 $u_0 = 0.0705"$

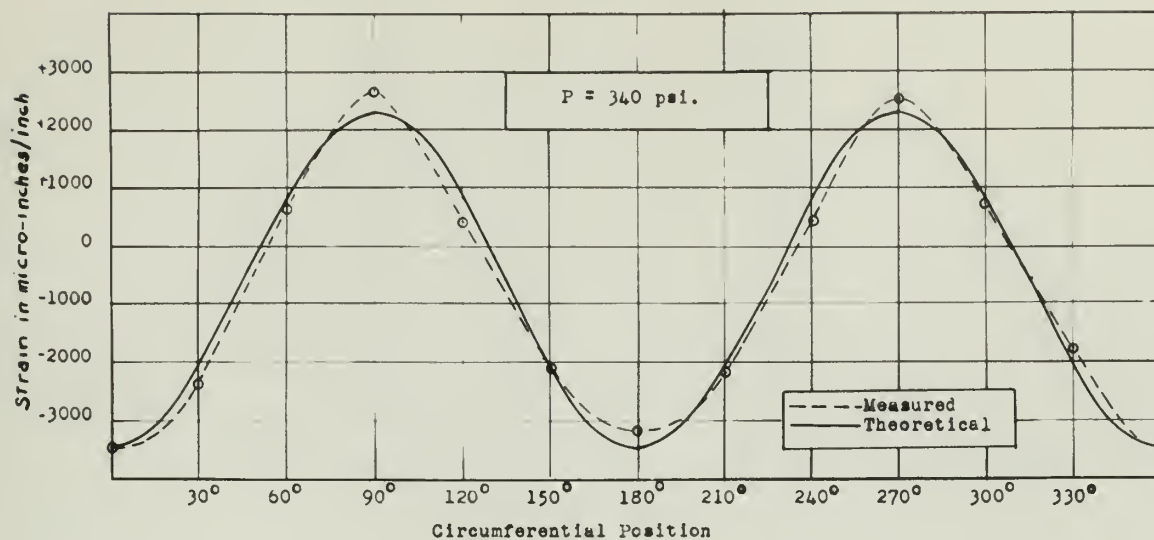


Figure **XII**

INITIAL CONFIGURATION
RING NO. 5

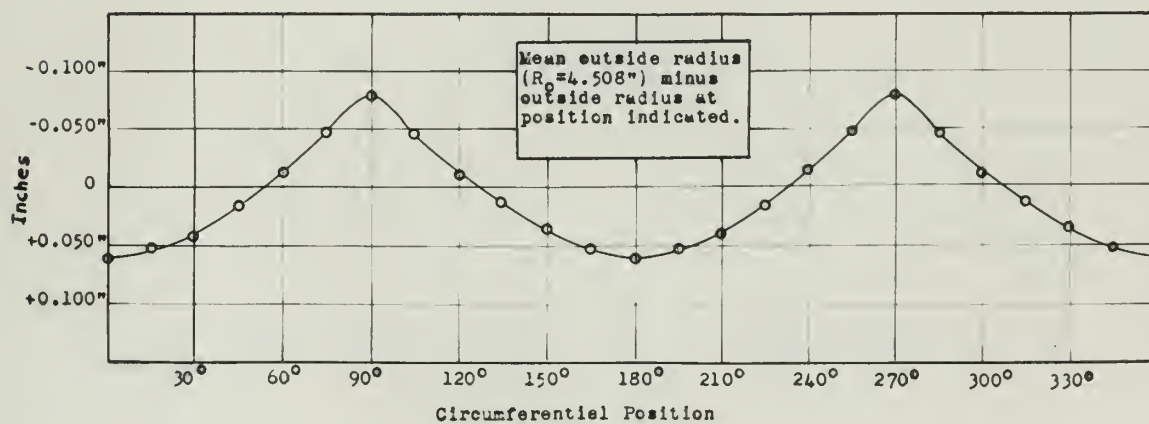


Figure **XIII**

VARIATION IN THICKNESS
RING NO. 5

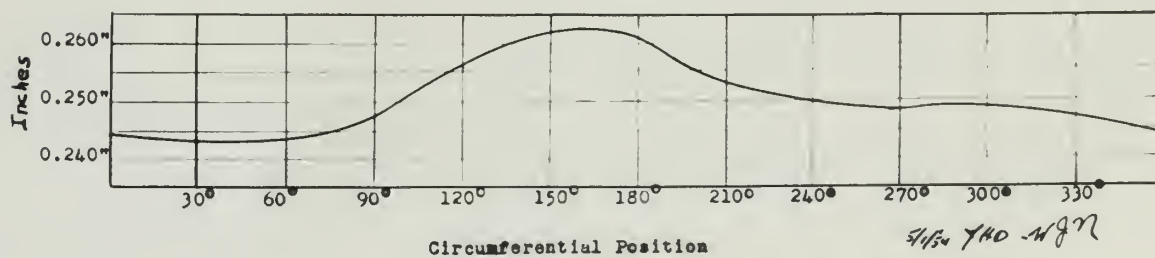


Figure XIV

CIRCUMFERENTIAL STRAIN DISTRIBUTION ON INNER SURFACE
RING NO. 6 $u_0 = 0.075"$

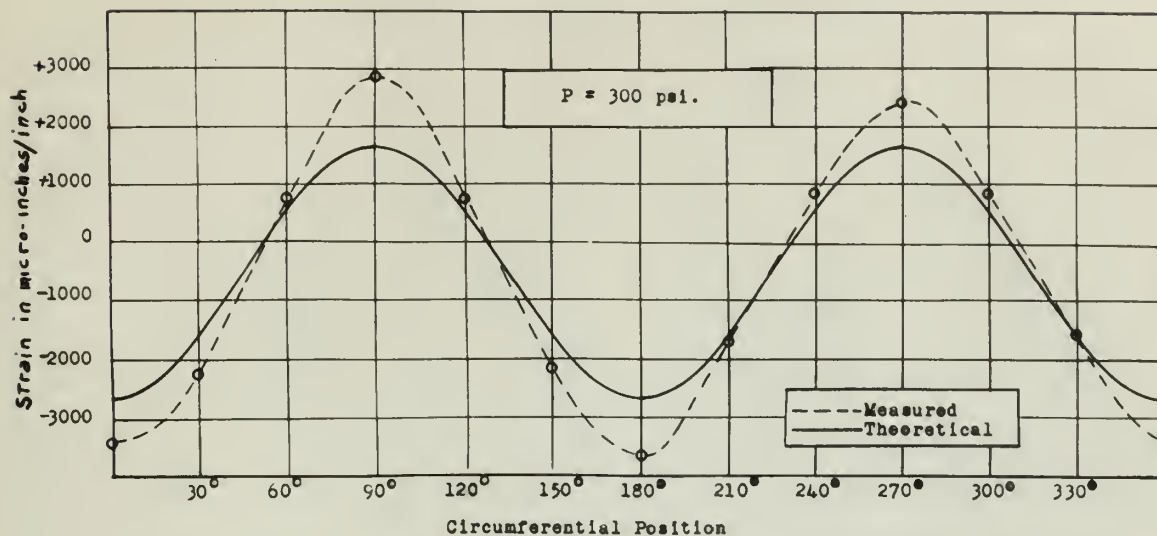


Figure XV

INITIAL CONFIGURATION
RING NO. 6

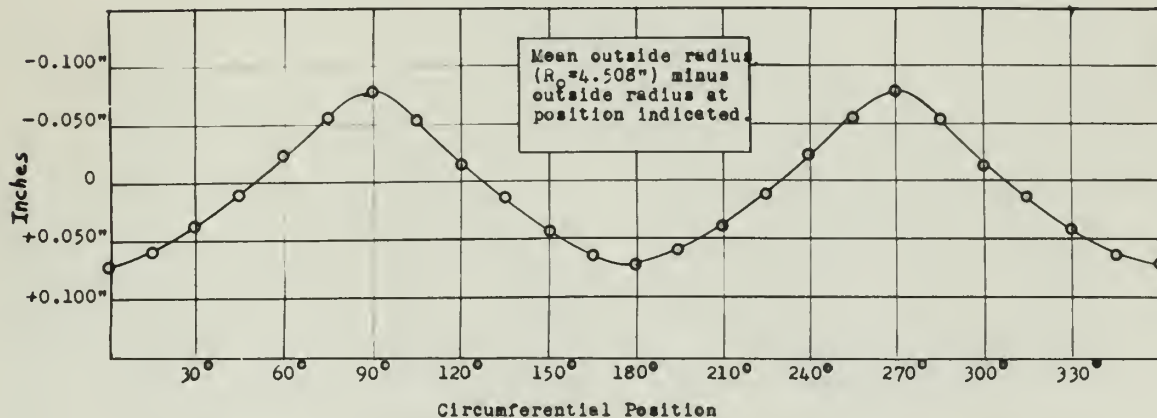


Figure XVI

VARIATION IN THICKNESS
RING NO. 6

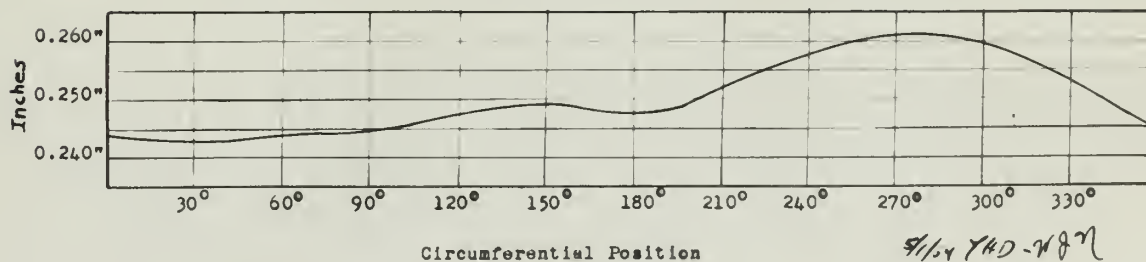




Figure XVII

CIRCUMFERENTIAL STRAIN DISTRIBUTION ON INNER SURFACE
RING NO. 7 $u_0 = 0.1535"$

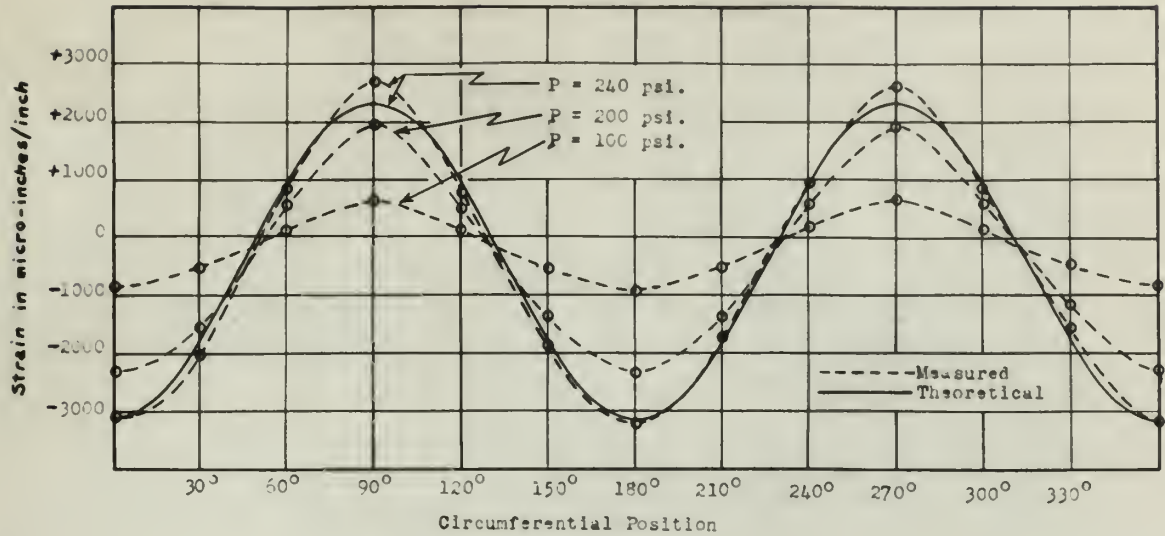


Figure XVIII

INITIAL CONFIGURATION
RING NO. 7

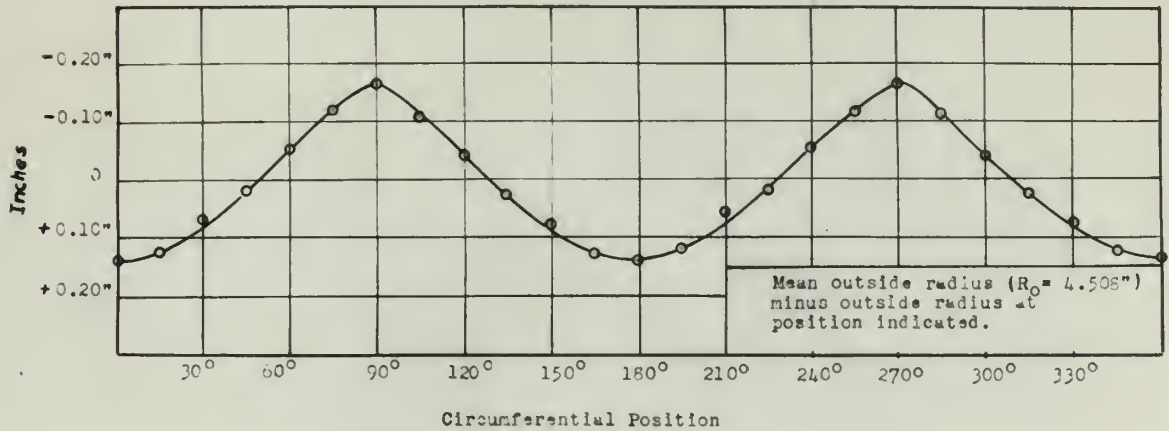
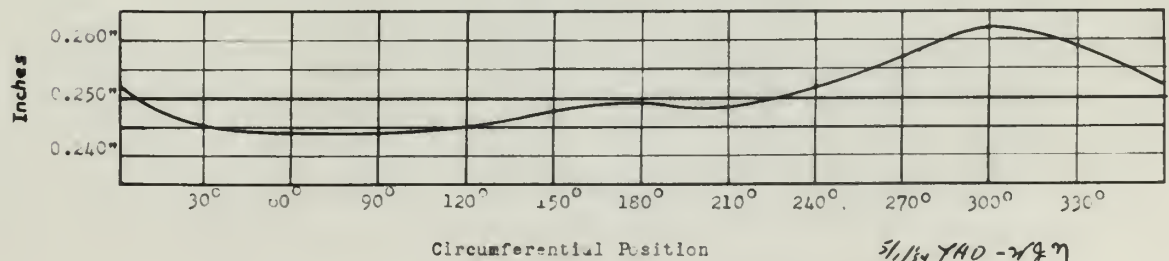


Figure XIX

VARIATION IN THICKNESS
RING NO. 7



5/1/54 YAO - 2/87

Figure XX

COMPARISON OF MEASURED AND THEORETICAL STRAINS

RING NO. 3 $u_0 = 0.027"$

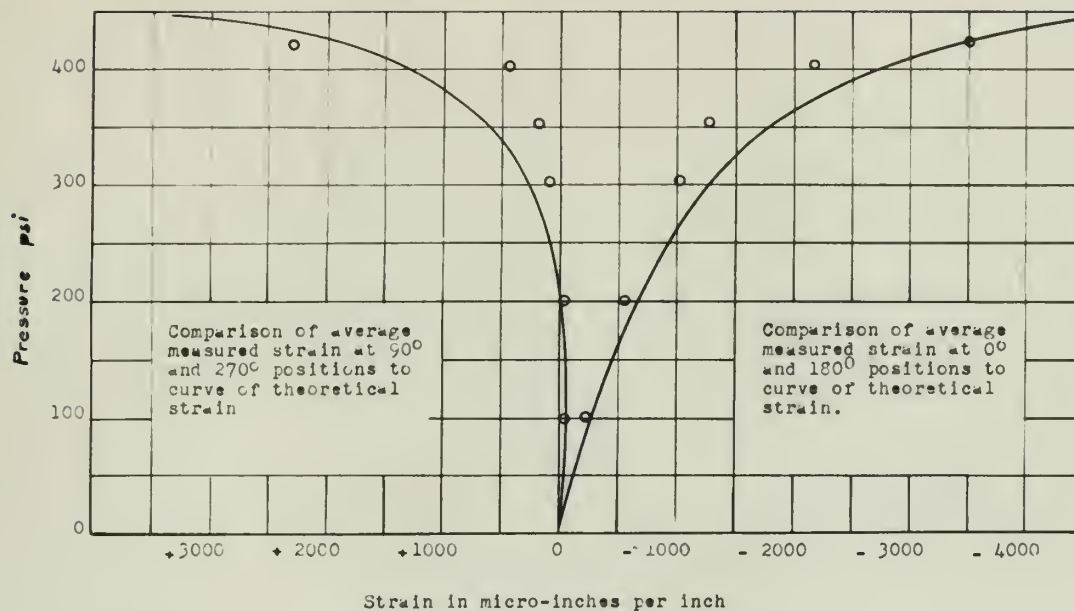


Figure XXI

COMPARISON OF MEASURED AND THEORETICAL STRAINS

RING NO. 4 $u_0 = 0.048"$

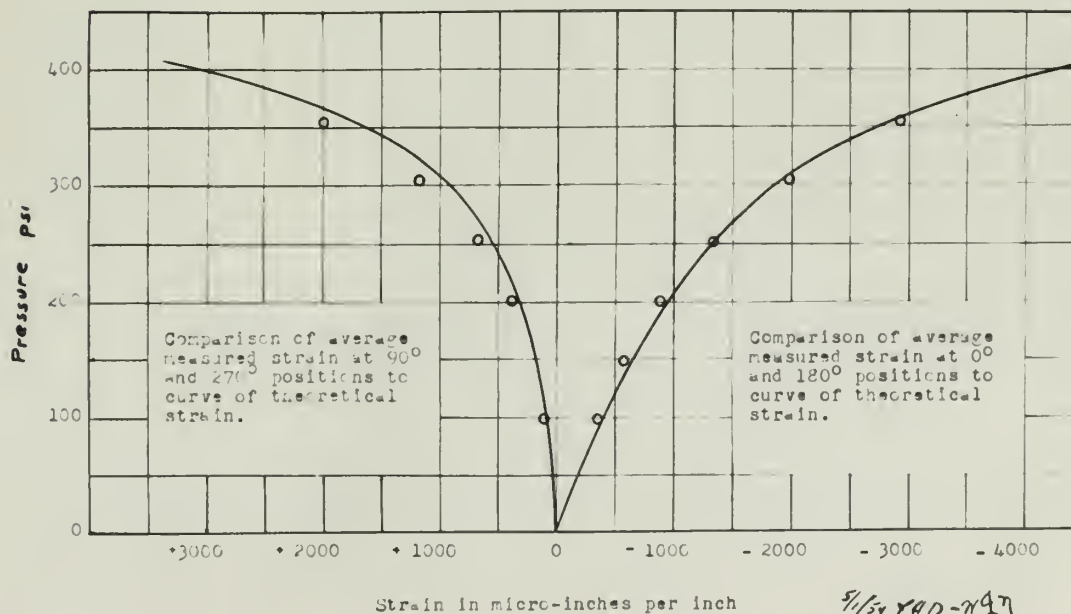


Figure XXII

COMPARISON OF MEASURED AND THEORETICAL STRAINS
RING NO. 5 $u_0 = 0.070"$

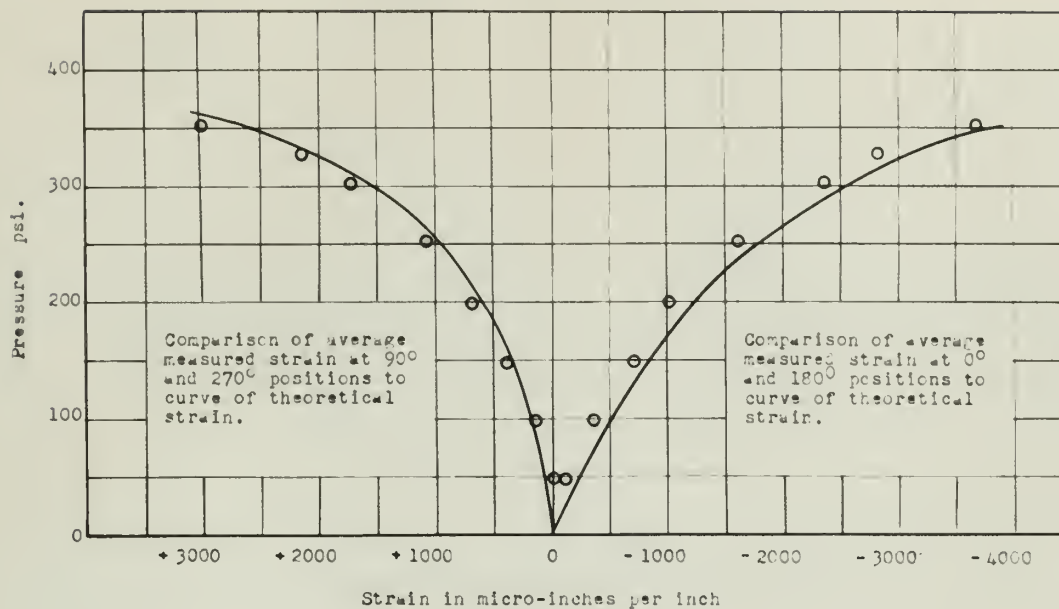
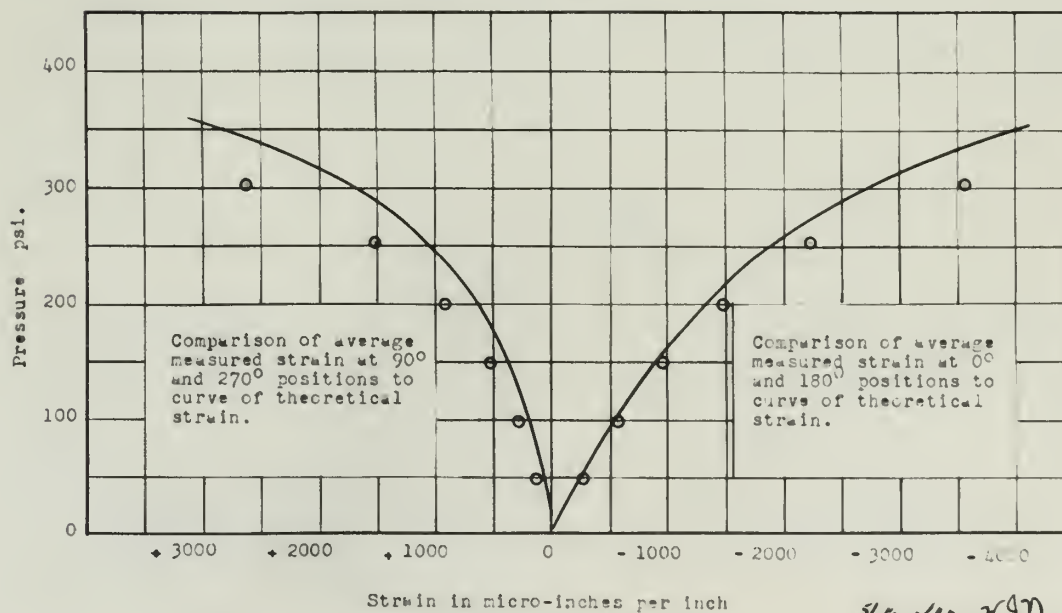


Figure XXIII

COMPARISON OF MEASURED AND THEORETICAL STRAINS
RING NO. 6 $u_0 = 0.075"$



5/15/40 - Wgn

Figure XXIV

COMPARISON OF MEASURED AND THEORETICAL STRAINS
RING NO. 7 $u_0 = 0.1635"$

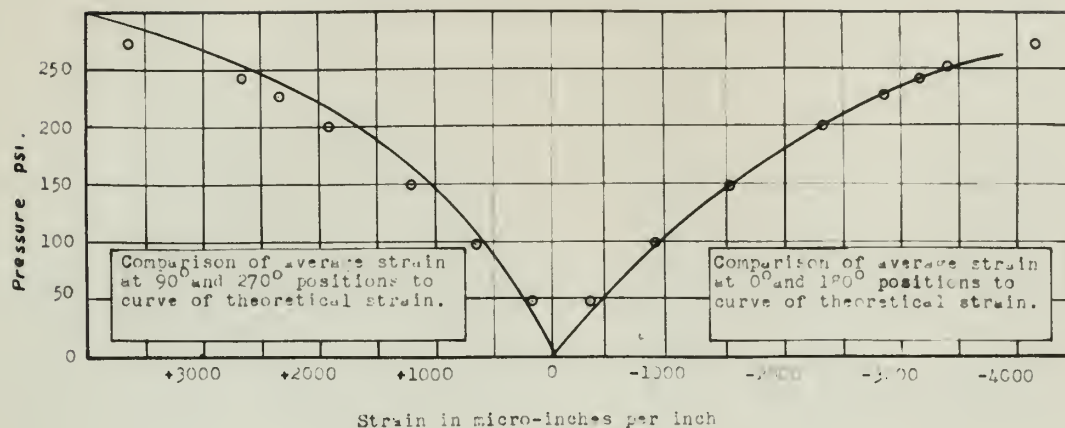


Figure XXV

COMPARISON OF MEASURED AND THEORETICAL STRAINS
RING NO. 8 $u_0 = 0.204"$

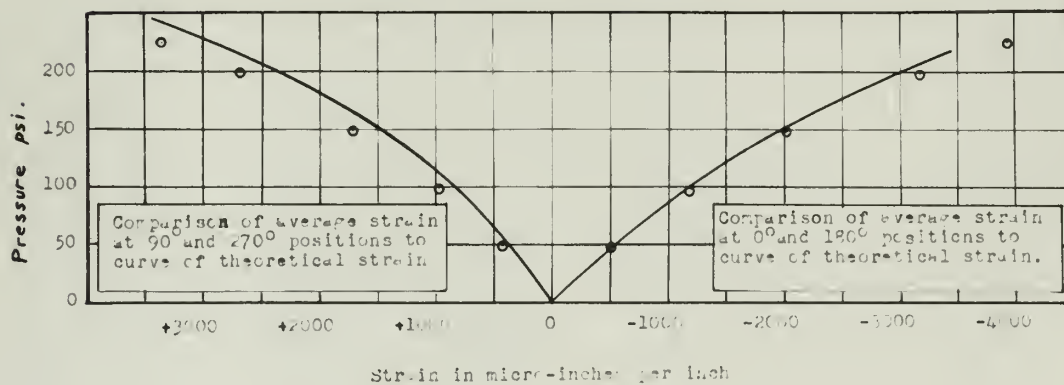
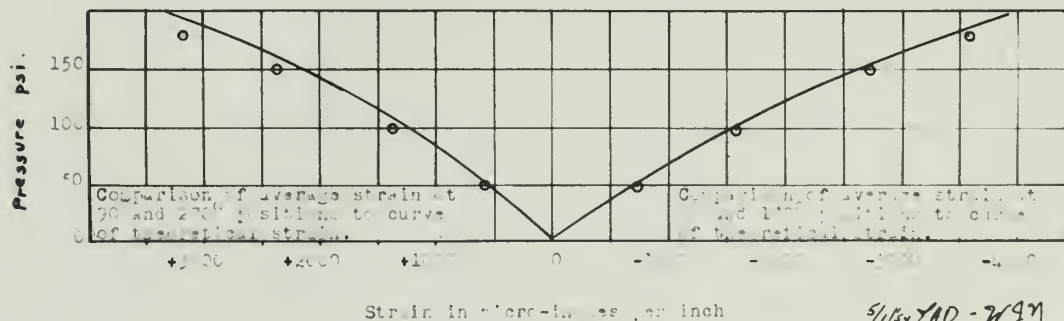


Figure XXVI

COMPARISON OF MEASURED AND THEORETICAL STRAINS
RING NO. 9 $u_0 = 0.223"$



5/1/54 YAD - W97



Figure XXVII

COMPARISON OF MEASURED AND PREDICTED BUCKLING PRESSURES

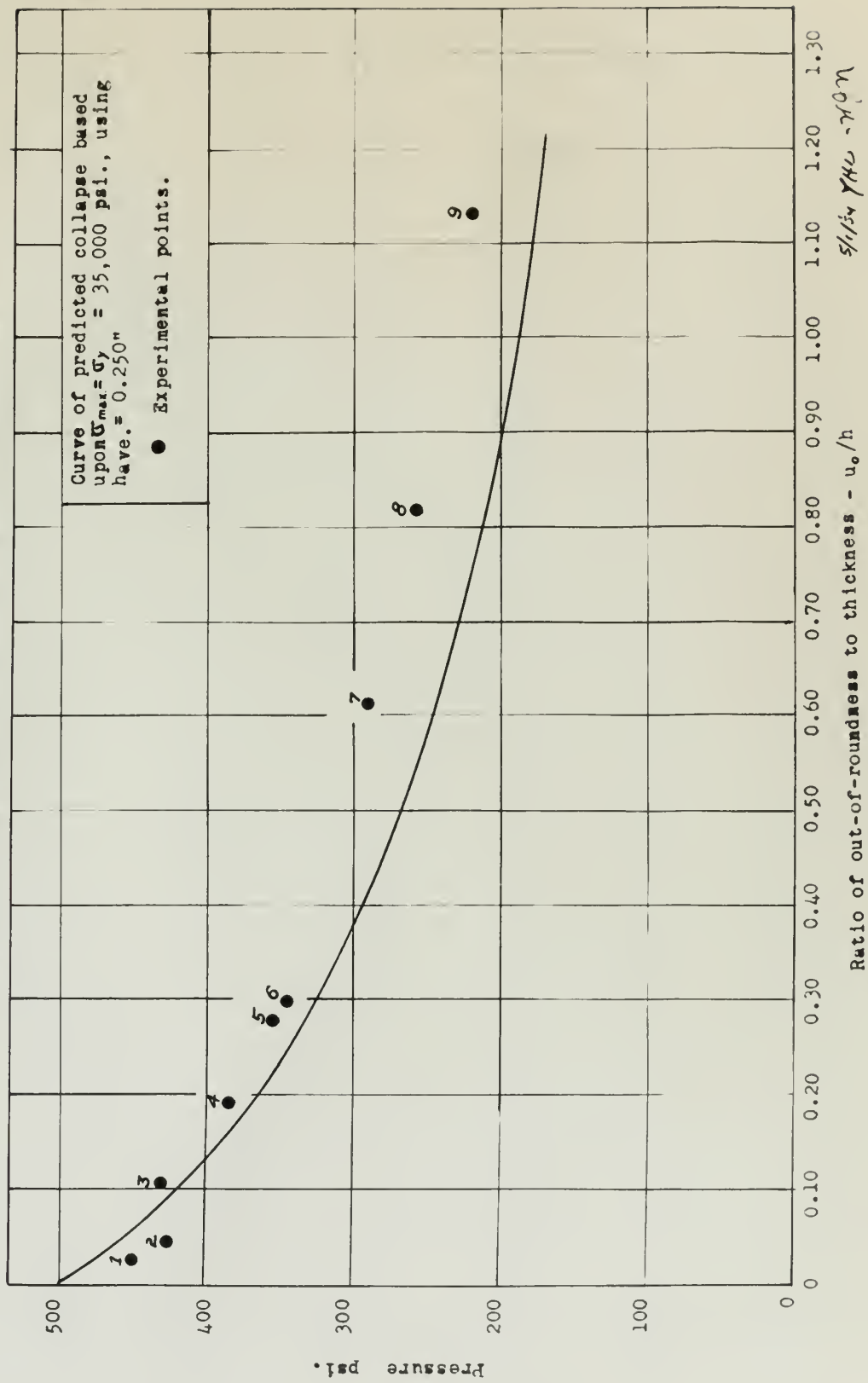
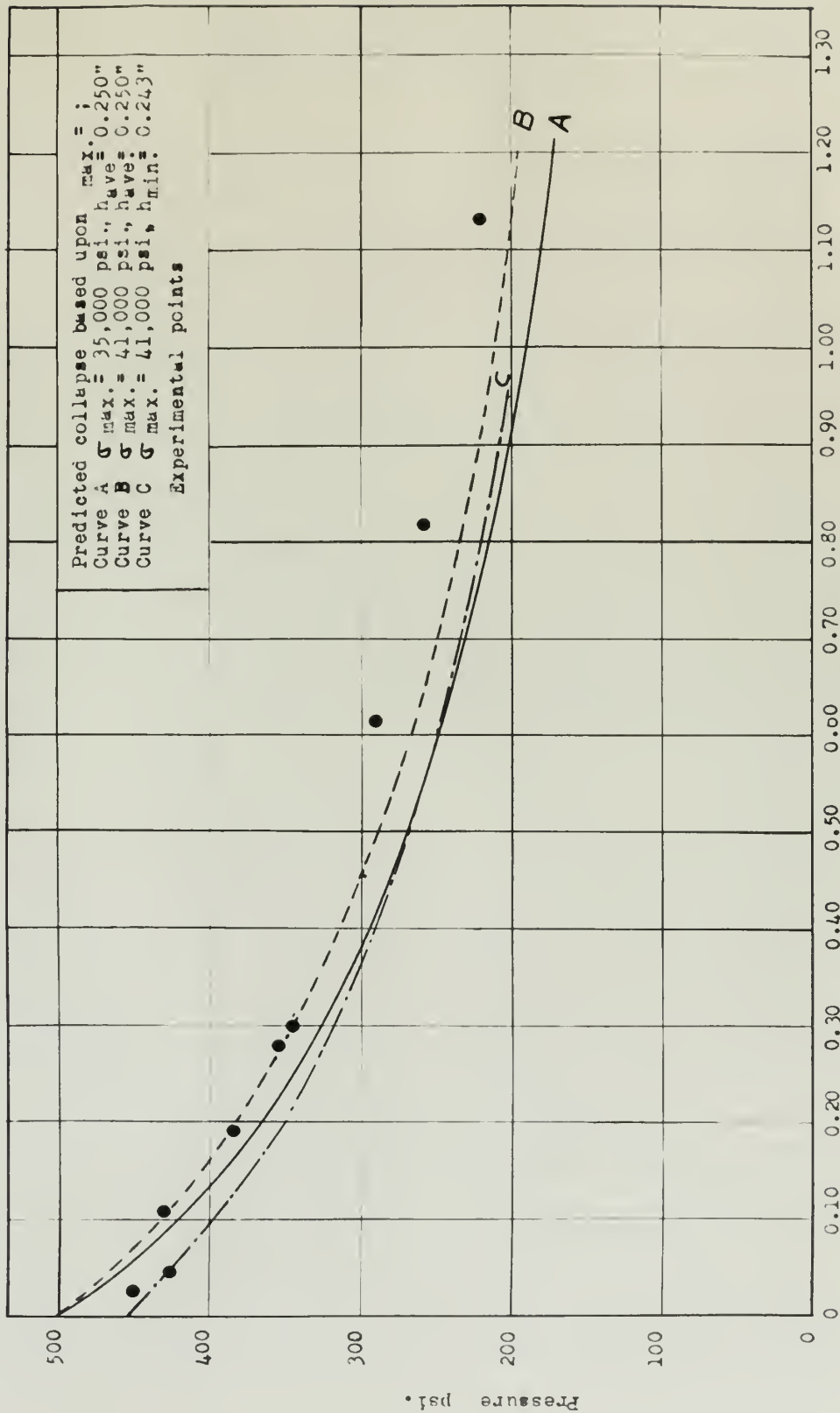


Figure XXVIII

COMPARISON OF MEASURED AND PREDICTED BUCKLING PRESSURES



Ratio of out-of-roundness to average thickness u_o/h_{ave} .

5/1/54 HAD-497



IV.

DISCUSSION OF RESULTS

Introduction

For convenience and continuity, the experimental results may be grouped in the following sequence:

1. Results of Compression Tests
2. Proof Test Data
3. Circumferential Strain Distributions
4. Maximum Strains
5. Collapse Pressures

This classification is convenient in that the significance of any group depends to some extent upon the interpretation and validity attached to one or more of the preceding groups.

Results of Compression Tests

The results obtained from the compression tests were probably the most significant factors in the correlation of experimental and theoretically predicted quantities.

The data obtained from the tests of the specimens were surprisingly consistent. See Figure III. Values of strain correlate particularly well at high stresses where some deviation might be expected. The possibility of introducing rather significant errors during the resetting of the Huggenberger Tensometers was considered,

DISCUSSION OF RESULTS

Introduction

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Results of Compression Tests

The results obtained from the compression tests were probably the most significant factors in the correlation of experimental and theoretically predicted quantities. The data obtained from the tests of the specimens were surprisingly consistent. See Figure III. Values of strain correlate particularly well with other factors where some deviation might be expected. The possibility of introducing further significant errors during the testing of the specimens was considered.

but a review of the data indicated that increments of strain before and subsequent to a reset were consistent. That the stress-strain curve shown in Figure III is representative of the alloy from which the rings were manufactured is implied in preceding sections. Appreciating the fact that the proof stress and ultimate strength of aluminum alloy varies considerably with the details of the heat treatment and possibly the original lot number, the authors specified that the specimens be obtained from the length of tubing from which the rings were manufactured and that all rings and specimens be heat treated as a group.

Young's Modulus, as determined from the stress-strain curve, was 10.8×10^6 psi. and can be compared to $E = 10 \times 10^6$ psi., the value generally given in handbooks. In the authors' opinion, Figure III justified the assumption of a linear stress-strain relationship up to a stress of 34,500 psi. and a 0.2% proof stress, designated yield stress, of 35,000 psi. Furthermore, it is to be noted that a very distinct departure from linearity did not occur until a stress of about 41,000 psi. was obtained.

These properties were commensurate with those characteristics considered most desirable in the material from which the rings were made. The existence of a linear stress-strain relationship over a large range of stress

but a review of the data indicated that independent of strain
 below and subsequent to a test were considered. That the
 stress-strain curve shown in Figure 11 is representative
 of the alloy from which the rings were manufactured is
 implied in preceding paragraphs. Appreciating the fact that
 the proof stress and ultimate strength of aluminum alloy
 varies considerably with the details of the heat treatment
 and possibly the original lot number, the authors specified
 that the specimens be obtained from the length of tubing
 from which the rings were manufactured and that all rings
 and specimens be heat treated as a group.

Young's modulus, as determined from the stress-strain
 curve, was 10.5×10^6 psi. and can be compared to
 9×10^6 psi., the value generally given in handbooks.
 In the authors' opinion, Figure 11 justifies the assumption
 of a linear stress-strain relationship up to a stress of
 24,000 psi. and a 0.2% proof stress, designated yield stress,
 of 32,000 psi. Furthermore, it is to be noted that a very
 distinct departure from linearity did not occur until a
 stress of about 41,000 psi. was obtained.

These properties were determined with single
 compressive specimens machined from bar stock in the material
 from which the rings were made. The diameter of a linear
 stress-strain relationship over a large range of stress

permitted most of the experimental strains to be compared directly with predictions based upon equation (3) without the complication of a decreasing modulus. For extrapolation of the results to steel rings, a plateau corresponding to a true yield stress was desirable. Although the authors did not anticipate such similarity at the time aluminum alloy was selected, the upper portion of the stress-strain curve is in most respects analogous to the compression test curves obtained for mild steel specimens tested by M. L. Pittman, Jr., and V. W. Rinehart (7).

It may be argued that the remainder of the results would have been of no consequence without the stress-strain data. The use of a standard stress-strain curve for aluminum alloy 61S-T6 would have involved an 8% error in Young's Modulus and a 14% error in the proof stress. Furthermore, the shape of the curve above the proof stress is valuable when analyzing the failure of a ring.

Proof Test Data

The authors have stated that the measured strains at a specified pressure were dependent to some extent upon the gasket and the type of lubricant used between the gasket and the Plexiglas surfaces. Figure IV shows this effect quantitatively. Black rubber-to-metal cement has been designated Lubricant #1 and clear rubber cement as

permitted heat of the experimental device to be compared directly with predictions based upon equation (3) without the complication of a decreasing modulus. The extrapolation of the results to steel rings, a plastic corresponding to a true yield stress was desirable. Although the authors did not anticipate such similarity at the time aluminum alloy was selected, the upper portion of the stress-strain curve is at least restricted analogous to the compression test curves obtained for mild steel specimens tested by R. L. Fittman, Jr., and V. W. Kinsman (7).

It may be argued that the magnitude of the results would have been of no consequence without the stress-strain data. The use of a standard stress-strain curve for aluminum alloy 618-T0 would have involved an 8% error in Young's modulus and a 14% error in the proof stress. Furthermore, the shape of the curve above the proof stress is variable when analyzing the failure of a ring.

Proof Test Data

The authors have stated that the measured strains of a specified pressure were dependent to some extent upon the method and the type of loadings used between the gauges and the elastic portions. Figure 17 shows this effect qualitatively. Black rubber-to-metal bonds have been designated maximum ϵ_1 and clear rubber bonds as

Lubricant #2. Although Lubricant #1 was considered most suitable on the basis of a preliminary investigation which has been described, Lubricant #2 appeared to be a reasonable second choice. The two gaskets were of the same dimensions and material but differed in that Gasket #1 had been used several times before while Gasket #2 had not. Relative to Gasket #2, Gasket #1 was very limber and appeared to have a slick, oily surface.

As shown by Figure IV the largest measured strains were attained with Lubricant #1 and Gasket #1. Lubricant #1, Gasket #2 were next in magnitude of strains while Lubricant #2, Gasket #1 gave the smallest measured strains. Considered as percentages, the differences in strains are quite significant. The authors reasoned that the gasket and lubricant were able to affect the strains by changing the magnitude of the friction force acting at the gasket-Flexiglas interface.

On the basis of the results indicated by Figure IV, the authors elected to use a well broken-in gasket in conjunction with Lubricant #1 for the remainder of the tests since this combination more nearly satisfied the requirement that the ring be completely free of restraint at the edges.

A thorough investigation of the friction force at the

On the basis of the results indicated by Figure 11, the authors selected for use a well-taken-in-hand specimen with dimensions of 10 mm by 10 mm by 10 mm. This specimen was used in the experiments described in this paper. The authors also noted that the specimen was well-taken-in-hand and was used in the experiments described in this paper.

gasket-Plexiglas interface should be a prerequisite for further experimentation based upon this thesis. The objectives of such an investigation should be a consideration of all likely gaskets and lubricants, a comparison of possible combinations with respect to effect upon strains, and a quantitative evaluation of the coefficient of friction characteristic of given combinations of gasket, lubricant and pressure.

Circumferential Strain Distribution

Figures VI, IX, XII, XV, and XVIII show typical out-of-roundness curves. Of the rings shown in this group, all were deformed intentionally with the exception of Ring No. 2, Figure VI. It is significant that the out-of-roundness obtained in Ring No. 2, while purely arbitrary, is very similar to a configuration given by $u_0 \cos 2 \theta$. The remainder of the plots are characterized by rather steep peaks in the vicinity of the minimum diameter and shallow peaks at points of maximum diameter. It is to be noted that the steep peaks occurred at points where the deforming loads were applied by the loading machine. The configurations are, however, reasonably similar to a cosine curve of the form $u_0 \cos 2 \theta$.

Figures VII, X, XIII, XVI, and XIX show the variation in thickness versus circumferential position. It is apparent that the relation of thickness and out-of-roundness at a

These findings suggest that the mechanism of action of the various experimental bases upon which the various observations of such an investigation would be a comparison of the results of all likely cases and instances, a comparison of results obtained with respect to effect upon the various and a qualitative evaluation of the conditions of reaction characteristics of these conditions of effect, however, and possibly.

Generalized Study Discussion

Figure VI, II, III, IV, and V show typical curves of various curves. Of the curves shown in this group, all were obtained independently of the reaction of the No. 2, Figure VI. It is significant that the out-of-phase curves obtained in this No. 2, while partly arbitrary, is very similar to a combination given by the No. 2. The reaction of the phase are characterized by rather sharp peaks in the vicinity of the minimum diameter and reaction peaks at points of maximum diameter. It is to be noted that the latter peaks occurred at points where the following points were obtained by the loading reaction. The observations are, however, reasonably similar to a series curve of the type of the No. 2. Figure VII, VIII, and IX show the reaction in the case of various different positions. It is to be noted that the reaction is relatively and independently of the

particular station is completely arbitrary. In general, the difference between minimum and maximum thickness is of the order of 0.02". Since the out-of-round plot is derived from the diameters read to the outer circumference, the corresponding configuration of the neutral axis may be assumed to deviate slightly from that of the plot. Since sections of thickness greater than average were generally opposite sections of less than average, the effect of this deviation upon the measured value of u_0 is assumed to be of no consequence except in the circular rings. During a preliminary reduction of data the authors considered the effect which a variation in thickness might have upon the plots of out-of-roundness and predicted strains. The plot of out-of-roundness for Ring No. 2 was altered to a minor extent. The plots for other rings were affected an insignificant amount. Correlation of measured and predicted strains was not improved.

With respect to circumferential strain distributions essentially three types of results were obtained. Figure V shows an arbitrary distribution while Figure XVII shows a well defined cosine distribution. Ring No. 3, Figure VIII, appears to combine some of the characteristics of the first two types.

The arbitrary distribution occurred in those rings

vertical section is completely arbitrary. In general, the difference between minimum and maximum thickness in the out-of-round plot is varied the order of 0.05". Since the out-of-round plot is varied from the diameter used for the outer circumference, the corresponding configuration of the central axis may be assumed to deviate slightly from that of the plot. Since sections of thickness greater than average were generally opposite sections of less than average, the effect of this deviation upon the measured value of μ is assumed to be of no consequence except in the circular rings. During a preliminary reduction of data the authors considered the effect which a variation in thickness might have upon the plots of out-of-roundness and predicted strains. The plot of out-of-roundness for Ring No. 1 was altered to a minor extent. The plots for other rings were altered in insignificant amounts. Deviation of measured and predicted strains was not improved.

With respect to circumferential strain distribution essentially three types of results were obtained. Figure V shows an extremely irregular distribution while Figure VII shows a well defined cosine distribution. When No. 3, Figure VIII, appears to combine some of the characteristics of the first two types.

The relative distribution obtained in these rings

which were essentially circular. The theoretical curve shown in Figure V is a plot of equation (3) with the measured out-of-roundness at a particular position substituted for $u_0 \cos 2' \theta$ and with $h = h_{ave} = 0.250"$. A point of interest is the fact that the small peaks which are visible at low pressures define a pattern which would seem to persist up to the maximum pressures obtained. The pattern is characterized by a relatively constant strain over 60° or 90° of the circumference and a fluctuating strain over the remaining 300° or 270° ; the fluctuations are a reasonable representation of two cycles of a cosine curve. The fact to be observed is that correlation between out-of-roundness and strain distribution is completely lacking.

Subsequent results demonstrate quite clearly that theoretical and measured strains agree satisfactorily when the initial configuration assumed in equation (3) is actually obtained in the ring. It is possible, then, that the method of measuring out-of-roundness was not sufficiently accurate in the case of Ring No. 2. In appraising the out-of-roundness measurement, it is convenient to visualize the initial configuration of the ring as a series of the form $R_0 + u_0 \cos 2 \theta + u_1 \cos 3 \theta + u_2 \cos 4 \theta - - - u_n \cos (n + 2) \theta$. The measurements from which Figure VI

which were essentially circular. The theoretical curve shown in Figure V is a plot of equation (5) with θ measured out-of-roundness as a function of position subtended for $u_0 \cos \theta$ and with $h = h_{avg} = 0.950$. A point of interest is the fact that the small peaks which are visible at low pressures define a pattern which would seem to persist up to the maximum pressures obtained. The pattern is characterized by a relatively constant strain over 60° or 90° of the circumference and a fluctuating strain over the remaining 300° or 270° ; the fluctuations are a reasonable representation of the cycles of a cosine curve. The fact to be observed is that correlation between out-of-roundness and strain distribution is completely lacking.

Subsequent results demonstrate quite clearly that theoretical and measured strain rates satisfactorily agree. The initial configuration assumed in equation (5) is actually obtained in the ring. It is possible, then, that the method of measuring out-of-roundness was not sufficiently accurate in the case of Ring No. 2. In appreciation for out-of-roundness measurement, it is convenient to visualize the initial configuration of the ring as a series of the form $u_0 + u_1 \cos \theta + u_2 \cos 2\theta + u_3 \cos 3\theta + \dots$

$u_0 \cos (n + \frac{1}{2}) \theta$. The measurements from which Figure VI

was obtained reflect only those terms of the form $u_0 \cos 2 \theta$, $u_2 \cos 4 \theta$, $u_4 \cos 6 \theta$, etc. It is to be observed that of these terms $u_0 \cos 2 \theta$ appears to predominate. Nevertheless, the assumption of a simple contour, $R_0 + u_0 \cos 2 \theta$, in equation (3) may not be justified since the effect of odd terms ($u_1 \cos 3 \theta$, $u_3 \cos 5 \theta$, etc.) is not indicated by Figure VI. On the other hand, the effect of this omission upon the predicted strain distribution is not so serious as might be expected. If a relation corresponding to equation (3) is derived assuming a configuration such as $R_0 + u_1 \cos 3 \theta$, it will be found that the maximum bending stress is proportional to $\frac{1}{1-P/P_{crit}}$ where

P_{crit} is the buckling pressure of a circular ring into $(n+2)$ lobes. The buckling pressure, P_{crit} , increases rapidly with the number of lobes; thus the value of P_{crit} for a two-lobe collapse is three-eighths of P_{crit} for a three lobe collapse and one-fifth of P_{crit} for a four lobe collapse. Consequently even if u_1 , u_2 , u_3 , etc. are comparable in magnitude to u_0 the effect upon the strain distribution will be considerably less since the bending stress is governed by $\frac{1}{1-P/P_{crit}}$. In view of these facts

the arbitrary strain distribution appears to be the result of a rather complicated combination of inadequate measure-

was obtained only those cases at the time
 as one of the cases of the same kind. It is to be
 observed that of these cases up to the present time
 domestic. Government, we are acquainted with a small number,
 the cases of the same kind (2) may not be limited since
 the effect of the same kind, the cases of the same kind, the
 not limited by the same kind. Of the same kind, the effect
 of this effect may be considered as a small number of
 not as limited as might be expected. It is a small number
 respectively as follows (2) is derived from a small number
 and as follows (2) is derived from a small number and
 cases being cases is proportional to $\frac{1}{1-\sqrt{2}}$ cases
 For the case of the same kind of a small number into
 (2) cases. The number of cases, the number of cases
 really with the number of cases, the number of cases
 for a two-fold cases is three-fold of cases for a
 three-fold cases and one-fold of cases for a four
 fold cases. Consequently even if the cases of the same kind
 comparable in magnitude to the effect upon the same kind
 distribution will be considerably less than the same kind
 cases is governed by $\frac{1}{1-\sqrt{2}}$ in view of these cases
 the effect upon the same kind of cases is to be considered
 of a small number of cases of cases of cases of cases

ment, variation of thickness, and localized surface irregularities or dents.

It follows, however, that a more thorough investigation of essentially circular rings is warranted. Rings could be machined as circular as possible and to a close tolerance in thickness. If the initial contour were then determined precisely, it would be possible to evaluate the effect of deviations from a basic two lobe pattern such as that given in Figure VI. It is the authors' opinion that under these circumstances the method of measuring out-of-roundness used in this thesis would prove useful as a practical measure, especially at high pressures.

Ring No. 3 appeared to be the borderline case. With reference to Figure VIII it is apparent that the circumferential strain distribution at 400 psi. deviated considerably from the theoretical curve, equation (3) with $n = n_{ave} = 0.250$ ". The distribution is not a clearly defined cosine curve, and the magnitude of the strains are generally below the predicted values. On the other hand, at 420 psi. there was a visible deflection of the ring and the strain distribution assumed a form which, except for a 4° - 8° phase shift, compared favorably with predicted values. At 400 psi. the deviations from a pure cosine curve are of the same character as the superposition of harmonics upon

ment, variation of thickness, and localized rupture
 irregularities of density.
 It follows, however, that a more thorough investigation
 of essentially different types is warranted. When
 could be mentioned as discovered in parallel and in a close
 coherence in thickness. If the initial contact were then
 determined precisely, it would be possible to evaluate the
 effect of deviation from a basis two fold greater than the
 that given in figure VI. It is the authors' opinion that
 under these circumstances the method of Hesse's 1937-38
 foundation used in this thesis would prove useful as a
 practical measure, especially at high pressure.
 Work no. 3 appeared to be the better one. With
 reference to figure VII it is apparent that the circum-
 ferential strain distribution at 400 psi. deviated con-
 siderably from the theoretical curve, equation (3) with
 $\mu = 0.25$. The distribution is not a clearly
 defined cosine curve, and the magnitude of the strain are
 generally below the predicted values. On the other hand,
 at 400 psi. there was a visible deviation of the strain
 the strain distribution followed a form which, though not a
 40-psi curve, compared favorably with predicted values.
 at 400 psi. the deviation from a pure cosine curve was of
 the same magnitude as the deviation of the 40-psi curve.

a fundamental wave form. Here it is possible to reason that the terms $u_1 \cos 3 \theta$, $u_2 \cos 4 \theta$, etc. are significant relative to the basic two lobe pattern given by $R_0 + u_0 \cos 2 \theta$. At 420 psi a two lobe strain pattern was clearly predominate; this would be expected since as the pressure approaches P_{crit} for the two lobe collapse pattern, the bending strains become significantly greater than corresponding strains resulting from the superposition of other basic configurations. However, in view of the added complication of the thickness variation and possible localized irregularities the authors do not feel justified in attributing the entire cause of the deviations to an inadequate out-of-round measurement. Again it is suggested that the effect of any deviation from a basic two lobe out-of-round pattern requires further investigation.

The last type of strain distribution, the clearly defined cosine curve, is exemplified by Rings No. 5 and 7, Figures XI-XIII and XVII-XIX. With the exception of Ring No. 6, the plots are typical of the results and correlation observed in Rings No. 4-9. Despite the peaked out-of-roundness curve and the variation in thickness, the correlation of measured and predicted strains is excellent. The manner in which the initial peaks of the strain curve progressively increase in amplitude is clearly demonstrated in Figure XVII. Any deviations from the basic two

in fundamental wave form. Here it is possible to reason that the strain ϵ_1 is $\epsilon_1 = \epsilon_0 + \epsilon_1$, where ϵ_0 is the strain relative to the basic two loop pattern given by $\epsilon_0 + \epsilon_1$ and ϵ_1 is the strain relative to the basic two loop pattern. At 450 psi a two loop pattern was clearly predominant; this would be expected since the pressure approaches zero for the two loop collapse pattern. The bending strains become significantly greater than the corresponding strains resulting from the superposition of other basic configurations. However, in view of the wide dispersion of the thickness variation and possible localized irregularities the authors do not feel justified in attributing the entire cause of the deviation to an inhomogeneous out-of-plane measurement. Again it is suggested that the effect of any deviation from a basic two loop out-of-plane pattern requires further investigation.

The basic type of strain distribution, the clearly defined cosine curve, as exemplified by strains ϵ_0 , ϵ_1 and ϵ_2 , is shown in Figs. 11-13 and 14-16. With the exception of Ring No. 6, the plots are typical of the results and compare favorably with Ring No. 4-2. Besides the basic out-of-plane curve and the variation in thickness, the correlation of measured and predicted strains is excellent. The manner in which the initial peaks of the cosine curve progressively increase in amplitude is directly demonstrated in Figure 11. Any deviation from the basic two

lobe pattern were insignificant at all pressures.

The test of Ring No. 6 differed from the others in that the Plexiglas surfaces were coated with black rubber-to-metal cement before the ring was placed in the test chamber. The gasket and outer surface of the ring were coated in the usual manner. This deviation from the usual test procedure was made in order to determine whether or not the friction between the gasket and Plexiglas could be reduced still further. The results of this test are shown in Figure XIV. The correlation of measured and predicted strains should be compared with that obtained in Figure XI since Rings No. 5 and 6 were of comparable out-of-roundness, $u_0 = 0.0705"$ and $u_0 = 0.075"$ respectively. Where excellent correlation was observed for Ring No. 5, the maximum strains in Ring No. 6 were as much as 40% above the predicted values. This discrepancy cannot be explained by the authors. Were it not for the fact that the results of Rings No. 4, 5, 7, 8, and 9 consistently plotted along the predicted strain curves, it might be assumed that in all cases other than Ring No. 6 a significant restraining force was present at the gasket-Plexiglas interface. The fact that such consistency does exist would imply that the test of Ring No. 6 was faulty in some respect. The results definitely indicate a need for further investigation of the friction force at

These results were insignificant at all points.

The test of Ring No. 6 differed from the others in that the flexural surfaces were covered with black rubber to prevent contact before the ring was placed in the test chamber. The result was a lower value of the rate of growth in the usual manner. This deviation from the usual test procedure was made in order to determine whether or not the friction between the contact and flexural surfaces was reduced still further. The results of this test are shown in Figure XIV. The correlation of measured and predicted strains should be compared with that obtained in Figure 11 since Rings No. 5 and 6 were of comparable out-of-roundness, $\mu = 0.0025$ and $\mu = 0.002$ respectively. Where excellent correlation was observed for Ring No. 5, the maximum strain in Ring No. 6 were as much as 40% above the predicted values. This discrepancy cannot be explained by the methods. Here it is for the first time the results of Rings No. 5, 6, 7, 8, and 9 consistently plotted along the same straight line. It might be supposed that in all cases other than Ring No. 6 a significant restraining force was present at the contact-flexural interface. The fact that such was not the case was not until Ring No. 6 was tested at the test of Ring No. 6 was fairly in line with the results of the other tests. A test for further investigation of the flexural force at

the gasket-Plexiglas interface, such as suggested previously.

Maximum Strains

In those rings which were intentionally deformed strain gages were placed at points of maximum (0° and 180°) and minimum (90° and 270°) diameters. The strains at 0° and 180° were averaged and plotted for Rings No. 3-9 on the right side of Figures XX - XXVI. Similarly, the average of the 90° and 270° readings are shown on the left side of the same plots. The theoretical curve is simply a plot of equation (3) with $\cos 2 \theta$ equal to plus or minus one and $h = h_{ave.} = 0.250$ ". The strains at 0° and 180° are particularly important in that they are normally used in the prediction of a collapse pressure; consequently, the strains at these positions are given added consideration.

With the exception of Rings No. 3 and 6, the correlation of measured and theoretical strain is entirely satisfactory. On the pressure scale the compressive strains are generally within a range of $\pm 5\frac{1}{2}$ psi. of the theoretical curve; this was the assumed accuracy of the recorded pressure. The tensile strains always appear below the theoretical curve; if the modulus of elasticity in tension were assumed to be slightly below 10.8×10^6 psi. the tensile strains would also correlate. In view of the consistency noted in Figures XXI, XXII, XXIV, XXV, and XXVI, the authors are inclined to think that this might actually be the case.

The gas-liquid interface, such as suggested previously.

Results

In these runs which were intentionally disturbed strain gauges were placed at points of maximum (0° and 180°) and minimum (90° and 270°) diameter. The strains at 0° and 180° were averaged and plotted for runs No. 5-9 on the right side of Figure 11 - 11V. Similarly, the average at the 90° and 270° readings are shown on the left side of the same plot. The theoretical curve is simply a plot of equation (2) with $\nu = 0.3$ equal to plus or minus one and $\mu = 0.3$. The strains at 0° and 180° are positive, the strains at 90° and 270° are negative. The strains at these positions are given in Figure 11. The correlation of measured and theoretical strains is highly satisfactory. On the pressure scale the compressive strains are generally within a range of $\pm 1\frac{1}{2}$ cal. of the theoretical curve. This was the assumed accuracy of the tested pressure. The general strain shape agrees with the theoretical curve; if the relation of diameter to radius were assumed to be slightly other than 1.5×10^3 cal. for small strains would also be satisfactory. In view of the consistency noted in Figures 11, 11V, 11V, and 11V, the strains are included to show that this is not generally the case.

In a sense Rings No. 3 and 6 were special cases. As pointed out Ring No. 3 appeared to be the dividing line between configurations in which the two lobe out-of-roundness pattern predominated and those which were affected by variations in thickness, harmonics, and/or local irregularities such as a small dent or gouge. Figure XX shows clearly that the measured strains were considerably below the predicted values up to a pressure of 400 psi; at this point the measured strains literally jumped to the theoretical curve. Figure XX tends to substantiate the remarks made previously; that is, in practical cases of out-of-roundness a two lobe configuration derived simply and directly may allow prediction of strains of acceptable accuracy at sufficiently high pressures, but further work is required before complete justification is obtained.

Figure XXIII shows merely that measured strains for Ring No. 6 were larger than the predicted strains over the entire range of pressures. The consistency lends some doubt to the authors' contention that the test may have been faulty.

In conjunction with the considerations of circumferential strain distribution, the results above indicate quite clearly that equation (3) is valid within the qualifications of the derivation; this conclusion is based upon the behavior of

[illegible]

Rings No. 4, 5, 7, 8, and 9 in which the assumption of a configuration $R_0 + u_0 \cos 2 \theta$ was obviously justified.

Collapse Pressure

Figure XXVII shows the collapse pressures of the nine rings plotted versus the ratio of out-of-roundness to average thickness and serves to identify the points shown on Figure XXVIII. The single curve is a plot of equation (4) with $\sigma_{\max} = 35,000$ psi. Figure XXVIII is more informative in that several variations of equation (4) are plotted; the values of σ_{\max} and h which apply to each curve are indicated in the legend.

Foregoing results (see Figure V) indicate that Rings No. 1 and 2 did not assume a two lobe configuration until the ring was in the process of failing completely; if a stable two lobe pattern existed it was possible over a very small range of pressure. Regardless of the cause of the random strain distribution, Figure XXVIII demonstrates clearly that the failure of Rings No. 1 and 2 could have been predicted by equation (4) in which σ_{\max} is 41,000 psi and $h = h_{\min.} = 0.243$ ".

There are two ways in which to view these results. First, it would appear that the minimum thickness, u_0 as measured, and the point at which the stress-strain curve departs significantly from linearity are the only factors

Figure 10. 1, 2, 3, 4, 5, 6, 7, 8, and 9 are the results of a
 configuration No. 1 in the 3.5 mm diameter, installed.

Colony Results

Figure 11 shows the colony pressure of the high
 stage plotted versus the ratio of out-of-roundness to
 average thickness and serves to identify the points shown
 on Figure 10. The single curve is a plot of equation
 (4) with $\Delta = 35,000$ psi. Figure 11 is more in-
 dicated in that several variations of equation (4) are plotted;
 the values of Δ and Δ which apply to each curve are
 indicated in the legend.

Regarding results (see Figure 11) indicate that range
 No. 1 and 2 did not assume a two lobe configuration until
 the time was in the process of falling completely; it is
 stable two lobe pattern existed it was possible over a very
 small range of pressure. Regardless of the cause of the
 problem, again distribution, Figure 11 demonstrates
 clearly that the ratio of range No. 1 and 2 could have
 been predicted by equation (4) in which $\Delta = 35,000$
 psi and $\Delta = 0.001$.

There are two ways in which to view these results.
 First, it would seem that the initial thickness, as
 measured, and the point at which the stress-strain curve
 departed significantly from linearity are the only factors

to be considered; hence, the use of equation (4) would be justified. This line of reasoning would infer that previous remarks relative to the importance of the basic two lobe configuration are entirely correct and that in practical cases of out-of-roundness one could neglect out-of-round other than that indicated by a simple consideration of diameters. On the other hand it is possible to reason that the variation in thickness was actually not significant; this statement could be based upon the collapse pressure of Rings No. 3, 4, and 5. Then the occurrence of collapse below the prediction based upon average thickness would be explained in terms of stress introduced by a superimposed but undetected configuration or by local irregularities. The authors can conclude only that in this case of practical out-of-roundness and arbitrary strain distribution the collapse was predicted by equation (4) using minimum thickness and $\sigma_{\max} = 41,000$ psi.

Rings No. 3, 4, and 5 differed from Rings No. 1 and 2 in that a clearly defined cosine type strain distribution was observed before failure. In each case the strain peaks corresponded to peaks in the out-of-roundness plot. Furthermore, the progressive failure which occurred in these rings may be contrasted to the rather sudden collapse of Rings No. 1 and 2.

to be considered; hence, the use of equation (4) would be justified. This line of reasoning would imply that previous remarks relative to the importance of the term $\frac{1}{2} \rho \omega^2 r^2$ in the configuration are entirely correct and that in previous cases of out-of-roundness the term $\frac{1}{2} \rho \omega^2 r^2$ is neglected out-of-round other than that indicated by a simple consideration of stresses. On the other hand it is possible to reason that the variation in thickness was actually not significant; this statement could be based upon the collapse pressure of Rings No. 3, 4, and 5. Then the occurrence of collapse below the critical pressure would suggest thickness would be explained in terms of stress introduced by a superimposed but undistorted configuration or by local irregularities. The authors can conclude only that in this case of practical out-of-roundness and existing grain distortion the collapse was predicted by equation (4) using average thickness and $D_{avg} = 41,000$ psi.

Rings No. 3, 4, and 5 differed from Rings No. 1 and 2 in that a slightly different grain type grain orientation was observed before failure. In some cases the grain lines corresponded to cracks in the out-of-roundness plot. Furthermore, the progressive failure which occurred in these rings may be associated in the collapse modes collapse at Rings No. 1 and 2.

Figure XXVIII shows that Rings No. 3, 4, and 5 collapsed at pressures very near to predictions given by equation (4) with $\sigma_{\max} = 41,000$ psi and $h = h_{\text{ave}}$. These results seem to justify the argument above that the variation in thickness in itself was not significant and that the behavior of Rings No. 1 and 2 should be explained on the basis of harmonics and/or local irregularities. Nevertheless, the authors' only conclusion is that for rings of u_0/h from 0.10 to 0.30 equation (4), when used with the stress at which marked departure from linearity occurs, will predict quite accurately the failures in those rings where an initial configuration of $R_0 + u_0 \cos 2 \theta$ is predominate.

Rings No. 7, 8, and 9 collapsed at pressures consistently above the predicted collapse pressure. Furthermore, stable configurations were obtained in which the maximum compressive stress was considerably above the yield point. Obviously as u_0 increases, the criteria used in computing the collapse pressure becomes more conservative. This result is explained by the fact that the hoop stress upon which the bending stress is superimposed was less at the time yielding occurred in the outer fibers; in addition, the change in stress with respect to pressure was also less in the rings of greater u_0 when this yielding began. Then to obtain instability in rings of greater out-of-roundness a larger increase in

Figure XVIII shows that since σ_1 , σ_2 , and σ_3 collapsed
the pressure was not in equilibrium (4) as
with $\sigma_1 = 41,000$ psi and $\sigma_2 = 0$ psi. These results seem to
justify the statement above that the variation in stresses
in itself was not significant and that the behavior of
stress σ_1 and σ_2 should be explained on the basis of
principles and/or local irregularities. Nevertheless, the
analysis and conclusion is that for time of test from
0.15 to 0.30 seconds (4) when used with the stress at which
marked departure from linearity occurs, will predict data
accurately the behavior in those cases where no initial
configuration of $\sigma_1 + \sigma_2 + \sigma_3$ is predominant.
Stress σ_1 , σ_2 , and σ_3 collapsed at pressures consistently
above the predicted collapse pressure. Furthermore, stable
configurations were obtained in which the maximum compressive
stress was consistently above the yield point. Obviously as
no increase, the criteria used in computing the collapse
pressure becomes more conservative. This result is explained
by the fact that the hoop stress upon which the bending stress
is superimposed was less at the time yielding occurred in
the outer fibers; in addition, the change in stress with
respect to pressure was also less in the case of stress
no when this relation began. Then to obtain instability in
times of tested out-of-roundness a larger increase in

pressure was required beyond the point of initial yield in the outer fibers. The authors conclude that for values of u_0/h greater than 0.30 and less than 1.10 the predicted collapse pressures are somewhat conservative but would be considered satisfactory for usual engineering design.

considered satisfactory for normal engineering design.

V.

CONCLUSIONS

1. The properties of aluminum alloy 61S-T6 were particularly well adapted to the objectives of this thesis.
2. The most significant problem encountered in the method of experimentation was the reduction of friction at the gasket-Plexiglas interface.
3. The relation for predicted strains in thin rings, equation (3), is valid when the assumed configuration, $R_0 + u_0 \cos 2 \theta$, is obtained.
4. In those cases where the equation for predicted strains is valid and the value of h/D is 0.0285 a failure criteria based upon the stress level in the outer fibers predicts quite accurately the collapse of thin rings in which u_0/h is between 0.10 and 0.30. For values of u_0/h between 0.30 and 1.10 the criteria is somewhat conservative but not to an extent which would be considered over-cautious in engineering design.
5. From the general trend of the experimental collapse pressures as u_0/h decreases, the authors conclude that the collapse pressure predicted for perfectly circular rings by the Levy Formula, $P_{crit} = \frac{3 E I}{R^3}$, is valid.

CONCLUSIONS

1. The properties of aluminum alloy 618-T3 were previously well adapted to the objectives of this investigation.
2. The most significant problem encountered in the method of experimentation was the reduction in friction at the contact-point interface.
3. The relation for predicted stress in shear stress, equation (3), is valid when the assumed coefficient, $\mu + \mu \cos 2\theta$, is obtained.
4. In those cases where the equation for predicted stress is valid and the value of μ/P is 0.025 or higher, criteria based upon the stress level in the outer fibers predicts quite accurately the collapse of thin rings in which θ is between 0.10 and 0.30. For values of μ/P between 0.10 and 1.10 the criteria is somewhat conservative but not to an extent which would be considered over-conditions in engineering design.
5. From the general trend of the experimental collapse pressure as μ/P increased, the authors conclude that the collapse pressure predicted for perfectly spherical rings by the Levy formula, $\sigma_{cr} = \frac{3}{8} \frac{E}{1 - \nu}$, is valid.

VI.

RECOMMENDATIONS

1. Aluminum alloy 61S-T6 should be used in any extension of the experimentation described in this thesis.
2. A thorough, quantitative investigation of the friction at the gasket-Plexiglas interface should precede any further use of a test apparatus such as that described herein.
3. An investigation should be made for the purpose of determining the effect of deviations from a basic two lobe pattern as measured in this thesis. The rings to be investigated should be of constant thickness and the initial configuration should be known precisely.
4. A series of rings, machined as circular as possible and to a close tolerance in thickness, should be tested in order to further substantiate the Levy Formula.
5. It is recommended that the scope of this experimentation be extended to include commercial shapes such as I or H sections.

FIGURE 1. ACPZ IMMUNOLOGY

1. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
2. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
3. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
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5. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
6. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
7. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
8. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
9. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:
10. It is recommended that the scope of this investigation should be extended to include commercial types such as those of the following:

VII.

A P P E N D I X

CONTENTS

1. A general account of the history of the subject, from the earliest times to the present day.
2. The various theories of the origin of the subject, and the evidence in support of each.
3. A detailed account of the various theories of the origin of the subject, and the evidence in support of each.
4. The various theories of the origin of the subject, and the evidence in support of each.

VII.

APPENDIX

1. A general account of the history of the subject, from the earliest times to the present day.
2. The various theories of the origin of the subject, and the evidence in support of each.
3. A detailed account of the various theories of the origin of the subject, and the evidence in support of each.
4. The various theories of the origin of the subject, and the evidence in support of each.

APPENDIX A

Details of Procedures

Selection of Material for Construction of Rings

Initially the authors considered the possibility of using steel in the manufacture of the rings. A length of centrifugally cast steel tubing of 18" outside diameter and 1-3/4" wall thickness was available. However, the expense of constructing a test apparatus to accommodate steel rings was considered prohibitive in view of the uncertainty of the results. Furthermore, it was felt that the visual test chamber made possible through the use of Flexiglas would be of tremendous advantage during the test phase; this innovation was not considered feasible in the preliminary design of an apparatus to accommodate the pressures necessary to collapse steel rings.

The use of plastics was also given consideration. The various types of plastics were gradually eliminated for one or more of the following reasons:

1. The relatively low modulus of plastics resulted in very low predicted collapse pressures.
2. The ultimate strength was generally low relative to available metals.
3. The extent of a linear stress-strain relationship was limited.

APPENDIX A

Details of Procedures

Selection of Material for Construction of Rings

Initially the authors considered the possibility of using steel in the construction of the rings. A number of essentially equal steel rings of 3/4" outside diameter and 1-3/4" wall thickness was available. However, the expense of constructing a test apparatus to accommodate steel rings was considered prohibitive in view of the uncertainty of the results. Furthermore, it was felt that the visual test chamber with possible through the use of fluorine would be of tremendous advantage during the test phase; this innovation was not considered feasible in the preliminary design of an apparatus to accommodate the pressure necessary to collapse steel rings.

The use of plastic was also given consideration. The various types of plastic were gradually eliminated for one or more of the following reasons:

1. The relatively low modulus of plastic resulted in very low hydraulic cylinder pressures.
2. The ultimate strength was generally low relative to available metals.
3. The extent of a linear stress-strain relationship was limited.

These considerations indicated that the range of pressures over which useful data could be obtained was very limited, and, further, that the accuracy of the results would be questionable unless extreme precautions were taken in instrumentation.

As noted in Section II, aluminum alloy 61S-T6 appeared to satisfy the requirements of moderate collapse pressures, an essentially linear stress-strain relationship over a considerable range of stresses, and availability.

Manufacture of Rings

As pointed out in Section II the aluminum alloy tubing as received was cut into lengths several times the intended width of the test rings. Alternatively, the rings could have been machined to final dimensions. However, it is rather doubtful that the edges of the finished ring would have remained planar after the ring had been subjected to deformation in a loading machine and a heat treatment. A ring warped in this manner would tend to bind against the plane sides of the annular test chamber within the test apparatus. Consequently, the machining of the rings to the specified width was scheduled as the last operation in the manufacturing process.

The heat treatment of the aluminum alloy was accomplished by the Forge Shop, Boston Naval Shipyard. Specifically, the

These considerations indicated that the range of pressures over which useful data could be obtained was very limited, and, further, that the accuracy of the results would be questionable unless extreme precautions were taken in instrumentation.

As noted in Section II, aluminum alloy 615-T6 appeared to satisfy the requirements of moderate collapse pressure, an essentially linear stress-strain relationship over a considerable range of stresses, and availability.

Manufacture of Rings

As pointed out in Section II the aluminum alloy tubing as received was cut into lengths several times the internal width of the test rings. Alternatively, the rings could have been machined to final dimensions. However, it is rather doubtful that the edges of the finished rings would have remained planar after the ring had been subjected to deformation in a loading machine and a heat treatment. A ring warped in this manner would tend to ring within the test plane edges of the universal test chamber within the test apparatus. Consequently, the accuracy of the data so obtained would be diminished as the test operation in the manufacturing process.

The heat treatment of the aluminum alloy was accomplished by the Forge Shop, Boston Naval Arsenal. Specifically, the

solution heat treatment consisted of a two hour soaking period at 970° F followed by a rapid quench in hot water at 180°F. The hot water quench was specified as an additional precaution against residual stresses. Actually the danger of residual stresses was not particularly significant in view of the rather thin thickness of alloy being quenched; nevertheless, the precaution was considered worthwhile since a hot water quench does not reduce the tensile properties appreciably below those obtainable with a cold water quench. The quench was followed by a precipitation heat treatment at 350°F for eight hours. The characteristics of aluminum alloy 61S-T6 as given in reference (5) are:

Tensile Strength	45,000 psi.
Yield Strength	40,000 psi.
E, Young's Modulus	10×10^6 psi.

Upon inspection of the sections after the heat treatment, it was found that the magnitude of the out-of-roundness introduced had not changed by any significant amount.

Instrumentation

The requirements placed upon the type strain gage selected were that the gage be small enough for mounting in the space available and that the gage length be sufficiently short with respect to the strain gradient. The SR-4, type A-7, strain gage satisfied these requirements. The gages

collected from specimens annealed at 1000°F. The hot water quench was specified as an period at 270°F followed by a rapid quench in hot water. Additional protection against residual stresses. Actually the danger of residual stresses was not particularly significant in view of the rather thin thickness of alloy being quenched; nevertheless, the protection was considered worthwhile since a hot water quench does not reduce the tensile properties appreciably below those obtainable with a cold water quench. The quench was followed by a prompt section heat treatment at 350°F for eight hours. The characteristics of aluminum alloy 618-T6 are given in

reference (2) are:

Tensile Strength	42,000 psi.
Yield Strength	40,000 psi.
E. Young's Modulus	10,100 psi.

Upon inspection of the sections after the heat treatment, it was found that the magnitude of the one-third reduction introduced had not changed by any significant amount.

Investigation

The requirements placed upon the type strain gage selected were that the gage be small enough for mounting in the space available and that the gage loading be sufficiently short with respect to the strain gradient. The 25-4 type A-7 strain gage satisfied these requirements. The gage

could be easily mounted within the width of the ring, and, furthermore, a 3/16" gage length was entirely adequate since the predicted stress gradient in the vicinity of points of maximum stress was very small.

A multiple selector switchbox was used to facilitate reading the strain indications. The switchbox was constructed to permit the operator to set the same initial reading on all strain gages.

Design and Manufacture of Test Apparatus

The primary concern of the authors in the initial design of a test apparatus was the application of a uniform pressure on the outer surface of the rings without the introduction of restraining forces on the ring edges. An annular test chamber appeared to be the most suitable in this respect. The annulus was designed such that the tolerance between the surfaces and the edges of the ring would be exceedingly small, i.e., order of 0.001-0.002". This would facilitate sealing and yet the ring would remain "free floating" so to speak. From preliminary calculations based on the Levy formula for the collapse of 9" diameter aluminum rings of the 1/4 inch thickness, it was estimated that the test chamber would be subjected to a pressure of approximately 500 psi. Because Plexiglas is beautifully clear and can be obtained in large thick sheets with polished surfaces, it was decided to design the test annulus with upper and

could be easily mounted within the limits of the ring, and, furthermore, a $3/16$ " pipe length was selected as being suitable since the predicted stress gradient in the vicinity of points of entrance stress was very small.

A multiple deflection switcher was used to facilitate reading the strain indications. The switcher was connected to permit the operator to set the same initial reading on all strain gages.

Design and Manufacture of Test Apparatus

The primary concern of the authors in the initial design of a test apparatus was the application of a uniform pressure on the outer surface of the ring without the introduction of restraining forces on the ring edges. An annular test chamber appeared to be the most suitable in this respect. The annulus was designed such that the tolerance between the surfaces and the edges of the ring would be exceedingly small, i.e., order of $0.001-0.003$ ". This would facilitate seating and yet the ring would remain "free-floating" so that from preliminary calculations based on the Levy formula for the collapse of 8" diameter aluminum rings of the 1/4 inch thickness, it was estimated that the test chamber would be subjected to a pressure of approximately 200 psi. Because plasticity is essentially elastic and can be obtained in large rings with polished surfaces, it was decided to design the test chamber with upper and

lower surfaces of this material. This construction would permit visual inspection of collapse and the possibility of obtaining deflection readings. Plexiglas, however, has a low modulus of elasticity and therefore required backing to prevent the surface from deflecting appreciably when subjected to the expected pressures. The top steel web assembly and the bottom steel reinforcing ring as indicated in Figure I were designed for limiting the deflections of the Plexiglas. The hole in the bottom Plexiglas and reinforcing plates was introduced to accommodate strain gage leads. The steel surfaces indicated to be surface ground in Figure I were so designated to provide good contact surfaces between the steel and the Plexiglas. The steel surfaces as received, particularly those of the spacer ring, would not have insured a uniform spacing of the two surfaces forming the annulus.

Although four $3/8$ " bolts would have been sufficient from considerations of strength alone, it was decided to use sixteen in order to obtain oil tightness.

Originally rubber gasket material was to be used between the spacer ring and the Plexiglas surfaces to prevent oil leakage at the junction of the steel and Plexiglas outside the annulus. However, the rubber gasket would have been compressed in varying amounts around the test apparatus,

lower portion of this material. This construction permits
 heavy axial loading of rollers and the possibility of
 excessive deflection resulting. However, as a
 low modulus of elasticity and therefore required bending to
 prevent the surface from deflection especially when con-
 sidered for the attached pressure. The steel used was generally
 and the roller steel reinforcing ring as indicated in
 Figure 1 was designed for limiting the deflection of the
 rollers. The hole in the bottom flange and roller
 plates was introduced to accommodate strain gage leads.
 The steel rollers indicated to be subject to
 Figure 1 were so designed to provide good contact surfaces
 between the steel and the flange. The steel rollers as
 received, particularly those of the square type, would not
 have insured a uniform spacing of the two contact rollers
 in operation.
 Although four 3/8" bolts would have been sufficient
 from consideration of strength alone, it was desired to
 use sixteen in order to obtain all thickness.
 Originally much greater material was to be used between
 the square ring and the flange rollers to prevent oil
 leakage at the junction of the steel and flange rollers
 the rollers. However, the roller bolts would have been
 compressed in rolling motion along the bolt openings.

depending on the tension in each bolt upon tightening up. It was possible to circumvent this difficulty by using paper of 0.010" thickness.

The pressure tap was designed to accommodate a 1/4" copper tube fitting. The air release tap diametrically opposite was so placed to vent the test chamber while oil was being introduced.

The cathetometer used to take deflection readings was mounted as indicated in Figure II and Photograph No. (4). The heavy brass plug on which the cathetometer is mounted was machined to fit snugly into the thick walled pipe which had been welded into the center of the top web assembly of the test apparatus; the thick-walled pipe was machined true to an axis perpendicular to the surface of the Plexiglas. A bench mark was established on the web assembly to facilitate taking deflection readings.

Proof Test of Apparatus

The objective of this phase of the testing was to solve as completely as possible the many unexpected problems of detail which plague any initial effort. In addition it was desirable to establish a standard test procedure to be used during the remainder of the experimental work.

Of concern to the authors during the design of the

depending on the location in which the specimen was
it was possible to determine this direction of strain
from the shape of the specimen, and the direction of strain
The specimen was then placed in a container in which
specimen was placed. The air pressure was then increased
opposite was an attempt to force the specimen into the
was being introduced.
The container was used to take additional readings and
mounted as indicated in Figure 11 and Figure 12. (A)
The heavy base was placed on the container as shown
was attached to the sample into the strain gauge type also
had been placed into the center of the top and capacity of
the test specimen; the thickness of the specimen was measured
first to the strain gauge and then to the center of the
specimen. A second strain gauge was attached to the top
assembly to facilitate taking additional readings.

Test of Specimen

The objective of this test of the specimen was to
solve as completely as possible the strain measurement problem
of strain which places any strain gauge. In addition
it was desired to establish a standard test procedure
to be used during the preparation of the specimen for
of course to the extent that the strain of the

apparatus was the need for an effective seal between the Plexiglas surfaces and the edges of the test ring. Preliminary considerations indicated that the most likely solution would be the use of either a leather or rubber gasket of slightly greater width than the ring. The gasket could be shaped to fold back against the Plexiglas surface when the top assembly was bolted down. See Figure XXIX (a). To facilitate assembly, the gasket could be glued to the outer circumference of the ring prior to insertion in the test chamber.

The following trial gaskets were selected for preliminary investigation:

1. A strip of bicycle tire inner tube cut to $3/4$ " width, 28" in length, and joined at the ends with rubber cement. Results: Not successful because a secure junction could not be obtained.
2. Rubber electrical tape applied to the outer circumference with the ends lapped.
Results: Extrusion of the tape between the Plexiglas surfaces and the edges of the ring occurred at pressures of about 400 psi.
3. Leather strip $3/4$ " wide, 28" in length with junction secured by a special leather cement.
Results: Not successful because the strength of the junction was unsatisfactory.

apparatus was the need for an effective seal between the
flexible members and the edges of the test piece. The
primary consideration indicated that the seal likely
would be the use of either a leather or rubber
gasket of slightly greater width than the rim. The
gasket could be shaped to fold back against the flexibles
surface when the top assembly was pulled down. See
Figure III (a). To facilitate assembly, the gasket could
be fitted to the outer circumference of the rim prior to
insertion in the test chamber.

The following trial gaskets were selected for pre-
liminary investigation:

1. A strip of bicycle tire inner tube cut to $3\frac{1}{4}$ "
width, 28" in length, and joined at the ends
with rubber cement. Result: Not successful
because a secure junction could not be obtained.
2. Rubber electrical tape applied to the outer cir-
cumference with the ends lapped.
Result: Extension of the tape between the flexi-
ble members and the edges of the rim caused
a protrusion of about 400 psi.
3. Leather strip $3\frac{1}{4}$ " wide, 28" in length with junction
secured by a special leather cement.
Result: Not successful because the extension of
the junction was unsatisfactory.

FIGURE XXIX

DETAILS OF GASKET

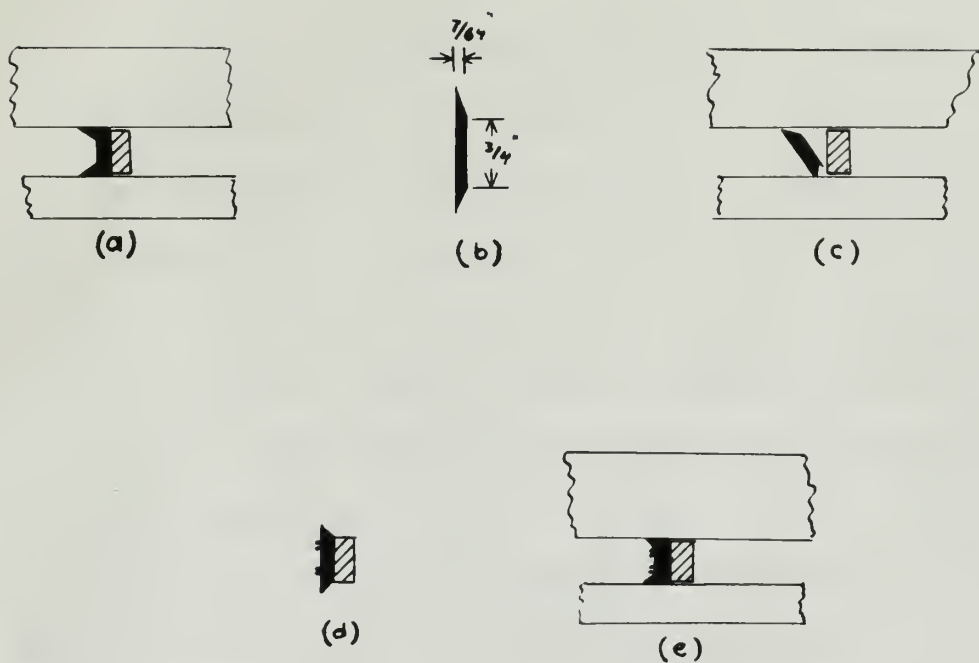
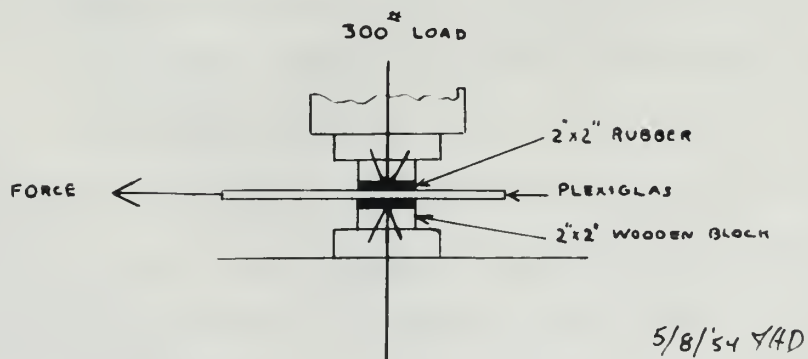


FIGURE XXX

DETAILS OF LUBRICANT TEST



4. Truck tire inner tube 9" in diameter cut to 3/4" width with edges beveled as indicated in Figure XXIX(b).

Results: Successful in holding pressures up to 875 psi.; it is interesting to note that even at this pressure the gasket did not extrude. Pressure was not increased beyond 875 psi. for fear of introducing a permanent set in the top Plexiglas surface. There was no junction difficulty in this particular case because the gasket was continuous.

Further experimentation with gasket No. 4 indicated that the 3/4" width was unnecessary and could be reduced to 1/2". The angle of the bevel was changed to 45°. The procedure for cutting the final gaskets was as follows:

1. Glue a 2" section of rubber tubing around the circumference of an 11" wooden disk attached to a lathe chuck.
2. Place the disk in the lathe and turn at slow speeds. (At high speeds the rubber was torn loose from the wooden disk)
3. Hold a sharp, pointed cutting tool at 45° to the edge of the disk and slowly press the tool into the rubber in much the same manner as if turning down wood.

4. Track size inner tube 2" in diameter out to 3/4" width with edges beveled as indicated in Figure XIX(c).

Results: Successful in building pressure up to 875 psi.; it is interesting to note that even at this pressure the wheel did not extrude. Pressure was not increased beyond 875 psi. for fear of producing a permanent set in the top Plexiglas window. There was no junction difficulty in this particular case because the wheel was continuous.

Further experimentation with wheel No. 4 indicated that the 3/4" width was unnecessary and could be reduced to 1/2". The angle of the bevel was changed to 45°. The procedure for making the final wheels was as follows:

1. Cut a 2" section of rubber tubing around the circumference of an 11" section disk attached to a latex wheel.
2. Place the disk in the lathe and turn it slow speeds. (At high speeds the rubber was torn from the wheel disk)
3. Hold a sharp, pointed cutting tool at 45° to the edge of the disk and slowly press the tool into the rubber in such a way as to remove as it possible down wood.

4. After the cutting blade has passed completely through the rubber, the tool is removed and inserted 1/2" from the first cut, the inclination being that required to produce a reverse 45° bevel.

Gaskets manufactured as above were glued to the outer circumference of the preliminary test rings with clear or black rubber cement. It was thought that a film of oil between the rubber gasket and the Plexiglas surfaces would be sufficient to lubricate the gasket and permit relatively friction free slipping. The first preliminary test ring was placed in the test chamber. Pressures well above the predicted failure pressure were applied without collapse of the ring. The possibility of passing through the first critical collapse pressure was discounted as a complete explanation because without noticeable restraint such could occur only by a combination of ideal conditions and fortuitious circumstances. The apparatus was repeatedly opened to adjust the gasket and, if possible to determine the cause of the restraint. After numerous trials the ring buckled at approximately 550 psi., well above the predicted collapse pressure.

The second preliminary test ring was similarly tested several times without success. At this point the authors

decided to direct all efforts toward the elimination of the restraining force which prevented the collapse of the rings at reasonable pressures. It was assumed that the restraining force occurred as a result of friction between the rubber gasket and the Plexiglas since without the gasket the rings were found to be completely free when enclosed in the test chamber.

Several types of lubricants were investigated by spreading the lubricant over two rubber 2"x2" squares and sandwiching a piece of Plexiglas between them. A load was applied to the rubber squares by means of wooden blocks placed between the rubber squares and the heads of a loading machine. See Figure XXX. The force necessary to move the Plexiglas under specified loads was observed.

The following lubricants were used:

1. Number 40 S.A.E. motor oil.

Results: Plexiglas could not be moved by a force less than 40 lbs. when a load of 300 lbs. was applied by the loading machine.

2. Silicone Compound DC4.

Results: Same as No. 1.

3. Vaseline

Results: Plexiglas could be moved by a force of approximately 8 lbs. when a load of 300 lbs. was applied by the loading machine.

desired to direct all efforts toward the elimination of the
resisting force which prevented the collapse of the chain
of resistance. It was assumed that the resistance

the force occurred as a result of friction between the
rope and the pulleys and the friction between the pulleys
the rope was found to be completely free when released
in the test chamber.

Several types of lubricants were investigated by spread-
ing the lubricant over two rubber "X"s, securing and sand-
wiching a piece of plastic between them. A load was
applied to the rubber squares by means of wooden blocks
placed between the rubber squares and the heads of a
loading machine. See Figure III. The force necessary to
move the plastic under specified loads was observed.

The following lubricants were used:

1. Lubricant 40 S.A.E. motor oil.

2. Grease; Plastigrip would not be used as a
force less than 40 lbs. when a load of 500 lbs.
was applied by the loading machine.

3. Silicone Grease, RCA.

4. Grease, same as No. 1.

5. Grease.

6. Grease; Plastigrip would be used as a force of
approximately 5 lbs. when a load of 500 lbs. was
applied by the loading machine.

4. Clear rubber cement applied to rubber and oil film on Plexiglas.

Results: Plexiglas could be moved by a force of approximately 6 lbs. when a load of 300 lbs. was applied by the loading machine.

5. Black rubber-to-metal cement applied to rubber and oil film on Plexiglas.

Results: Lubricant effective to the extent that the 2" rubber squares would not remain in place under wooden blocks as the load was being applied.

On the basis of the preceding results it was concluded that the black rubber-to-metal cement was the most suitable lubricant. The rubber gasket was cemented to the outer circumference of the preliminary test ring with a large amount of cement. The aluminum ring was then placed in the test apparatus and the top assembly bolted in position. It was noted at this time that the gasket would not remain in position on the aluminum ring but pulled away in the manner indicated in Figure XXIX (c). The top assembly was removed and several strands of light string were passed around the outer circumference of the gasket. See Figure XXIX (d). The beveled edges were observed to fold back uniformly along the surface of the Plexiglas when the ring and gasket were again placed in the test chamber and the top assembly had been positioned and bolted down. See Figure XXIX (e).

4. Clear rubber cement applied to rubber and all this on Plexiglas.

Result: Plexiglas could be moved by a force of approximately 5 lbs. when a load of 500 lbs. was applied by the loading machine.

5. Black rubber-to-metal cement applied to rubber and all this on Plexiglas.

Result: Imperfect adhesive to the extent that the 9" rubber squares would not remain in place under wooden blocks as the load was being applied. On the basis of the preceding results it was concluded that the black rubber-to-metal cement was the best adhesive

available. The rubber cement was cemented to the outer circumference of the preliminary test ring with a large amount of cement. The aluminum ring was then placed in the test apparatus and the top assembly bolted in position. It was noted at this time that the cement would not remain in position on the aluminum ring but pulled away in the corner indicated in Figure III (a). The top assembly was removed and several strands of light string were passed around the outer circumference of the pulley. See Figure III (a).

The results were observed in this test assembly along the surface of the Plexiglas when the ring was tested were again placed in the test chamber and the top assembly had been positioned and bolted down. See Figure III (a).

Pressure was applied and the preliminary test ring buckled very close to the predicted collapse pressure.

The test procedure with respect to gasket material, gasket size, and lubrication was thus established. It is worthwhile to note that the best results were obtained when the black cement was applied to the beveled surfaces, the inner surface of the gasket, and the outer surface of the ring. The cement was allowed to dry slightly after application.

In an effort to obtain deflection readings the authors considered testing the rings in two phases. During the first phase clear rubber cement would be used as the lubricant and both deflection readings and strain readings were to be recorded up to the pressure at which collapse was expected. The pressure would then be released and the ring removed for lubrication with the black cement. During the second phase strain readings were to be taken until actual collapse occurred. However, as pointed out previously strains taken during the two phases did not correlate. Furthermore, the cathetometer proved impractical. The third preliminary test ring was strain gaged and tested with various lubricants and gasket material to show that the strains did in fact depend upon these factors. Results of these tests were presented in Figure IV.

Pressure was applied and the preliminary test was conducted very close to the predicted collapse pressure.

The test procedure with respect to gas pressure, gas flow, and deflection was also maintained. It is worthwhile to note that the test results were obtained when the black cement was applied to the devalued surface, the inner surface of the gas, and the outer surface of the ring. The cement was allowed to dry slightly after application.

In an effort to obtain deflection readings and stresses considered testing the rings in two phases. During the first phase clear tubes cement would be used as the instrument and both deflection readings and strain readings were to be recorded up to the pressure at which collapse was expected. The pressure would then be released and the ring removed for inspection with the black cement. During the second phase strain readings were to be taken until actual collapse occurred. However, as pointed out previously, strain readings taken during the two phases did not differ. Furthermore, the measurements proved impractical. The third preliminary test ring was strain gaged and tested with various instruments and gases resulting in some loss of strain due to loss of gas upon release. Results of these tests were presented in Figure IV.

During the proof testing period oil leakage around the bolt holes was noted. A strip of rubber electrical tape 3/4" in width was glued to the inner circumference of the spacer ring. The edges of this seal folded back toward the center of the test chamber and were pressed tightly against the Plexiglas when pressure was applied. Holes were cut in the electrical tape in way of the pressure tape and air release tape. Little or no oil leakage was observed around the bolt holes throughout the remainder of the tests.

Test of Compression Specimens

Since the value of E, Young's Modulus, and σ_y were of primary importance in the correlation of experimental data with predicted stress distribution and collapse pressure, it was immediately apparent to the authors that the values of E and σ_y for the material used should be determined experimentally rather than relying on handbook values which represent average data at best.

Four 2 1/2" wide semi-circular sections were cut from the original piece of aluminum tubing to provide the material from which tensile specimens were to be machined. The semi-circular sections had to be flattened before machining; therefore, the sections were annealed prior to the flattening process to prevent cracking. Annealing was carried out as recommended by the Metals Handbook (5). To insure a reasonably complete anneal, the semi-circular

During the short testing period all leakage around the
 half coils was noted. A strip of rubber electrical tape
 3/4" in width was glued to the inner circumference of the
 spaced ring. The edges of this seal joined each other at the
 center of the test chamber and were pressed tightly against
 the flanges when pressure was applied. Holes were cut in
 the electrical tape in way of the pressure taps and air
 release taps. Little or no oil leakage was observed around
 the coil coils throughout the remainder of the tests.

Test of Compression Specimens

Other the value of σ , Young's modulus, and ν were
 of primary importance in the correlation of experimental
 data with predicted stress distribution and collapse pressure,
 it was immediately apparent to the authors that the values
 of E and ν for the material used should be determined ex-
 perimentally rather than relying on handbook values which
 represent average data at best.
 Four 3 1/2" wide semi-circular sections were cut from
 the original piece of aluminum tubing to provide the
 material from which tensile specimens were to be machined.
 The semi-circular sections had to be flattened before
 machining; therefore, the sections were annealed prior to
 the flattening process to prevent cracking. Annealing was
 carried out as recommended by the Metals Handbook (5).
 To insure a reasonably complete anneal, the semi-circular

sections were heated for two hours at 750°F, then slowly cooled (not faster than 50°F per hour) until a temperature of 500° was reached. Below 500°F the rate of cooling was not important and the sections were removed from the furnace and allowed to air cool to room temperature.

The sections were then flattened in a mechanical press. No difficulties were experienced in the flattening operation. Cracks did not occur in the sections and very little spring-back was observed when the upper head of the press was lifted from the sections. From all indications, the aluminum alloy was sufficiently annealed by the process described above.

To obtain uniformity of physical properties in the flattened sections, they were heat treated along with the ring sections as described in the material related to the manufacture of the rings.

Unfortunately the flat bar specimens were not quite as true as the authors had anticipated, and it was feared that appreciable bending stresses might be introduced in the tensile specimens during the application of loads. Initial unfairness could have been eliminated by machining the flat surfaces; however, this procedure would not have been too practical since the original wall thickness of the aluminum tubing was only 0.250". A further reduction in thickness would have increased the difficulty of mounting

sections were heated for 24 hours at 750°. They slowly cooled (the faster they cooled the better) until a temperature of 500° was reached. Below 500° the rate of cooling was not so rapid and the sections were cooled from the furnace and allowed to cool to room temperature.

The sections were then flattened in a mechanical press. No difficulties were experienced in the flattening operation. Cracks did not occur in the sections and very little deformation was observed when the upper head of the press was lifted from the sections. From all indications, the aluminum alloy was satisfactorily annealed by the process described above.

To obtain uniformity of physical properties in the flattened sections, they were heat treated along with the ring sections as described in the material related to the heat treatment of the rings.

Unfortunately, the flat bar specimens were not quite as long as the rings had anticipated, and it was found that approximately similar stresses could be introduced in the tensile specimens during the flattening process. Initial determinations have been obtained by bending the flat rectangular members, and preliminary work has been completed since the original work was completed at the aluminum foundry was 0.005". A further reduction in thickness would have introduced the difficulty of obtaining

of Tensometers. It was then decided to use three compression specimens cut from the flattest portions of the section. Since the length of the specimens required was of the order of 2.5", no appreciable difficulty was experienced in finding planar portions of the sections for this purpose.

Three compression specimens were manufactured by the Department of Mechanical Engineering machine shop.

Specification for compression test specimens:

Using the same material from which the rings were manufactured, three specimens are to be cut and milled to the dimensions of 0.84 x 2.53 inches, these dimensions conforming to the proportions of 1 to 3 as recommended by ASTM standards.

The compression specimen test block and two Type A Huggenberger Tensometers were furnished by the Department of Mechanical Engineering, M.I.T. Load was applied to the block by means of a 10,000 pound capacity loading machine.

As recommended by the Metals Handbook, the Tensometers were mounted on opposite sides of the test specimen. The gage points were located symmetrically with respect to the middle of the length of the specimen and not closer to the end of the specimen than a distance approximately equal to the width of the specimen.

It was found necessary to reset the Huggenberger Tensometers at least four times during each specimen run.

of Tensometers. It was then decided to use three compression specimens and from the flattest portions of the section. Since the length of the specimens required was of the order of 7.5", an appreciable difficulty was experienced in finding planar portions of the sections for this purpose. Three compression specimens were manufactured by the

Department of Mechanical Engineering machine shop.

Specifications for compression test specimens:

Using the same material from which the rings were manufactured, three specimens are to be cut and rolled to the dimensions of 0.84 x 2.53 inches, these dimensions conforming to the proportions of 1 to 3 as recommended by

ASTM standards.

The compression specimen test block and two type A Hydrostatic Tensometers were furnished by the Department of Mechanical Engineering, M.I.T. Load was applied to the block by means of a 10,000 pound capacity loading machine. As recommended by the Metals Handbook, the Tensometers were mounted on opposite sides of the test specimen. The gage points were located symmetrically with respect to the middle of the length of the specimen and not closer to the end of the specimen than a distance approximately equal to the width of the specimen.

It was found necessary to treat the Hydrostatic Tensometers at least four times during each specimen run.

APPENDIX B

Sample Calculations and Summary of Data

Calculation of R_o , Radius to Outer Circumference

The radius to the outer circumference of each ring was measured indirectly by fitting a wire of known diameter around the outer surface of the ring, scribing the wire in position, and then measuring the distance between the scribe marks on a 36" rule.

$$\text{Then } R_o = \frac{l}{2\pi} - d, \text{ where}$$

R_o = radius to outer circumference of the ring
in inches.

l = length of wrapper wire measured between
scribe marks in inches.

d = diameter of wrapper wire in inches.

Thus for Ring No. 1:

$$R_o = \frac{28.391}{2\pi} - 0.01"$$

$$R_o = 4.509"$$

The value of R_o used in the remaining calculations is the average of all measured radii.

Compression Test

The strains given by opposite Huggenberger Tensometers at a specified load were averaged to avoid errors due to bending. The stress corresponding to this strain was obtained by dividing the load by the original cross sectional area of the specimen.

Appendix 2

Sample Calculations and Summary of Data

Calculation of R₀ Values as Outer Circumference

The radius to the outer circumference of each lead was measured individually by passing a wire of known diameter around the outer surface of the coil, adding the wire in position, and then measuring the distance between the points where on a 36" rule.

$$R_0 = \frac{l}{2\pi} = 1.5708 \text{ inches}$$

R_0 = Radius to outer circumference of the coil in inches.

l = length of wrapper wire measured between center marks in inches.

d = diameter of wrapper wire in inches.

Thus for lead 1:

$$R_0 = \frac{55.531}{2\pi} = 8.871$$

$$R_0 = 8.871$$

The value of R_0 used in the remaining calculations

in the system of all measured radii.

Correcting Term

The values given by opposite instruments; temperature of a corrected lead were averaged to avoid errors due to position. The values corresponding to this system are up-
dated by division and lead by the actual cross-sectional area of the specimen.

The following calculation applies in the case of
Specimen No. 1 at a load of 4500 lbs.

Data - Specimen No. 1

Huggenberger No. 2201				Huggenberger No. 2195		
Calibration Factor 1040				Calibration Factor 1051		
Load lbs.	Rdg.	Rdg	Rdg	Rdg	Rdg	Rdg
0	1.50			1.5		
500	1.30	-0.20	-0.20	1.24	-0.26	-0.26
1000	1.09	-0.21	-0.41	1.00	-0.24	-0.30
1500	0.86	-0.23	-0.64	0.74	-0.26	-0.76
2000	0.63	-0.23	-0.87	0.50	-0.24	-1.00
2500	0.43	-0.20	-1.07	0.28	-0.22	-1.22
3000	0.20	-0.23	-1.30	0.05	-0.23	-1.45
Reset	1.50			1.50		
3500	1.33	-0.17	-1.47	1.28	-0.22	-1.67
4000	1.11	-0.22	-1.69	1.04	-0.24	-1.91
4500	0.89	-0.22	-1.91	0.81	-0.23	-2.14

$$\text{Strain} = \frac{\text{Tensometer Reading}}{\text{Calibration Factor}} \text{ in inches/inch}$$

$$\epsilon_1 = \text{Strain in inches/inch given by Tensometer No. 2201}$$

$$\epsilon_2 = \text{Strain in inches/inch given by Tensometer No. 2195}$$

$$\epsilon = \text{Average measured strain in micro-inches/inch}$$

$$\epsilon_1 = \frac{-1.91}{1040} = -0.001836 \text{ inches/inch}$$

$$\epsilon_2 = \frac{-2.14}{1051} = -0.002036 \text{ inches/inch}$$

$$\epsilon = \frac{(-0.001836) + (-0.002036)}{2} \times 10^6 = -1936 \text{ micro-inches/inch}$$

$$\sigma = \frac{L}{txw}$$

$$\sigma = \text{stress in psi.}$$

$$L = \text{load in lbs.}$$

The following calculation applies to the case of
Specimen No. 1 at a load of 4500 lbs.

Note - Specimen No. 1

Load lbs.	Strain in. in.	Strain mm. mm.	Strain mm. mm.	Strain mm. mm.	Strain mm. mm.
0	0	0	0	0	0
500	0.000	0.000	0.000	0.000	0.000
1000	0.000	0.000	0.000	0.000	0.000
1500	0.000	0.000	0.000	0.000	0.000
2000	0.000	0.000	0.000	0.000	0.000
2500	0.000	0.000	0.000	0.000	0.000
3000	0.000	0.000	0.000	0.000	0.000
3500	0.000	0.000	0.000	0.000	0.000
4000	0.000	0.000	0.000	0.000	0.000
4500	0.000	0.000	0.000	0.000	0.000

$$\text{Strain} = \frac{\text{Transverse Reading}}{\text{Calibration Factor}}$$

$$E = \frac{\text{Strain in inches/inch given by Transducer}}{\text{No. 2501}}$$

$$E = \frac{\text{Strain in inches/inch given by Transducer}}{\text{No. 2192}}$$

$$E = \frac{\text{Average measured strain in microstrain/inch}}$$

$$E = \frac{-1.91}{1940} = -0.0009845 \text{ inches/inch}$$

$$E = \frac{-2.14}{1021} = -0.0021058 \text{ inches/inch}$$

$$E = \frac{(-0.0009845 + (-0.0021058)) \times 10^6}{2} = -154.5 \text{ ksi}$$

$$G = \frac{1}{2}$$

$$G = \text{Strain in psi}$$

$$G = \text{Load in lbs.}$$

t = thickness of specimen in inches

w = width of specimen in inches

$$\sigma = \frac{4,500}{(0.201)(0.841)} = 20,500 \text{ psi}$$

The results of the Compression Test calculations are presented as experimental points in Figure III.

Circumferential Strains

Except for the case of Ring No. 2 all predicted circumferential strain distributions were computed directly from equation (3).

$$\epsilon = \frac{P R_o}{E h} + \frac{6 P R u_o h \cos 2 \theta}{E h^3 - 4 P R^3}$$

All factors are defined in SYMBOLS AND ABBREVIATIONS, page vi.

For Ring No. 7, $u_o = 0.1535"$; then for $P = 240 \text{ psi}$.

$$\epsilon = \frac{(240)(4.508)}{(0.250)(10.8 \times 10^6)} + \frac{(6)(240)(4.383)(0.1535)(0.250) \cos 2 \theta}{(10.8 \times 10^6)(0.250)^3 - (4)(240)(4.383)^3}$$

$$\epsilon = (400.7) + (2754.8 \cos 2 \theta) \text{ micro-inches/inch}$$

θ	$\cos 2 \theta$	$2754.8 \cos 2 \theta$	ϵ
$0^\circ \text{ \& } 180^\circ$	+1	+2754.8	+3155.5
$30^\circ \text{ \& } 210^\circ$	+1/2	+1377.4	+1778.1
$60^\circ \text{ \& } 240^\circ$	-1/2	-1377.4	-976.7
$90^\circ \text{ \& } 270^\circ$	-1	-2754.8	-2354.1
$120^\circ \text{ \& } 300^\circ$	-1/2	-1377.4	-976.7
$150^\circ \text{ \& } 330^\circ$	+1/2	+1377.4	+1778.1

r = thickness of specimen in inches

w = width of specimen in inches

$$d = \frac{4,500}{(0.25)(0.041)} = 30,500 \text{ psi}$$

The results of the compression test calculations are

presented as suggested values in Table III.

Circumferential Strain

Except for the case of Ring No. 2 all predicted circumferential strain distributions were compared directly

from equation (3).

$$\epsilon = \frac{r}{m} + \frac{D}{m} \cos n \cos \theta$$

All results are defined in SYMBOLS AND ABBREVIATIONS, page vi.

For Ring No. 7, $\theta = 0.155^\circ$; then for $r = 240$ psi.

$$\epsilon = \frac{(0.155)(10.8 \times 10^6)}{(10.8 \times 10^6)(0.25)(0.041)} + \frac{(0.155)(10.8 \times 10^6)(0.25)(0.041)}{(10.8 \times 10^6)(0.25)(0.041)} \cos \theta$$

$\epsilon = (10.8 \times 10^6)(0.25)(0.041) \cos \theta$ (circumferential)

θ	$\cos \theta$	ϵ
0° & 180°	1	$+17,544.0$
30° & 210°	$+0.87$	$+15,277.8$
60° & 240°	-0.5	$-8,772.4$
90° & 270°	-1	$-17,544.0$
120° & 300°	-0.5	$-8,772.4$
150° & 330°	$+0.87$	$+15,277.8$

Since ϵ has been defined as a compression strain the plotted values become:

Circumferential Position	Strain in micro-inches/inch
0° & 180°	-3155.5
30° & 210°	-1788.1
60° & 240°	+976.7
90° & 270°	+2354.1
120° & 300°	+976.7
150° & 330°	-1778.1

The circumferential strain distribution for Ring No. 2 was computed using the measured out-of-roundness at each station instead of $u_0 \cos 2 \theta$.

$$\epsilon = \frac{P R_0}{E h} + \frac{6 P R u h}{E h^3 - 4 P R^3} \quad (5)$$

Data:

Circumferential Position	u	Sign as determined by equation (5)
0° & 180°	0.0075"	-
30° & 210°	0.0800"	+
50° & 230°	0.0125"	+
80° & 260°	0.0065"	+
110° & 290°	0.0045"	-
150° & 330°	0.0125"	-

at $P = 400$ psi.

Since ϵ has been defined as a compression strain the plotted

values become:

Circumferential Position Strain in zero-increase/line

0° & 180°	-0.0025
30° & 210°	-0.0018
60° & 240°	+0.0007
90° & 270°	+0.0014
120° & 300°	+0.0017
150° & 330°	-0.0018

The circumferential strain distribution for line No. 2 was computed using the measured out-of-roundness at each station instead of r_0 for r .

$$\epsilon = \frac{r_0}{r} + \frac{r_0}{r} \left(\frac{r_0}{r} - 1 \right) \quad (2)$$

Date:

Circumferential Position Strain as determined by

equation (2)

0° & 180°	0.0015
30° & 210°	0.0000
60° & 240°	0.0115
90° & 270°	0.0025
120° & 300°	0.0045
150° & 330°	0.0125

at $r = 400$ mil.

$$\epsilon = \frac{(400)(4.508)}{(0.250)(10.8 \times 10^6)} + \frac{(6)(400)(4.383)(0.250) u}{(10.8 \times 10^6)(0.250)^3 - (4)(400)(4.383)^3}$$

$$\epsilon = 667.85 + 83353.4 u \text{ micro-inches/inch}$$

Substituting data:

Circumferential Position	Plotted Strain in micro-inches/inch
0° & 180°	-42.7
30° & 210°	-1334.65
50° & 230°	-1709.77
80° & 260°	-1209.63
110° & 290°	-292.76
150° & 330°	+374.08

A summary of all such calculated values is given in the form of curves, see Figures V, VIII, XI, XIV, and XVII.

Maximum Strains

The maximum strains at specified pressures were computed using equation (3). $\cos 2 \theta$ becomes +1 for 0° and 180° positions and -1 for 90° and 270° positions.

$$\epsilon = \frac{P R_0}{E h} + \frac{6 P R u_0 h}{E h^3 - 4 P R^3} \quad (6)$$

$$\epsilon = \frac{P R_0}{E h} - \frac{6 P R u_0 h}{E h^3 - 4 P R^3} \quad (7)$$

ϵ for a ring of given u_0 is computed from equations (6) and (7) at several increments of pressure. For Ring No. 5, $u_0 = 0.070$ "; then at 50 psi.:

$$3 = \frac{(10.50)(10.50 \times 10^6)}{(10.50)(10.50 \times 10^6)} + \frac{(10.50)(10.50 \times 10^6)}{(10.50)(10.50 \times 10^6)}$$

$$3 = 10.50 + 10.50 \times 10^6$$

Calculation of

Electrochemical

in photo-chemical

10.50	10.50
10.50	10.50
10.50	10.50
10.50	10.50
10.50	10.50
10.50	10.50

A summary of all such calculated values is given in the form of curves, see Figures V, VIII, XI, XIV, and XVII.

References

The various values of potential pressure were measured using a potentiometer (1). The values of E_0 and E_{100} are given in Table I for 10^6 and 10^7 potentials.

$$(6) \quad \frac{E_{100} - E_0}{E_{100} - E_0} = \frac{E_{100} - E_0}{E_{100} - E_0}$$

$$(7) \quad \frac{E_{100} - E_0}{E_{100} - E_0} = \frac{E_{100} - E_0}{E_{100} - E_0}$$

For a line of given E_0 in the region of the curve (6) and (7) of several increments of potential, the area A_0 is $A_0 = 0.001$ from 0 to 1.

ϵ at 0° and 180°

$$\epsilon = \frac{(50)(4.508)}{(10.8 \times 10^6)(0.250)} + \frac{(6)(50)(4.383)(0.070)(0.250)}{(10.8 \times 10^6)(0.250)^3 - (4)(50)(4.383)^3}$$

$\epsilon = 234.9$ micro-inches/inch (Plotted with - sign)

ϵ at 90° and 270°

$$\epsilon = \frac{(50)(4.508)}{(10.8 \times 10^6)(0.250)} - \frac{(6)(50)(4.383)(0.070)(0.250)}{(10.8 \times 10^6)(0.250)^3 - (4)(50)(4.383)^3}$$

$\epsilon = -68.0$ micro-inches/inch (Plotted with + sign)

Pressure psi.	Plotted value at 0° and 180°	Plotted value at 90° and 270°
50	-234.9	+68.0
100	-507.7	+173.8
150	-834.3	+333.5
200	-1,241.7	+573.9
250	-1,778.1	+943.4
300	-2,539.9	+1,538.2
350	-3,750.7	+2,582.1
400	-6,077.3	+4,741.7

A summary of all such calculated values is given in the form of plots, see Figures XX-XXVI.

Collapse Pressures

The curves of predicted collapse pressures were derived indirectly by first plotting equation (4). Stresses at various pressures were computed for several values of u_0 .

A cross curve of predicted collapse pressures was then drawn using the values given by the intersection of a single ordinate 35,000 psi. or 41,000 psi. with the curves of stress versus pressure.

To compute one point for stress-pressure curves:

$$u_0 = 0.050", P = 50 \text{ psi.}, h = 0.250"$$

$$\sigma = \frac{P R_0}{h} + \frac{6 P R u_0 h E}{E h^3 - 4 P R^3}$$

$$\sigma = \frac{(50)(4.508)}{(0.250)} + \frac{(6)(50)(0.050)(4.383)(0.250)(10.8 \times 10^6)}{(10.8 \times 10^6)(0.250)^3 - (4)(50)(4.383)^3}$$

$$\sigma = 2,090 \text{ psi.}, \text{ maximum compressive stress}$$

Similarly:

Pressure psi.	σ psi.
100	4,431
200	10,609
250	15,005
300	21,139
350	30,737
400	50,206

Values for $\sigma_{\text{max.}}$ were computed at several increments of pressure for the following values of u_0 : 0.050", 0.100", 0.150", 0.200", and 0.250". The procedure was repeated using $h = h_{\text{min.}} = 0.243"$.

The results of these computations are presented as curves, Figure XXXI. Shown in the same Figure are ordinates

A cross curve of isotherms obtained from the data drawn using the values given by the intersection of a single ordinate 25,000 psi. or 41,000 psi. with the curve of isotherms pressure.

The average and point for stress-pressure curves:

$$p_0 = 0.050, \quad T = 50 \text{ psi.}, \quad n = 0.250$$

$$d = \frac{1}{n} \left(\frac{p_0}{p} + \frac{p}{p_0} \right) \frac{p}{p_0 - p}$$

$$d = \frac{(50)(4.5051) + (4)(50)(0.0501)(0.250)(10.8 \times 10^8)}{(0.250) (10.8 \times 10^8) (10.250) - (4)(50)(4.5051)}$$

$$d = 2,020 \text{ psi.}, \text{ maximum compressive stress}$$

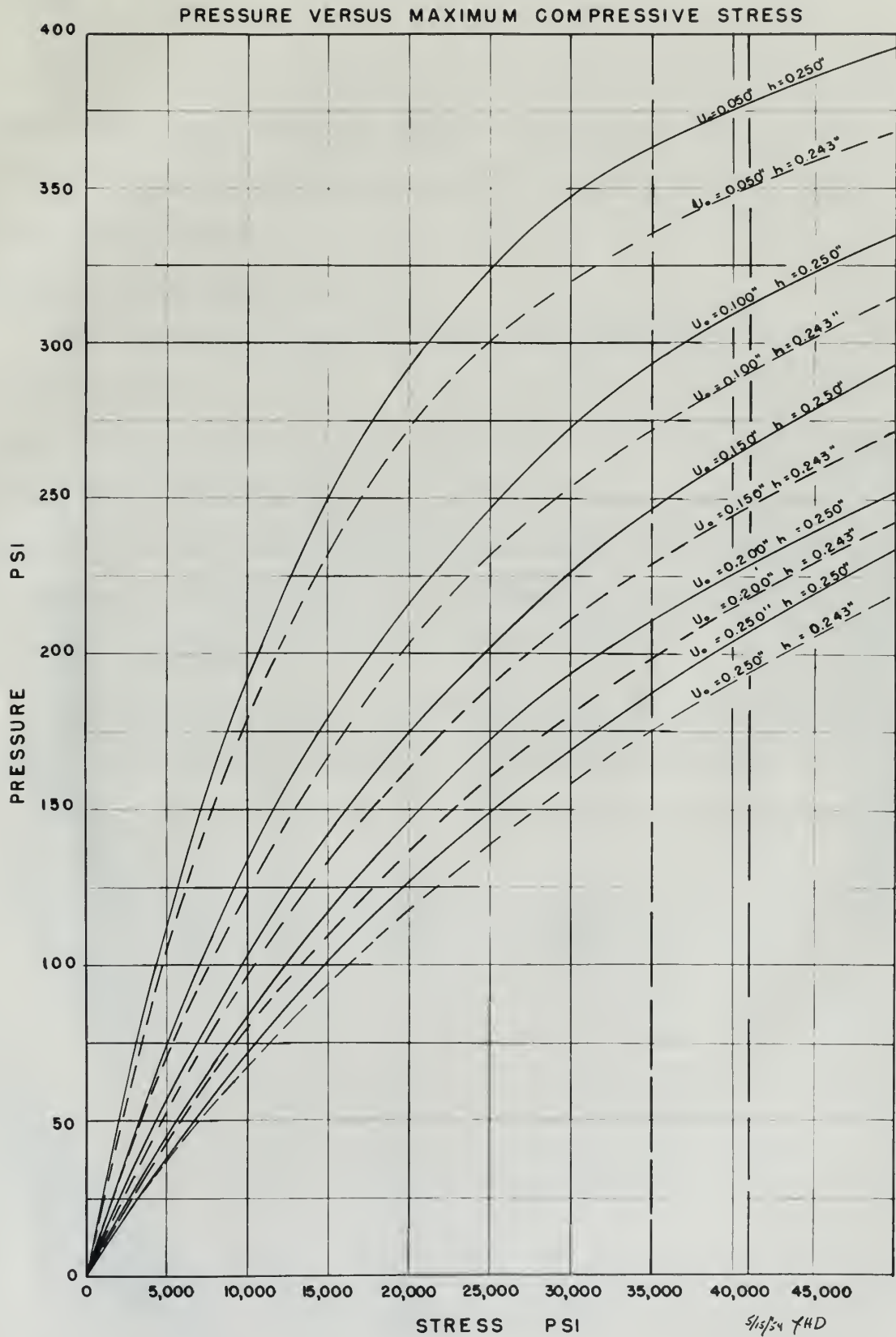
Stress:

Pressure psi.	d psi.
100	4,431
200	10,602
300	15,002
400	20,132
500	25,737
600	30,728

Values for d are, with rounded to several hundredths of pressure for the following values of p : 0.050, 0.100, 0.150, 0.200, and 0.250. The pressure was selected using $n = 0.250$.

The results of these calculations are presented as curves, Figure 111. Some of the data points are indicated

FIGURE XXXI



at 35,000 psi. and 41,000 psi. The intersection of pressure-stress curve with these ordinates determined the points through which the predicted collapse pressure curves were drawn; see Figures XXVII and XXVIII.

Experimental Strain Readings

The strain data was checked for significant errors by plotting measured strains versus pressure for each strain gage. There were no marked deviations from the fair curves which were drawn through the points. These curves were also used to determine the experimental points shown in the plots of circumferential strain distribution in Figures V, VIII, XI, XIV, and XVII.

The measured values of strain shown in Figures IV and XX-XXVI were taken directly from the original data.

of 25,000 psi. was 41,000 psi. The intersection of pressure
 curves with these isotherms determined the points
 through which the predicted isotherm pressure curves were
 drawn, see figures XIV and XV.

Experimental Strain Results

The strain data was checked for significant errors by
 plotting measured strain versus pressure for each strain
 gage. There were no noted deviations from the full curves
 which were drawn through the points. These curves were also
 used to determine the experimental points shown in the plots
 of circumferential strain distribution in figures V, VII,
 XI, XIV, and XV.

The measured values of strain shown in figure IV are
 11-15% were taken directly from the original data.

APPENDIX C

Original Data

Table I is a tabulation of the data taken during the test of the three compression specimens.

Table II is a record of the data recorded during the investigation of the effects of lubricants and gasket material upon the strain distribution around the inner circumference of the aluminum rings.

Tables III - XI contain data relating to physical dimensions, circumferential strain distribution, collapse pressures and diametral readings for each ring.

APPENDIX C

Original Data

Table I is a tabulation of the data taken during the

test of the three compression specimens.

Table II is a record of the data recorded during the

investigation of the effects of lubrication and heating

material upon the strain distribution around the inner

circumference of the specimen rings.

Table III - XI contain data relating to physical

properties, circumferential strain distribution, collapse

pressure and lateral readings for each ring.

Table I
Data for Stress-Strain Curve in Compression

Load lbs.	Specimen #1 H#2201	Specimen #2		H#2201	H#2195	Specimen #3	
		H#2195	H#2201			H#2201	H#2195
0	1.50	1.50	1.60	1.58	1.60	1.58	1.60
500	1.30	1.24					
1000	1.09	1.00	1.40	0.94	0.95	1.26	0.95
1500	0.86	0.74					
2000	0.63	0.50	0.96	0.41	0.55	0.82	0.55
2500	0.43	0.28					
3000	0.20	0.05	0.59	0.03	0.19	0.37	0.19
Reset to	1.50	1.50	1.55	1.62	1.49	1.59	1.49
3500	1.33	1.28					
4000	1.11	1.04	1.05	1.15	1.06	1.15	1.06
4500	0.89	0.81					
5000	0.66	0.57	0.62	0.69	0.60	0.70	0.60
5500	0.45	0.35	0.45	0.52	0.35	0.46	0.35
6000	0.21	0.10	0.22	0.29	0.10	0.24	0.10
Reset to	1.50	1.50	1.60	1.55	1.53	1.54	1.53
6500	1.30	---	1.36	1.30	1.43	1.40	1.43
7000	1.05	---	1.10	1.03	1.18	1.11	1.18
7500	0.82	---	0.90	0.80	0.84	0.78	0.84
8000	0.56	---	0.65	0.55 Reset to	0.50	0.38	0.40
8500	0.25	---	0.35	0.19 Reset to	1.50	1.50	1.50
Reset to	1.50	1.50	1.50	1.60	1.07	1.10	1.07
9000	1.01	1.12	1.34	1.40			
9110				Reset to			
9500	0.01	0.30	0.54	0.36	-0.12	0.11	-0.12
Reset to	1.50	1.50	1.55	1.45	1.55	1.47	1.55
9630	0.02	0.23			0.10	0.29	0.10
9650			0.39	-0.15			

TABLE 1 SUMMARY OF DATA FOR THE STUDY

Item	Location	Altitude (ft)	Area (sq ft)	Volume (cu ft)	Weight (lb)	Notes
1	Point A	100.0	100.0	100.0	100.0	
2	Point B	200.0	200.0	200.0	200.0	
3	Point C	300.0	300.0	300.0	300.0	
4	Point D	400.0	400.0	400.0	400.0	
5	Point E	500.0	500.0	500.0	500.0	
6	Point F	600.0	600.0	600.0	600.0	
7	Point G	700.0	700.0	700.0	700.0	
8	Point H	800.0	800.0	800.0	800.0	
9	Point I	900.0	900.0	900.0	900.0	
10	Point J	1000.0	1000.0	1000.0	1000.0	
11	Point K	1100.0	1100.0	1100.0	1100.0	
12	Point L	1200.0	1200.0	1200.0	1200.0	
13	Point M	1300.0	1300.0	1300.0	1300.0	
14	Point N	1400.0	1400.0	1400.0	1400.0	
15	Point O	1500.0	1500.0	1500.0	1500.0	
16	Point P	1600.0	1600.0	1600.0	1600.0	
17	Point Q	1700.0	1700.0	1700.0	1700.0	
18	Point R	1800.0	1800.0	1800.0	1800.0	
19	Point S	1900.0	1900.0	1900.0	1900.0	
20	Point T	2000.0	2000.0	2000.0	2000.0	
21	Point U	2100.0	2100.0	2100.0	2100.0	
22	Point V	2200.0	2200.0	2200.0	2200.0	
23	Point W	2300.0	2300.0	2300.0	2300.0	
24	Point X	2400.0	2400.0	2400.0	2400.0	
25	Point Y	2500.0	2500.0	2500.0	2500.0	
26	Point Z	2600.0	2600.0	2600.0	2600.0	
27	Point AA	2700.0	2700.0	2700.0	2700.0	
28	Point AB	2800.0	2800.0	2800.0	2800.0	
29	Point AC	2900.0	2900.0	2900.0	2900.0	
30	Point AD	3000.0	3000.0	3000.0	3000.0	
31	Point AE	3100.0	3100.0	3100.0	3100.0	
32	Point AF	3200.0	3200.0	3200.0	3200.0	
33	Point AG	3300.0	3300.0	3300.0	3300.0	
34	Point AH	3400.0	3400.0	3400.0	3400.0	
35	Point AI	3500.0	3500.0	3500.0	3500.0	
36	Point AJ	3600.0	3600.0	3600.0	3600.0	
37	Point AK	3700.0	3700.0	3700.0	3700.0	
38	Point AL	3800.0	3800.0	3800.0	3800.0	
39	Point AM	3900.0	3900.0	3900.0	3900.0	
40	Point AN	4000.0	4000.0	4000.0	4000.0	
41	Point AO	4100.0	4100.0	4100.0	4100.0	
42	Point AP	4200.0	4200.0	4200.0	4200.0	
43	Point AQ	4300.0	4300.0	4300.0	4300.0	
44	Point AR	4400.0	4400.0	4400.0	4400.0	
45	Point AS	4500.0	4500.0	4500.0	4500.0	
46	Point AT	4600.0	4600.0	4600.0	4600.0	
47	Point AU	4700.0	4700.0	4700.0	4700.0	
48	Point AV	4800.0	4800.0	4800.0	4800.0	
49	Point AW	4900.0	4900.0	4900.0	4900.0	
50	Point AX	5000.0	5000.0	5000.0	5000.0	

Table II
Strain Data for Preliminary Test Ring

Lubricant #2		Gasket #1		Strains in Micro-inch per inch at Positions indicated									
Pressure	0°	30°	60°	90°	120°	150°	180°	210°	240°	270°	300°	330°	
psi													
0	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000
99	6820	6920	---	6800	6695	6800	6890	6900	6830	6875	6780	---	---
200	6615	6830	---	6600	6375	6580	6765	6800	6660	6750	6570	---	---
303	6375	6800	---	6385	5920	6270	6660	6790	6580	6630	6215	---	---
403	6120	7650	---	6190	5200	5690	6510	7010	6780	6540	5630	---	---

Lubricant #1		Gasket #1											
Pressure	0°	30°	60°	90°	120°	150°	180°	210°	240°	270°	300°	330°	
psi													
0	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000
99	6770	6900	---	6815	6680	6760	6855	6890	6850	6885	6765	---	---
200	6530	6870	---	6650	6285	6460	6700	6840	6770	6785	6480	---	---
303	6270	7100	---	6400	5500	5875	6635	7055	6860	6610	5870	---	---

Lubricant #1		Gasket #2											
Pressure	0°	30°	60°	90°	120°	150°	180°	210°	240°	270°	300°	330°	
psi													
0	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000	7000
99	6760	6880	---	6850	6700	6760	6840	6880	6860	6905	6790	---	---
200	6520	6810	---	6670	6350	6500	6700	6800	6735	6800	6550	---	---
303	6260	6840	---	6460	5840	6120	6550	6820	6730	6695	6150	---	---
403	6070	7430	---	6150	4810	5305	6450	7345	7180	6540	7240	---	---

Table III
Dimensions and Strain Data for Ring #1

Angle	h in.	b in	0	Strains in micro-inches per inch at pressures indicated				
				99	200	303	403	453
0°	0.247	0.444	11,000	10,780	10,610	10,500	10,270	10,120
30°	0.250	0.445	11,000	10,805	10,670	10,515	10,285	10,160
60°	0.253	0.445	11,000	10,850	10,680	10,460	10,305	10,215
90°	0.261	0.444	11,000	10,860	10,640	10,370	10,200	10,120
120°	0.261	0.445	11,000	10,820	10,590	10,310	10,090	9,970
150°	0.256	0.444						
180°	0.247	0.445	11,000	10,860	10,790	10,790	10,730	10,710
210°	0.242	0.446	11,000	10,830	10,750	10,670	10,545	10,460
240°	0.243	0.445	11,000	10,805	10,560	10,255	9,910	9,730
270°	0.243	0.444	11,000	10,800	10,480	10,120	9,830	9,715
300°	0.246	0.444	11,000	10,870	10,700	10,530	10,470	10,430
330°	0.248	0.445	11,000	10,840	10,705	10,605	10,530	10,460

Collapse Pressure = 450 psi

Outside Radius = 4.509"

Diameter readings in inches	
0-180°	9.012
15-195°	9.008
30-210°	9.008
45-225°	9.016
60-240°	9.027
75-255°	9.035
	90-270°
	105-285°
	120-300°
	135-315°
	150-330°
	165-345°

On 19 July 2001

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^aOR
2.06E-07

Case	Age	Sex	Duration	Findings
1	44.5	M	1.5 yr	Normal
2	44.5	M	1.5 yr	Normal
3	44.5	M	1.5 yr	Normal
4	44.5	M	1.5 yr	Normal
5	44.5	M	1.5 yr	Normal
6	44.5	M	1.5 yr	Normal
7	44.5	M	1.5 yr	Normal
8	44.5	M	1.5 yr	Normal
9	44.5	M	1.5 yr	Normal
10	44.5	M	1.5 yr	Normal
11	44.5	M	1.5 yr	Normal
12	44.5	M	1.5 yr	Normal
13	44.5	M	1.5 yr	Normal
14	44.5	M	1.5 yr	Normal
15	44.5	M	1.5 yr	Normal
16	44.5	M	1.5 yr	Normal
17	44.5	M	1.5 yr	Normal
18	44.5	M	1.5 yr	Normal
19	44.5	M	1.5 yr	Normal
20	44.5	M	1.5 yr	Normal
21	44.5	M	1.5 yr	Normal
22	44.5	M	1.5 yr	Normal
23	44.5	M	1.5 yr	Normal
24	44.5	M	1.5 yr	Normal
25	44.5	M	1.5 yr	Normal
26	44.5	M	1.5 yr	Normal
27	44.5	M	1.5 yr	Normal
28	44.5	M	1.5 yr	Normal
29	44.5	M	1.5 yr	Normal
30	44.5	M	1.5 yr	Normal
31	44.5	M	1.5 yr	Normal
32	44.5	M	1.5 yr	Normal
33	44.5	M	1.5 yr	Normal
34	44.5	M	1.5 yr	Normal
35	44.5	M	1.5 yr	Normal
36	44.5	M	1.5 yr	Normal
37	44.5	M	1.5 yr	Normal
38	44.5	M	1.5 yr	Normal
39	44.5	M	1.5 yr	Normal
40	44.5	M	1.5 yr	Normal
41	44.5	M	1.5 yr	Normal
42	44.5	M	1.5 yr	Normal
43	44.5	M	1.5 yr	Normal
44	44.5	M	1.5 yr	Normal
45	44.5	M	1.5 yr	Normal
46	44.5	M	1.5 yr	Normal
47	44.5	M	1.5 yr	Normal
48	44.5	M	1.5 yr	Normal
49	44.5	M	1.5 yr	Normal
50	44.5	M	1.5 yr	Normal
51	44.5	M	1.5 yr	Normal
52	44.5	M	1.5 yr	Normal
53	44.5	M	1.5 yr	Normal
54	44.5	M	1.5 yr	Normal
55	44.5	M	1.5 yr	Normal
56	44.5	M	1.5 yr	Normal
57	44.5	M	1.5 yr	Normal
58	44.5	M	1.5 yr	Normal
59	44.5	M	1.5 yr	Normal
60	44.5	M	1.5 yr	Normal
61	44.5	M	1.5 yr	Normal
62	44.5	M	1.5 yr	Normal
63	44.5	M	1.5 yr	Normal
64	44.5	M	1.5 yr	Normal
65	44.5	M	1.5 yr	Normal
66	44.5	M	1.5 yr	Normal
67	44.5	M	1.5 yr	Normal
68	44.5	M	1.5 yr	Normal
69	44.5	M	1.5 yr	Normal
70	44.5	M	1.5 yr	Normal
71	44.5	M	1.5 yr	Normal
72	44.5	M	1.5 yr	Normal
73	44.5	M	1.5 yr	Normal
74	44.5	M	1.5 yr	Normal
75	44.5	M	1.5 yr	Normal
76	44.5	M	1.5 yr	Normal
77	44.5	M	1.5 yr	Normal
78	44.5	M	1.5 yr	Normal
79	44.5	M	1.5 yr	Normal
80	44.5	M	1.5 yr	Normal
81	44.5	M	1.5 yr	Normal
82	44.5	M	1.5 yr	Normal
83	44.5	M	1.5 yr	Normal
84	44.5	M	1.5 yr	Normal
85	4			

Table IV

Dimensions and Strain Data for Ring #2

Angle	h in.	b in.	Strains in micro-inches per inch at pressures indicated			
			0	99	200	303 403
0°	0.244	0.448	11,000	10,820	10,690	10,590 11,670
30°	0.246	0.447	11,000	10,810	10,605	10,320 9,550
60°	0.249	0.448	11,000	10,780	10,520	10,100 8,030
90°	0.247	0.448	11,000	10,765	10,475	10,090 8,640
120°	0.250	0.448	11,000			
150°	0.252	0.448	11,000	10,800	10,580	10,250 8,850
180°	0.261	0.448	11,000	10,885	10,770	10,755 11,160
210°	0.262	0.448	11,000	10,850	10,750	10,840 12,335
240°	0.256	0.448	11,000	10,825	10,650	10,555 10,930
270°	0.247	0.448	11,000	10,920	10,885	11,040 12,720
300°	0.243	0.448	11,000	10,905	10,840	10,895 12,030
330°	0.243	0.448	11,000	10,790	10,540	10,220 9,580

Collapse Pressure = 425 psi

Diameter readings in inches.

0-180°	9.020	90-270°	9.042
15-195°	9.030	105-285°	9.027
30-210°	9.050	120-300°	9.020
45-225°	9.060	135-315°	9.013
60-240°	9.057	150-330°	9.010

Outside radius = 4.510"

Calculus: $\text{length} = 1'' \times 2\pi r$

Temperature, °C	Time, min	Temperature, °C	Time, min
50.0	0.5	100.0	0.5
75.0	0.5	125.0	0.5
100.0	0.5	150.0	0.5
125.0	0.5	175.0	0.5
150.0	0.5	200.0	0.5
175.0	0.5	225.0	0.5
200.0	0.5	250.0	0.5
225.0	0.5	275.0	0.5
250.0	0.5	300.0	0.5
275.0	0.5	325.0	0.5
300.0	0.5	350.0	0.5
325.0	0.5	375.0	0.5
350.0	0.5	400.0	0.5
375.0	0.5	425.0	0.5
400.0	0.5	450.0	0.5
425.0	0.5	475.0	0.5
450.0	0.5	500.0	0.5
475.0	0.5	525.0	0.5
500.0	0.5	550.0	0.5
525.0	0.5	575.0	0.5
550.0	0.5	600.0	0.5
575.0	0.5	625.0	0.5
600.0	0.5	650.0	0.5
625.0	0.5	675.0	0.5
650.0	0.5	700.0	0.5
675.0	0.5	725.0	0.5
700.0	0.5	750.0	0.5
725.0	0.5	775.0	0.5
750.0	0.5	800.0	0.5
775.0	0.5	825.0	0.5
800.0	0.5	850.0	0.5
825.0	0.5	875.0	0.5
850.0	0.5	900.0	0.5
875.0	0.5	925.0	0.5
900.0	0.5	950.0	0.5
925.0	0.5	975.0	0.5
950.0	0.5	1000.0	0.5

[illegible][illegible][illegible]

ANALYSIS The authors used descriptive statistics to analyze the data.

References

Table V
Dimensions and Strain Data for Ring #2

Angle	h in.	b in	0	99	200	303	353	403	423
0°	0.246	0.444	9,000	8,745	8,450	7,940	7,690	6,225	5,230
30°	0.249	0.443	9,000	8,730	8,385	7,830	7,550	6,280	6,080
60°	0.249	0.445	9,000	8,840	8,640	8,475	8,390	8,450	9,150
90°	0.251	0.444	9,000	8,950	8,960	9,130	9,240	9,480	11,360
120°	0.254	0.443	9,000	8,895	8,840	8,930	8,980	9,040	10,010
150°	0.261	0.444	9,000	8,815	8,640	8,460	8,360	8,180	7,520
180°	0.262	0.445	9,000	8,750	8,460	8,035	7,775	7,405	5,760
210°	0.256	0.445	9,000	8,770	8,490	8,070	7,830	7,530	6,410
240°	0.247	0.444	9,000	8,840	8,660	8,490	8,410	8,355	8,860
270°	0.243	0.445	9,000	8,940	8,940	9,060	9,110	9,370	11,220
300°	0.243	0.445	9,000	8,755	8,520	8,360	8,410	8,660	9,920
330°	0.244	0.445	9,000	8,850	8,735	8,650	8,570	8,280	7,530

Collapse Pressure = 430 psi.

Diameter readings in inches.

0.130°	9.072	90-270°	8.965
15-195°	9.071	105-285°	8.986
30-210°	9.061	120-300°	9.007
45-225°	9.041	135-315°	9.032
60-240°	9.017	150-330°	9.047
75-255°	8.983	165-345°	9.064

Outside Radius = 4.510"

TABLE
Characterized according to the amount

but I wish to say that I am not a

[illegible]

Fig. 4 - initial situation

ОТДЕЛЕНИЕ ВОЕННЫХ НАУК

8	975-06	270.0
8	702-592	170.0
8	130-300	120.0
8	112-311	110.0
8	120-130	110.0
8	102-202	100.0
8	100-200	100.0

Table VI
Dimensions and Strain Data for Ring # 4

Angle	h in.	b in	Strains in micro-inches per inch at pressures indicated						
			0	99	150	200	252	303	353
0°	0.261	0.445	7,000	6,670	6,460	6,170	5,720	5,140	378
30°	0.254	0.446	7,000	6,765	6,600	6,410	6,140	5,800	3,340
60°	0.251	0.446	7,000	6,900	6,850	6,855	6,925	7,090	4,910
90°	0.246	0.444	7,000	7,080	7,170	7,350	7,670	8,180	7,810
120°	0.249	0.444	7,000	6,880	6,860	6,880	6,955	7,110	9,950
150°	0.246	0.444	7,000	6,740	6,600	6,390	6,115	5,700	7,820
180°	0.244	0.445	7,000	6,630	6,400	6,045	5,580	4,900	—
210°	0.244	0.445	7,000	6,720	6,550	6,340	6,090	5,770	—
240°	0.245	0.445	7,000	6,950	6,940	7,015	7,150	7,470	—
270°	0.248	0.444	7,000	7,120	7,215	7,420	7,690	8,205	—
300°	0.255	0.444	7,000	6,855	6,790	6,775	6,760	6,820	—
330°	0.262	0.444	7,000	6,710	6,570	6,345	6,100	5,630	—

Collapse Pressure = 383 psi

Outside Radius = 4.506"

Diameter readings in inches

0-180°	9.102	90-270°	8.910
15-195°	9.099	105-285°	8.953
30-210°	9.069	120-300°	9.006
45-225°	9.028	135-315°	9.047
60-240°	8.986	150-330°	9.075
75-255°	8.940	165-345°	9.097

Table IV

Analysis of Variance for Error and Regression

Source	SS	df	MS	F	Prob > F
Regression	10.00	1	10.00	1.00	.3183
Error	10.00	1	10.00	1.00	.3183
Total	20.00	2			

MS = Mean Square

Prob > F = Probability of F-value exceeding critical value

df = degrees of freedom

Regression	10.00	1	10.00	1.00	.3183
Error	10.00	1	10.00	1.00	.3183
Total	20.00	2			

Table VII
Dimensions and Strain Data for Ring #5

Angle	h in.	b in.	0	Strains in micro-inches per inch at pressures indicated							
				49	99	150	200	252	303	328	353
0°	0.244	0.444	9,000	8,910	8,625	8,290	7,880	7,390	6,630	6,120	5,250
30°	0.243	0.441	9,000	8,870	8,650	8,400	8,100	7,750	7,260	6,960	6,455
60°	0.243	0.444	9,000	8,860	8,860	8,880	8,930	9,030	9,280	9,495	9,880
90°	0.247	0.445	9,000	9,020	9,175	9,415	9,720	10,140	10,800	11,245	12,100
120°	0.256	0.445	9,000	8,925	8,890	8,890	8,925	9,000	9,170	9,295	9,580
150°	0.262	0.444	9,000	8,875	8,695	8,470	8,210	7,900	7,470	7,200	6,760
180°	0.261	0.443	9,000	8,850	8,605	8,260	7,860	7,360	6,650	6,200	5,380
210°	0.253	0.445	9,000	8,900	8,710	8,470	8,215	7,880	7,430	7,130	7,595
240°	0.250	0.445	9,000	8,930	8,910	8,920	8,960	9,030	9,200	9,320	9,590
270°	0.248	0.444	9,000	8,955	9,100	9,330	9,600	9,990	10,625	11,060	11,905
300°	0.249	0.445	9,000	8,900	8,885	8,915	8,965	9,075	9,320	9,530	9,965
330°	0.247	0.445	9,000	8,920	8,765	8,560	8,340	8,090	7,710	7,485	7,115

Collapse Pressure = 365 psi.

Diameter readings in inches

0-180°	9.138	90-270°	8.856
15-195°	9.122	105-285°	8.923
30-210°	9.097	120-300°	8.994
45-225°	9.046	135-315°	9.041
60-240°	8.985	150-330°	9.086
75-255°	8.920	165-345°	9.122

Outside Radius = 4.508"

Table VIII
Dimensions and Strain Data for Ring #6

Angle	h in	b in	0	Strains in micro-inches per inch at pressures indicated				
				49	99	150	200	252
0°	0.244	0.444	9,000	8,740	8,460	8,105	7,580	6,850
30°	0.243	0.445	9,000	8,780	8,560	8,290	7,945	7,475
60°	0.244	0.445	9,000	8,960	8,950	8,970	9,080	9,290
90°	0.244	0.445	9,000	9,155	9,340	9,610	10,035	10,690
120°	0.248	0.445	9,000	8,990	8,990	9,020	9,120	9,315
150°	0.249	0.445	9,000	8,830	8,600	8,340	8,005	7,540
180°	0.247	0.445	9,000	8,720	8,390	7,975	7,445	6,670
210°	0.252	0.445	9,000	8,840	8,685	8,490	8,240	7,885
240°	0.257	0.445	9,000	8,995	9,015	9,075	9,200	9,410
270°	0.261	0.445	9,000	9,100	9,230	9,430	9,770	10,330
300°	0.260	0.444	9,000	8,990	8,980	9,010	9,115	9,330
330°	0.253	0.444	9,000	8,870	8,705	8,520	8,280	7,950

Collapse Pressure = 345 psi.

Outside Radius = 4.506"

Diameter readings in inches		
0-180°	9.158	8.858
15-195°	9.134	8.908
30-210°	9.093	8.986
45-225°	9.038	9.042
60-240°	8.968	8.910
75-255°	8.905	9.142

Table IX
Dimensions and Strain Data for Ring #7.

Angle θ in.	b in	Strains in micro-inches per inch at pressures indicated									
		0	49	99	150	200	226	242	273	288	
0°	0.252	0.444	9,000	8,695	8,160	7,510	6,700	6,230	5,890	4,920	3,775
30°	0.245	0.444	9,000	8,850	8,465	8,010	7,460	7,180	6,960	6,375	5,850
60°	0.244	0.443	9,000	9,020	9,130	9,280	9,550	9,735	9,880	10,300	10,655
90°	0.244	0.444	9,000	9,140	9,630	10,210	10,950	11,380	11,700	12,725	14,110
120°	0.245	0.444	9,000	8,990	9,110	9,260	9,500	9,660	9,770	10,150	10,450
150°	0.248	0.443	9,000	8,850	8,460	8,130	7,630	7,340	7,140	6,590	6,090
180°	0.249	0.444	9,000	8,685	8,090	7,460	6,635	6,140	5,790	4,760	3,510
210°	0.248	0.444	9,000	8,790	8,485	8,110	7,660	7,420	7,260	6,810	6,465
240°	0.252	0.444	9,000	9,010	9,150	9,290	9,570	9,750	9,900	10,355	10,780
270°	0.257	0.444	9,000	9,230	9,680	10,200	10,910	11,330	11,620	12,570	13,590
300°	0.262	0.443	9,000	8,985	9,130	9,295	9,560	9,710	9,815	10,210	10,575
330°	0.259	0.442	9,000	8,800	8,520	8,220	7,820	7,585	7,430	7,025	6,730

Collapse Pressure = 290 psi

Outside Radius = 4.508"

Diameter readings in inches.

0-180°	9.293	90-270°	8.679
15-195°	9.259	105-285°	8.793
30-210°	9.155	120-300°	8.929
45-225°	9.052	135-315°	9.067
60-240°	8.905	150-330°	9.168
75-255°	8.775	165-345°	9.265

Table X
Dimensions and Strain Data for Ring #8

Angle	h in.	b in	0	49	99	150	200	226	242	252
			Strains in micro-inches per inch at pressure indicated							
0°	0.263	0.445	9,000	8,660	8,240	7,690	6,935	6,570	6,250	5,940
30°	0.260	0.444	9,000	9,075	9,165	9,325	9,600	9,770	9,900	10,050
60°	0.253	0.445	9,000	9,400	9,930	10,635	11,580	12,200	12,705	13,260
90°	0.250	0.445	9,000	9,110	9,310	9,650	10,170	10,530	10,810	11,110
120°	0.243	0.444	9,000	8,730	8,440	8,100	7,620	7,360	7,185	7,040
150°	0.249	0.445	9,000	8,470	7,800	6,955	5,820	5,070	4,385	3,540
180°	0.246	0.445	9,000	8,680	8,260	7,710	7,040	6,630	6,330	6,050
210°	0.244	0.445	9,000	9,080	9,215	9,430	9,790	10,040	10,230	10,440
240°	0.243	0.446	9,000	9,440	10,010	10,775	11,310	12,520	13,110	13,860
270°	0.244	0.445	9,000	9,075	9,220	9,460	9,835	10,080	10,275	10,465
300°	0.247	0.445	9,000	8,675	8,310	7,835	7,240	6,870	6,585	6,320
330°	0.257	0.444	9,000							

Collapse Pressure = 257 psi

Outside Radius = 4.509"

Diameter readings in inches

0-180°	9.386	90-270°	8.571
15-195°	9.341	105-285°	8.686
30-210°	9.207	120-300°	8.847
45-225°	9.066	135-315°	9.026
60-240°	8.879	150-330°	9.167
75-255°	8.659	165-345°	9.286

Table 1. Physical properties of water at 1 atm

Temp, °C	Temp, °F	Density, g/cm ³	Specific Volume, cm ³ /g	Enthalpy, kJ/kg	Entropy, kJ/kg·K
0	32	999.84	1.00016	0.01	0.001
10	50	999.70	1.00030	42.01	0.151
20	68	998.21	1.00170	83.94	0.296
30	86	996.58	1.00340	125.76	0.437
40	104	994.79	1.00520	167.57	0.575
50	122	992.72	1.00710	209.38	0.711
60	140	990.38	1.00910	251.18	0.845
70	158	987.77	1.01120	292.98	0.978
80	176	984.89	1.01340	334.79	1.110
90	194	981.75	1.01570	376.60	1.241
100	212	978.41	1.01810	418.41	1.371
110	230	974.79	1.02060	460.22	1.501
120	248	970.91	1.02320	502.03	1.631
130	266	966.78	1.02590	543.84	1.761
140	284	962.41	1.02870	585.65	1.891
150	302	957.81	1.03160	627.46	2.021
160	320	952.99	1.03460	669.27	2.151
170	338	947.96	1.03770	711.08	2.281
180	356	942.73	1.04090	752.89	2.411
190	374	937.31	1.04420	794.70	2.541
200	392	931.70	1.04760	836.51	2.671
210	410	925.91	1.05110	878.32	2.801
220	428	920.00	1.05470	920.13	2.931
230	446	913.99	1.05840	961.94	3.061
240	464	907.89	1.06220	1003.75	3.191
250	482	901.70	1.06610	1045.56	3.321
260	500	895.43	1.07010	1087.37	3.451
270	518	889.09	1.07420	1129.18	3.581
280	536	882.69	1.07840	1170.99	3.711
290	554	876.24	1.08270	1212.80	3.841
300	572	869.75	1.08710	1254.61	3.971
310	590	863.22	1.09160	1296.42	4.101
320	608	856.65	1.09620	1338.23	4.231
330	626	850.05	1.10090	1379.04	4.361
340	644	843.42	1.10570	1419.85	4.491
350	662	836.76	1.11060	1460.66	4.621
360	680	830.07	1.11560	1501.47	4.751
370	698	823.36	1.12070	1542.28	4.881
380	716	816.63	1.12590	1583.09	5.011
390	734	809.88	1.13120	1623.90	5.141
400	752	803.11	1.13660	1664.71	5.271
410	770	796.32	1.14210	1705.52	5.401
420	788	789.51	1.14770	1746.33	5.531
430	806	782.68	1.15340	1787.14	5.661
440	824	775.83	1.15920	1827.95	5.791
450	842	768.96	1.16510	1868.76	5.921
460	860	762.07	1.17110	1909.57	6.051
470	878	755.17	1.17720	1950.38	6.181
480	896	748.26	1.18340	1991.19	6.311
490	914	741.34	1.18970	2032.00	6.441
500	932	734.41	1.19610	2072.81	6.571

Source: NIST Thermophysical Properties of Matter

0	32	999.84	1.00016	0.01	0.001	0.001
10	50	999.70	1.00030	42.01	0.151	0.001
20	68	998.21	1.00170	83.94	0.296	0.001
30	86	996.58	1.00340	125.76	0.437	0.001
40	104	994.79	1.00520	167.57	0.575	0.001
50	122	992.72	1.00710	209.38	0.711	0.001
60	140	990.38	1.00910	251.18	0.845	0.001
70	158	987.77	1.01120	292.98	0.978	0.001
80	176	984.89	1.01340	334.79	1.110	0.001
90	194	981.75	1.01570	376.60	1.241	0.001
100	212	978.41	1.01810	418.41	1.371	0.001
110	230	974.79	1.02060	460.22	1.501	0.001
120	248	970.91	1.02320	502.03	1.631	0.001
130	266	966.78	1.02590	543.84	1.761	0.001
140	284	962.41	1.02870	585.65	1.891	0.001
150	302	957.81	1.03160	627.46	2.021	0.001
160	320	952.99	1.03460	669.27	2.151	0.001
170	338	947.96	1.03770	711.08	2.281	0.001
180	356	942.73	1.04090	752.89	2.411	0.001
190	374	937.31	1.04420	794.70	2.541	0.001
200	392	931.70	1.04760	836.51	2.671	0.001
210	410	925.91	1.05110	878.32	2.801	0.001
220	428	920.00	1.05470	920.13	2.931	0.001
230	446	913.99	1.05840	961.94	3.061	0.001
240	464	907.89	1.06220	1003.75	3.191	0.001
250	482	901.70	1.06610	1045.56	3.321	0.001
260	500	895.43	1.07010	1087.37	3.451	0.001
270	518	889.09	1.07420	1129.18	3.581	0.001
280	536	882.69	1.07840	1170.99	3.711	0.001
290	554	876.24	1.08270	1212.80	3.841	0.001
300	572	869.75	1.08710	1254.61	3.971	0.001
310	590	863.22	1.09160	1296.42	4.101	0.001
320	608	856.65	1.09620	1338.23	4.231	0.001
330	626	850.05	1.10090	1379.04	4.361	0.001
340	644	843.42	1.10570	1419.85	4.491	0.001
350	662	836.76	1.11060	1460.66	4.621	0.001
360	680	830.07	1.11560	1501.47	4.751	0.001
370	698	823.36	1.12070	1542.28	4.881	0.001
380	716	816.63	1.12590	1583.09	5.011	0.001
390	734	809.88	1.13120	1623.90	5.141	0.001
400	752	803.11	1.13660	1664.71	5.271	0.001
410	770	796.32	1.14210	1705.52	5.401	0.001
420	788	789.51	1.14770	1746.33	5.531	0.001
430	806	782.68	1.15340	1787.14	5.661	0.001
440	824	775.83	1.15920	1827.95	5.791	0.001
450	842	768.96	1.16510	1868.76	5.921	0.001
460	860	762.07	1.17110	1909.57	6.051	0.001
470	878	755.17	1.17720	1950.38	6.181	0.001
480	896	748.26	1.18340	1991.19	6.311	0.001
490	914	741.34	1.18970	2032.00	6.441	0.001
500	932	734.41	1.19610	2072.81	6.571	0.001

Table XI
Dimensions and Strain Data for Ring #2

Angle	h in.	b in	0	Strains in micro-inches per inch at pressures indicated.				
				49	99	150	180	200
0°	0.248	0.443	9,000	8,290	7,425	6,260	5,380	4,560
30°	0.244	0.444	9,000	8,570	8,055	7,390	6,910	6,495
60°	0.244	0.444	9,000	9,150	9,395	9,780	10,120	10,410
90°	0.243	0.444	9,000	9,620	10,430	11,490	12,320	13,130
120°	0.244	0.443	9,000	9,190	9,435	9,820	10,140	10,420
150°	0.252	0.444	9,000					
180°	0.261	0.444	9,000	8,290	7,415	6,280	5,430	4,630
210°	0.261	0.445	9,000	8,640	8,220	7,670	7,285	6,960
240°	0.258	0.444	9,000	9,160	9,390	9,745	10,040	10,310
270°	0.251	0.445	9,000	9,570	10,310	11,275	12,025	12,750
300°	0.249	0.443	9,000	9,180	9,440	9,850	10,200	10,520
330°	0.250	0.443	9,000	8,650	8,215	7,645	7,260	6,930

Collapse Pressure = 220 psi

Outside Radius = 4.510"

Diameter readings in inches.

0-180°	9.532	90-270°	8.401
15-195°	9.452	105-285°	8.549
30-210°	9.254	120-300°	8.798
45-225°	9.021	135-315°	9.042
60-240°	8.794	150-330°	9.259
75-255°	8.575	165-345°	9.447

II. RESULTS

Calculated curves are shown in Fig. 1 for various values of α .

α	$\alpha = 0$	$\alpha = 0.1$	$\alpha = 0.2$	$\alpha = 0.3$	$\alpha = 0.4$	$\alpha = 0.5$	$\alpha = 0.6$	$\alpha = 0.7$	$\alpha = 0.8$	$\alpha = 0.9$	$\alpha = 1.0$
0	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
10	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
20	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
30	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
40	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
50	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
60	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
70	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
80	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
90	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
100	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000

Fig. 1. $\alpha = 0.1, 0.2, 0.3, 0.4, 0.5, 0.6, 0.7, 0.8, 0.9, 1.0$

Fig. 2. $\alpha = 0.1, 0.2, 0.3, 0.4, 0.5, 0.6, 0.7, 0.8, 0.9, 1.0$

0.1	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.2	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.3	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.4	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.5	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.6	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.7	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.8	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
0.9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
1.0	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000

APPENDIX D

Literature Citations

- (1) J. M. MONCRIEFF, "The Practical Column Under Central or Eccentric Loads", Transactions of the American Society of Civil Engineers, Vol. XLV, 1901, pp. 334-431.
- (2) M. LEVY, "Buckling of Circular Rings and Tubes Under Uniform External Pressure", J. Math Pure et Appl. (Liouville), Vol. 10, No. 3, 1884, pp. 5.
- (3) A. P. BORESI, Thesis submitted at University of Illinois, 1952.
- (4) S. P. TIMOSHENKO, "Strength of Materials", Part II, Second Edition, August 1941, D. Van Nostrand Company, Inc., 250 Fourth Avenue, New York, pp. 101 -103, 216 - 224.
- (5) "Metals Handbook", 1948 Edition, pp. 109-111, 822.
- (6) D. A. POLYCHRON, "On the Buckling of Intermittently Supported Plates", Doctor of Science Thesis, M.I.T., 1949, pp. 64-73.
- (7) M. L. PITTMAN and V. W. RINEHART, "On Providing Uniform Edge Compression Loads for Wide Flat Plates", N.A. Thesis, 1954.

APPENDIX B

Literature Cited

- (1) L. W. MOOREHEAD, "The Practical Column Under General or Eccentric Loads", Transactions of the American Society of Civil Engineers, Vol. XIV, 1901, pp. 334-351.
- (2) W. LEVY, "Buckling of Circular Rings and Tubes Under Uniform External Pressure", J. Math. Phys. 27, 1948, (Madisonville), Vol. 10, No. 3, 1944, pp. 3.
- (3) A. P. BOGOSI, Thesis submitted at University of Illinois, 1952.
- (4) R. F. TIMOSHENKO, "Strength of Materials", Part II, Second Edition, August 1941, D. Van Nostrand Company, Inc., 250 Fourth Avenue, New York, pp. 101-103, 216-224.
- (5) "Steel Handbook", 1948 Edition, pp. 109-111, 822.
- (6) D. A. HUYCHEN, "On the Buckling of Internally Supported Pipes", Doctor of Science Thesis, M.I.T., 1948, pp. 64-73.
- (7) R. L. FITTMAN and V. W. RIMBART, "On Providing Uniform Edge Compression Loads for Wide Flat Plates", R.A. Wessels, 1954.

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